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WIND-TUNNEL INVESTIGATION OF THE AERODYNAMIC AND STRUCTURAL-DEFLECTION CHARACTERISTICS OF AN INFLATABLE AIRPLANE

By H. Clyde McLemore

Langley Research Center
Langley Station, Hampton, Va.

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SUMMARY

An investigation has been conducted in the Langley full-scale tunnel to determine the aerodynamic and structural-deflection characteristics of a single-place inflatable airplane over a range of test velocities from about 77 ft/sec to 135 ft/sec. Aerodynamic force data, wing-distortion photographs, and wing-guy-cable loads were recorded at each test speed for a range of angles of attack to the stall or to wing buckle.

At a normal inflation pressure of 7.0 lb/sq in. load factors of the order of 4.50 could be obtained for a test velocity of 134.3 ft/sec. For a reduced inflation pressure of 4.8 lb/sq in. a maximum load factor of 3.09 was obtained at a test velocity of 113.0 ft/sec. At an inflation pressure of 2.0 lb/sq in. a maximum load factor of 1.33 was obtained for a test velocity of 78.5 ft/sec, and the airplane would not be safe for flight at such a low inflation pressure.

INTRODUCTION

Because of wing-buckling failures during flight tests of inflatable airplanes, evaluating teams have jointly recommended some changes in the restraint-cable attachment points. These changes are intended to increase the wing load factor before wing buckle. The inflatable airplane is certain to be torn by the propeller and deflated if buckling occurs during flight; therefore, a configuration having a wing cable and structural relationship sufficiently strong to preclude a wing buckle within the normal operating limits of the airplane would certainly be desirable.

The inflatable airplane of reference 1 (identified herein as Inflatoplane I) was previously tested in the Langley full-scale tunnel where the largest load factor obtainable prior to buckle was found to be

about 2.5 at the normal inflation pressure of 7.0 lb/sq in. The intent of the present series of tests of an inflatable airplane (identified herein as Inflatoplane II), which were also made in the Langley full-scale tunnel, was to determine whether load factors of the order of 4.50 to 5.00 could be obtained with appropriate modifications.

The tests of the present investigation were conducted at angles of attack varying from about -8° to that required for maximum lift or for wing buckle. Measurements of the surface-pressure distributions were made for most of the test configurations, and for one of the best configurations the wing structural integrity was determined with full aileron deflection at high wing loadings. The tests were conducted for a range of Reynolds numbers, based on wing chord, varying from 2.57×10^6 to 4.57×10^6 which corresponds to a test velocity range of about 77 ft/sec to 135 ft/sec.

SYMBOLS

The data of the subject report are referred to the stability system of axes the origin of which is located at the model center of gravity located longitudinally at fuselage station 72.7 and vertically at water line 45.3. See figure 1.

b	wing span, ft
c	wing chord, ft
$\bar{c}$	mean aerodynamic chord, ft
C_D	drag coefficient, Drag/$q_\infty S$
C_L	lift coefficient, Lift/$q_\infty S$
C_l	rolling-moment coefficient, Rolling moment/$q_\infty Sb$
C_m	pitching-moment coefficient, Pitching moment/$q_\infty S\bar{c}$
C_n	yawing-moment coefficient, Yawing moment/$q_\infty Sb$
C_p	pressure coefficient, $\frac{p_l - p_\infty}{q_\infty}$

C_Y side-force coefficient, Side force/$q_\infty S$

c_n wing-section normal-force coefficient, $\int_0^{1.0} C_{p,l}\, d(x/c) - \int_0^{1.0} C_{p,u}\, d(x/c)$

i_w mean incidence of wing (angle between wing chord and water line 50, see fig. 1), deg

n load factor, $\frac{C_L q_\infty}{W/S}$

p wing and fuselage inflation pressure, lb/sq in.

p_l local static pressure, lb/sq ft

p_∞ free-stream static pressure, lb/sq ft

q_∞ free-stream dynamic pressure, lb/sq ft

S wing area, sq ft

V_∞ free-stream velocity, ft/sec

W weight of airplane, 550 lb

x chordwise distance measured parallel to the plane of symmetry, ft

y lateral distance measured perpendicular to the plane of symmetry, ft

α_f angle of attack of fuselage (angle between relative wind and water line 50, see fig. 1), deg

δ_a right-hand aileron deflection (positive when trailing edge deflected down), deg

δ_r rudder deflection (positive when trailing edge deflected toward the pilot's right hand), deg

Subscripts:

u	upper surface
l	lower surface
max	maximum

AIRPLANE AND APPARATUS

The inflatable airplane used in the present investigation was composed of pneumatic structure throughout with the exceptions of the engine, engine mount, landing gear, support cables, and miscellaneous short control members. All inflatable components were interconnected in a manner allowing, during flight operation, a small compressor on the 40-horsepower air-cooled engine to maintain a constant regulated pressure in the system even with moderate leakage. An external air supply was substituted for the normal engine compressor because the engine was not operated during the tests. The wing and tail surfaces are woven in a manner such that the upper and lower airfoil surfaces were connected internally by nylon drop threads varying in length to produce the shape desired in any surface when inflated. A circular cross-sectional fuselage was utilized with a fuel bag internally mounted, and the cockpit section was constructed of sections of an inflatable material 2 inches thick.

Each wing panel was restrained by two guy cables on the upper surface and by either three or four cables attached to the lower surface. The two upper cables were anchored to the engine pylon and the lower cables were attached to the landing gear and to the fuselage rearward of the wing trailing edge. Each guy cable had a calibrated strain-gage link inserted in the cable and the instrumentation was such that all cable loads were recorded simultaneously. A general layout with pertinent geometric data is shown in figure 1, and a photograph of a similar airplane, Inflatoplane I, mounted in the Langley full-scale tunnel is given as figure 2. No such photograph of the present airplane was obtained as the tests were terminated, because of wing failure, before any general photographs were made. The principal external differences between Inflatoplane I and Inflatoplane II are that the configuration used in this investigation has a slightly larger wing, no cockpit canopy, and better shape control of the airfoil contour. The propeller was not installed for the present tests.

The airplane was mounted for tests on the conventional six-component tunnel mechanical balance system. A special yoke was utilized to mount

the airplane so that restraining loads were transmitted to the fuselage through strap attachments located beneath the wing quarter-chord point; thus, the wings were free to deflect while being restrained, as in flight, only by the normal wing-fuselage and guy-cable attachments. Provision was made to change cable configurations by means of extra attachment points beneath the wing. The various lower surface cable attachment points with their number designations are shown in figure 3. Cable configuration was designated in terms of the numbers of the attachment points. For example, a cable configuration designation of 2-5-8 means that three lower surface cables were used and these cables were attached to the wing at positions 2, 5, and 8. The rear mounting strut was attached to a saddle strapped to the rearward portion of the fuselage and was connected by cables to the front support yoke to prevent longitudinal tail strut loads from being transmitted into the fuselage. Restraint cables were mounted above each wing to control the amount of deflection during buckling and, thus, minimize structural damage caused by the buckles.

An actuator system was installed in the cockpit to allow remote operation of the ailerons. Movie and still cameras were set up to record the deflection of the right-hand wing panel under various loading conditions. The right-hand wing panel was chosen for photographic study because the wing contours and geometry were more uniform than those of the left-hand panel, and in the event of wing buckle, the right-hand panel was expected to buckle first.

The right-hand wing panel was equipped with five (10- and 12-tube) plastic belts glued to the wing surface in a chordwise direction with a hole punched in each tube to provide wing-surface orifices for determining the pressure-distribution characteristics of the wing under various loading conditions. The spanwise locations of the pressure-distribution belts were 16.21, 28.95, 50.00, 68.00, and 89.20 percent of the semispan and are referred to hereinafter in this report as stations 1 to 5, respectively. (See fig. 4.) The chordwise locations of the orifices at the 5 spanwise stations are also given in figure 4. The orifice pressures were measured on a multiple-tube manometer and photographically recorded.

TESTS AND CORRECTIONS

The primary objective of the present tests was to determine the maximum loading, prior to wing buckle, of the unmodified Inflatoplane II (cable configuration 2-5-8) and if possible to improve the loading characteristics of the airplane by the addition of a wing guy cable or by relocation of some or all of the wing restraint guy cables. In addition

to the aerodynamic force, the wing-deflection characteristics and the pressure-distribution characteristics were determined for all the test configurations; however, camera malfunction during the tests caused some portions of the photographs to be unreadable. Consequently, complete pressure-distribution data for all stations and test configurations are not presented. The tests were conducted at various airspeeds ranging from approximately 77 ft/sec to 135 ft/sec corresponding to Reynolds numbers, based on wing chord, of 2.57×10^6 to 4.57×10^6. The angle of attack of the fuselage increased at each airspeed from about -8° to the angle at which the wing stalled or buckled. The wing had an average incidence of 7.2° relative to water line 50. (See fig. 1.) Most of the tests were conducted at the normal inflation pressure of 7.0 lb/sq in.; however, a few tests were conducted at inflation pressures of 4.8 lb/sq in. and 2.0 lb/sq in.

Wing cable loads were recorded for all test configurations, and the wing deflections during most of the tests were photographically recorded by still and movie cameras. The cable positions available for use during the tests are shown in figure 3. The method used for all the tests, after force data on the original configuration were obtained, was to first visually observe the wing stalling or buckling characteristics and the cable loads for a given test velocity. Further force and pressure measurements were then made for only the better configurations. The cable positions shown in figure 3 were utilized in the following manner: one or two cables attached to position 1, 2, or 3; one to position 4, 5, or 6; and one to position 7, 8, or 9. A few visual tests were made for modified cable arrangements having the attachment points at the landing gear, moved forward but these tests did not warrant further investigation because of poor loading characteristics.

For the aileron tests, only the right-hand aileron was deflected so that the effect of individual aileron movement, up or down, could be determined. For the tests with the rudder deflected, the rudder was locked by cables to a fully deflected position of full right rudder.

All the data presented in this paper have been corrected for airstream misalinement ($\Delta\alpha = -0.5^\circ$; $\Delta C_L = -0.0087 C_L$; $\Delta C_D = -0.0087 C_D$), buoyancy ($\Delta C_D = 0.0$), and jet boundary ($\Delta\alpha = -0.760 C_L$; $\Delta C_D = -0.013 C_L^2$). Support strut tares were not measured since major emphasis was placed on obtaining loads information. All drag results, therefore, include the tare drag of the support system.

RESULTS AND DISCUSSION

A motion-picture film supplement to this paper has been prepared and is available on loan. A request card form and a description of the

film will be found at the back of this paper on the page immediately preceding the abstract and index pages.

Static Longitudinal and Wing-Buckling Characteristics

The variations of the lift, drag, and pitching-moment characteristics of the inflatable airplane for the normal inflation pressure of 7.0 lb/sq in., for several test airspeeds and two wing-guy-cable configurations are shown in figures 5 and 6. The data of figure 5 were obtained by utilizing the basic airplane as received from the manufacturer. In general the data of figure 5 show that the airplane is longitudinally stable through the stall. The right-hand wing buckled just prior to $C_{L,max}$ at the test velocity of 124.3 ft/sec which resulted in a load factor of 4.19. The maximum load factor achieved for a speed at which the airplane reached stall was 3.63 at a velocity of 112.7 ft/sec.

In an attempt to increase the load on the wing prior to stall or buckle, qualitative tests were made of various cable arrangements. Apparently, the best of the arrangements tested was configuration 1-3-6-8. During these tests, however, the largest load factor obtained was only about 4.22, which was below the desired value of 4.50 to 5.00.

It was noted that the landing-gear structure was deflecting into the belly of the fuselage just prior to wing buckle. This deflection was attributed to the wing-guy-cable load because the guy cable was anchored to the landing-gear structure. This deflection would effectively lengthen the guy cable which would in turn allow the wing to deflect more and thus buckle more readily. A 9-inch-square, 1/2-inch-thick plate of aluminum was placed between the landing-gear tubular structure and the inflated belly of the fuselage to provide additional load-bearing area and, thus, minimize the landing-gear deflection under cable load. In addition to the belly plate, the guy cables were heavily tightened in the static condition. The results of the tests made with these modifications to cable configuration 1-3-6-8 are shown in figure 6. The maximum load factor at a velocity of 123.6 ft/sec remained at 4.22 which indicated that the cable and plate modification had not improved the load-carrying capabilities of the airplane. The wing deflection occurred in a different manner with the belly plate installed; however, to be sure of the best cable configuration, the cables were rerigged to the original cable configuration 2-5-8 with the belly plate installed and with the guy cables heavily tightened. For this configuration at a velocity of 122.0 ft/sec the wing did not buckle and the load factor at $C_{L,max}$ was 4.17. The test velocity was arbitrarily increased to 134.3 ft/sec to produce a buckle in order to determine the maximum load factor of cable configuration 2-5-8. The data for this configuration are shown in figure 6. The load factor for this test condition was 4.50 which was within the desired load range so no further attempts were made to increase the airplane load factor.

Two types of buckles were noticed during the tests. A root buckle occurred when the larger of two shear wrinkles at the outboard edge of the fuselage bulkhead straps lengthened to about the quarter chord. These wrinkles first progressed rearward and outward at about 45° and then straight rearward. As the wrinkles neared the quarter chord several short wrinkles appeared parallel to the long wrinkle just before buckle occurred at the root section. A patch buckle (buckle at cable attachment point) occurred just over one of the patches that supported the lower cables. Depressions in the wing upper surface above the cable attachment points can be seen in the film supplement. As the cable load increased the depressions deepened. A patch buckle occurred when numerous short chordwise wrinkles appeared in the depressed areas of the inboard cable attachment points. A patch buckle was usually preceded by an erratic rise and fall of the portion of the wing outboard of the outboard cable. An example of a typical load buildup and buckle is shown in the photographs of figure 7. This figure also shows the effect of the forward and rearward cables over the wing in limiting the upward movement of the wing which thus minimized damage when the wing buckled.

In order to determine whether the fuselage and wing structure would be strong enough to withstand flight at some reduced speed with a reduced inflation pressure, several tests were conducted at reduced inflation pressures. The results of these tests are shown in figure 8. At an inflation pressure of 4.8 lb/sq in. and a test speed of 93.6 ft/sec the wing reached $C_{L,max}$ and a load factor of 2.32 without buckling. The velocity was arbitrarily increased to 113.0 ft/sec and the maximum load factor obtained at buckle was 3.09. At an inflation pressure of 2.0 lb/sq in. the maximum load factor reached at buckle was 1.33 at a velocity of 78.5 ft/sec. At this speed one would be flying very close to buckle in unaccelerated level flight; therefore, an inflation pressure of 2.0 lb/sq in. should be considered to be too low for flight.

During some of the high-loading tests, which were repeated to obtain pressure-distribution data that were not obtained earlier because of camera malfunction, the wing tore loose from the wing-fuselage bulkhead. The wing was damaged severely and the test program was terminated. It was later found during study of the motion-picture film that the failure resulted from a broken right-hand wing guy cable which had probably been damaged during previous buckle tests. The broken cable was attached to the wing in the number 5 position. Incidentally, when the cable broke the load factor was estimated to be about 4.32, and the cable load was estimated to be approximately 1,100 pounds, which is well below the design breaking strength of about 3,700 pounds.

Static Lateral Characteristics

The static lateral characteristics of the airplane with right-hand aileron deflections of -15°, 0°, and 26° are shown in figures 9 and 10. Deflection of one aileron was done to isolate the effects of up and down deflection. The reason for the reduced rolling moment with aileron deflection and with increased speed is that the deflected aileron was twisting the wing. Incidentally, the motion-picture film of the aileron tests shows considerable aileron flutter or buffet. One contributing factor to the small amount of flutter which occurred during some of the tests simulating normal flight conditions was weak bungee chords in the aileron restraint system. The problems that could arise in the event of aileron flutter should definitely be recognized and appropriate steps should be taken to prevent their occurrence in flight.

Aileron deflection is seen to produce adverse yaw (fig. 10). Tests for rudder effectiveness were not made, but visual tests for fuselage torsional stiffness were made for several airspeeds with the rudder fully deflected and single-point data were obtained for each airspeed. The data, figure 11, showed that the rudder would provide adequate moments to counteract the adverse yaw and still have sufficient power for maneuvering.

Aerodynamic wing-guy-cable loads for several of the test conditions are given without analysis in figures 12 to 15. The initial loading of the cables for the zero-speed condition was not determined. For each test the load recording instruments were set at zero load for the zero-speed condition; thus, the data show the change in cable loading caused by aerodynamic forces and moments.

Pressure-Distribution Characteristics

A complete listing of the pressure coefficients obtained during the subject investigation is given in tables 1 to 16. A listing of the tables with pertinent information concerning them is as follows:

Table	Cable configuration	Guy-cable tension	Belly plate	q_∞, lb/sq ft	p, lb/sq in.	α_f, deg	δ_a, deg
1	2-5-8	Light	Off	6.7	7.0	-8.5 to 11.8	0
2	2-5-8	Light	Off	9.9	7.0	-8.5 to 12.9	0
3	2-5-8	Light	Off	14.2	7.0	-8.5 to 11.9	0
4	1-3-6-8	Light	Off	17.1	7.0	-8.4 to 3.7	0
5	1-3-6-8	Heavy	On	17.0	7.0	-6.6 to 4.7	0
6	2-5-8	Heavy	On	6.9	7.0	-8.5 to 11.8	0
7	2-5-8	Heavy	On	10.1	7.0	-8.5 to 11.9	0
8	2-5-8	Heavy	On	14.1	7.0	-8.5 to 10.9	0
9	2-5-8	Heavy	On	16.9	7.0	-6.5 to 9.4	0
10	2-5-8	Heavy	On	20.1	7.0	-6.6 to 2.2	0
11	2-5-8	Light	Off	9.9	4.8	-8.5 to 9.8	0
12	2-5-8	Light	Off	11.6	4.8	-8.6 to 1.8	0
13	2-5-8	Light	Off	6.9	2.0	-8.6 to -1.1	0
14	2-5-8	Heavy	On	7.0	7.0	-4.8 to 10.9	-15 to 26
15	2-5-8	Heavy	On	10.2 or 10.8	7.0	-4.8 to 6.7	-15 to 26
16	2-5-8	Heavy	On	14.4 or 14.7	7.0	-4.8 to 7.4	-15 to 26

The wing-surface-pressure distributions are shown in figure 16 for two cable configurations. The data of figure 16 are not intended to show detailed differences in the loading characteristics of the two configurations but are intended to be representative diagrams of the loading of the inflatable wing for varying angles of attack and airspeeds. For detailed analysis, use must be made of the tabulated values of the pressure coefficients.

The pressure-distribution characteristics of the wing with aileron deflection and for two test velocity conditions are shown in figure 17. As one might expect of an inflatable wing, aileron deflection is seen to adversely affect the chordwise loading on the wing forward of the aileron which is the result of wing twist caused by the aileron deflection.

Span Loading Characteristics

Complete span loading plots cannot be made because the spanwise orifice stations were not close enough to the airplane vertical plane of symmetry to accurately determine the fairing of the curves for the inboard locations and because of camera malfunction some of the data for the inboard stations were not obtained; however, the general span loading characteristics of the configurations are shown in figures 18 and 19. Apparently, the reason that the loading shown in figure 18(c) at $\alpha_f = 2.2^\circ$ is lower than that at $\alpha_f = 1.8^\circ$ is that the manometer was photographed just as the wing buckled, because the data of figure 6 and visual observations of the wing showed that maximum lift should have been experienced at $\alpha_f = 2.2^\circ$.

The general span loading characteristics of the wing with the right-hand aileron deflected are shown in figure 19 for test velocities of 94.0 ft/sec and 113.7 ft/sec.

CONCLUDING REMARKS

The results of wind-tunnel tests of an inflatable airplane in the Langley full-scale tunnel indicate that by proper selection of the attachment points of the wing restraint guy cables, by restricting the movement of the lower cable attachment point into the belly of the fuselage, and by heavily tightening the lower guy cables in the static condition, a load factor of 4.50 at a normal inflation pressure of 7.0 lb/sq in. can be obtained prior to buckle for a test velocity of 134.3 ft/sec. For an inflation pressure of 4.8 lb/sq in. a maximum load factor of 3.09 was obtained at a test velocity of 113.0 ft/sec but

for an inflation pressure of 2.0 lb/sq in. the maximum load factor obtained at a test velocity of 78.5 ft/sec was 1.33 which is considered to be too low for flight.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Air Force Base, Va., February 13, 1962.

REFERENCE

1. Cocke, Bennie W., Jr.: Wind-Tunnel Investigation of the Aerodynamic and Structural Deflection Characteristics of the Goodyear Inflatoplane. NACA RM L58E09, 1958.

TABLE I

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.848 .125 -.714 -.518 -.625 -.607 -.554 -.304	.938 .304 -.348 -.366 -.482 -.473 -.357 -.464	.893 .411 -.089 -.214 -.393 -.384 -.295 -.321
Lower	.0500 .1500 .4000 .7000	-.732 -.607 -.348 -.143	-.750 -.580 -.286 .045	-.696 -.491 -.188 .036
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.384 -.357 -.268 -.179 -.089 -.000	-.482 -.313 -.259 -.268 -.107 -.027	-.357 -.339 -.411 -.250 -.063 -.018
Lower	.8100 .8340 .9250 .9750	-.000 .152 -.036 -.018	.125 .259 .089 .027	.223 .179 .080 .045

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:					
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000			1.045	.973	.884	
	.0100			-.339	-.205	-.054	
	.0250			-.902	-.964	-.554	
	.0500			-.902	-.964	-.679	
	.1000			-.625	-.813	-.643	
	.2250			-.473	-.571	-.554	
	.4500			-.366	-.446	-.384	
	.7500			-.188	-.509	-.393	
Lower	.0500			-.098	-.339	-.393	
	.1500			-.107	-.366	-.366	
	.4000			-.107	-.286	-.188	
	.7000			.125	.071	-.036	
Aileron							
Upper	.8050			-.071	-.482	-.402	
	.8150			-.054	-.339	-.411	
	.8350			-.107	-.286	-.527	
	.8750			-.036	-.304	-.321	
	.9250			.125	-.152	-.179	
	.9750			.125	-.018	-.054	
Lower	.8100			.250	.214	.196	
	.8340			.161	.232	.152	
	.9250			.161	.063	.018	
	.9750			.196	-.018	-.018	

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.7$ lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.295 -1.688 -2.188 -1.911 -1.473 -.839 -.589 -.170	.759 -1.304 -2.054 -1.893 -1.205 -.813 -.500 -.464	.813 -.777 -1.241 -1.205 -.938 -.643 -.375 -.375
Lower	.0500 .1500 .4000 .7000	.330 .188 -.080 -.080	.259 -.063 -.063 .152	.071 -.098 -.071 -.009
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.223 -.223 -.223 -.116 -.045 -.045	-.402 -.286 -.188 -.179 -.098 -.054	-.348 -.357 -.473 -.295 -.188 -.098
Lower	.8100 .8340 .9250 .9750	.143 .134 .080 -.063	.286 .277 .080 .045	.205 .152 -.045 -.063

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = 2.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-.446 -3.098 -3.652 -3.179 -2.107 -1.321 -.714 -.339 .161 .107 .116 .071
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.661 .286 .071 -.054 -.036 -.071 -.071

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.464 -2.946 -3.429 -2.875 -1.929 -1.259 -.705 -.321	.143 -2.500 -3.071 -2.759 -1.643 -.938 -.509 -.321	.393 -1.679 -1.929 -1.804 -1.196 -.768 -.366 -.446
Lower	.0500 .1500 .4000 .7000	.420 .116 .107 .241	.580 .232 -.009 .223	.384 .089 -.027 -.036
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.321 -.196 -.205 -.089 -.018 -.089	-.384 -.080 -.098 -.107 -.098 -.098	-.464 -.429 -.509 -.366 -.250 -.196
Lower	.8100 .8340 .9250 .9750	.161 .214 -.036 -.089	.313 .366 .098 .000	.170 .161 -.089 -.170

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.7$ lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.634	-1.598
	.0100	-4.643	-4.857
	.0250	-4.482	-5.063
	.0500	-4.089	-4.366
	.1000	-2.304	-2.393
	.2250	-1.357	-1.339
	.4500	-.920	-.741
	.7500	-.268	-.464
	.8350	-.063	.054
	.8750	.018	-.107
	.9250	-.107	-.089
	.9750	.018	.018
Lower	.0500	.866	.866
	.1500	.589	.384
	.4000	.232	.089
	.7000	.152	.125
	.8530	.063	-.107
	.9250	-.188	-.152
	.9750	-.071	-.170

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.741	-.741	-.223
	.0100	-4.473	-3.830	-2.625
	.0250	-4.786	-4.196	-2.696
	.0500	-4.107	-3.866	-2.464
	.1000	-2.589	-2.071	-1.482
	.2250	-1.393	-1.116	-.839
	.4500	-.732	-.563	-.438
	.7500	-.304	-.277	-.589
Lower	.0500	.714	.857	.634
	.1500	.393	.482	.241
	.4000	.045	.196	.027
	.7000	.018	.268	-.063
Aileron				
Upper	.8050	-.446	-.259	-.670
	.8150	-.241	-.098	-.598
	.8350	-.232	-.080	-.750
	.8750	-.161	-.107	-.589
	.9250	-.054	-.080	-.420
	.9750	-.205	-.071	-.313
Lower	.8100	.080	.330	.080
	.8340	.232	.366	.116
	.9250	-.143	.152	-.098
	.9750	-.080	.045	-.179

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 8.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.464	-2.420
	.0100	-5.652	-5.991
	.0250	-5.205	-5.848
	.0500	-4.268	-4.821
	.1000	-2.384	-2.777
	.2250	-1.589	-1.393
	.4500	-.679	-.741
	.7500	-.286	-.188
	.8350	-.045	.152
	.8750	-.036	-.125
	.9250	-.143	-.143
	.9750	-.080	-.036
Lower	.0500	.813	.929
	.1500	.411	.732
	.4000	.330	.161
	.7000	.179	.080
	.8530	-.045	.143
	.9250	-.196	-.045
	.9750	-.107	-.080

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.500	-1.268	-.527
	.0100	-5.607	-4.616	-3.143
	.0250	-5.598	-4.875	-3.116
	.0500	-4.643	-4.518	-2.821
	.1000	-2.661	-2.295	-1.634
	.2250	-1.482	-1.232	-.911
	.4500	-.768	-.607	-.491
	.7500	-.286	-.286	-.652
Lower	.0500	.830	.938	.714
	.1500	.607	.554	.313
	.4000	.268	.196	.018
	.7000	.125	.259	-.107
Aileron				
Upper	.8050	-.250	-.268	-.821
	.8150	-.196	-.143	-.661
	.8350	-.170	-.098	-.875
	.8750	-.143	-.125	-.714
	.9250	-.125	-.107	-.446
	.9750	-.098	-.107	-.357
Lower	.8100	.295	.348	.036
	.8340	.330	.348	.036
	.9250	-.018	.125	-.179
	.9750	-.000	.009	-.250

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.7$ lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.223	-1.348	-.643
	.0100	-5.277	-4.723	-3.205
	.0250	-5.679	-4.893	-3.125
	.0500	-4.420	-4.607	-2.848
	.1000	-2.571	-2.304	-1.670
	.2250	-1.402	-1.313	-.955
	.4500	-.634	-.634	-.500
	.7500	-.098	-.366	-.732
Lower	.0500	1.018	.875	.652
	.1500	.670	.482	.277
	.4000	.366	.161	.009
	.7000	.098	.161	-.125
Aileron				
Upper	.8050	-.098	-.304	-.813
	.8150	-.161	-.161	-.643
	.8350	.045	-.107	-.839
	.8750	-.054	-.143	-.723
	.9250	-.063	-.161	-.482
	.9750	-.036	-.179	-.357
Lower	.8100	.170	.250	-.027
	.8340	.295	.375	.054
	.9250	.152	.045	-.161
	.9750	-.116	-.036	-.259

TABLE I.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.804	-2.125
	.0100	-4.518	-5.402
	.0250	-4.277	-5.411
	.0500	-3.661	-4.545
	.1000	-2.179	-2.455
	.2250	-1.295	-1.295
	.4500	-.804	-.571
	.7500	-.473	-.143
	.8350	-.491	-.009
	.8750	-.339	-.188
	.9250	-.339	-.232
	.9750	-.232	-.134
Lower	.0500	.875	.821
	.1500	.696	.518
	.4000	.420	.161
	.7000	.045	.098
	.8530	.018	-.009
	.9250	-.339	-.009
	.9750	-.214	-.134

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.598	-1.589	-.902
	.0100	-5.670	-5.098	-3.536
	.0250	-5.705	-5.152	-3.348
	.0500	-4.813	-4.866	-2.973
	.1000	-2.732	-2.384	-1.714
	.2250	-1.384	-1.277	-.982
	.4500	-.679	-.607	-.491
	.7500	-.241	-.268	-.705
Lower	.0500	.973	.929	.750
	.1500	.625	.580	.357
	.4000	.259	.223	.009
	.7000	.125	.188	-.143
Aileron				
Upper	.8050	-.313	-.232	-.857
	.8150	-.277	-.134	-.554
	.8350	-.205	-.089	-.866
	.8750	-.152	-.143	-.750
	.9250	-.098	-.125	-.455
	.9750	-.098	-.116	-.330
Lower	.8100	.170	.277	-.027
	.8340	.277	.366	.036
	.9250	-.036	.098	-.161
	.9750	-.116	.009	-.277

TABLE I.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.7 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(i) $\alpha_f = 11.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-2.036	-2.232	-2.750	-1.536	-.804
	.0100	-4.643	-5.661	-5.679	-4.938	-3.429
	.0250	-4.277	-5.625	-5.875	-5.071	-3.250
	.0500	-3.518	-4.438	-4.723	-4.804	-2.938
	.1000	-2.125	-2.527	-3.027	-2.348	-1.661
	.2250	-1.393	-1.250	-1.464	-1.250	-.955
	.4500	-.580	-.688	-.786	-.589	-.509
	.7500	-.688	-.446	-.286	-.304	-.696
	.8350	-.393	-.089			
	.8750	-.679	-.179			
	.9250	-.536	-.134			
	.9750	-.482	-.196			
Lower	.0500	.652	.866	.911	.929	.714
	.1500	.250	.634	.616	.554	.313
	.4000	.223	.268	.179	.232	.018
	.7000	-.071	.125	.116	.241	-.170
	.8530	-.143	-.089			
	.9250	-.509	-.170			
	.9750	-.554	-.179			
Aileron						
Upper	.8050			-.286	-.241	-.839
	.8150			-.259	-.098	-.563
	.8350			-.196	-.036	-.857
	.8750			-.179	-.143	-.732
	.9250			-.098	-.089	-.482
	.9750			-.134	-.098	-.321
Lower	.8100			.179	.286	-.027
	.8340			.250	.420	.018
	.9250			-.027	.098	-.161
	.9750			-.107	.000	-.277

TABLE II

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.849 .349 -.193 -.205 -.572 -.464 -.404 -.373	.880 .512 -.114 -.199 -.349 -.343 -.277 -.464	.873 .536 .030 -.072 -.301 -.337 -.259 -.349
Lower	.0500 .1500 .4000 .7000	-.729 -.536 -.398 -.084	-.855 -.572 -.283 .054	-.783 -.482 -.199 .066
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.386 -.361 -.289 -.193 -.108 -.064	-.476 -.289 -.066 -.283 -.084 -.006	-.343 -.319 -.422 -.217 -.054 .024
Lower	.8100 .8340 .9250 .9750	.006 .012 .024 .060	.175 .253 .096 .078	.193 .181 .066 .066

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.753 -.669 -1.133 -1.205 -1.133 -.735 -.530 -.506	1.000 -.169 -.940 -.952 -.783 -.560 -.386 -.476	.976 .012 -.482 -.542 -.554 -.500 -.343 -.355
Lower	.0500 .1500 .4000 .7000	-.289 -.307 -.271 -.145	-.277 -.289 -.193 .096	-.361 -.349 -.169 .024
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.482 -.386 -.301 -.253 -.193 -.120	-.464 -.307 -.084 -.223 -.102 -.018	-.361 -.349 -.506 -.307 -.157 -.060
Lower	.8100 .8340 .9250 .9750	.018 .036 -.066 -.000	.217 .265 .078 .042	.229 .193 .036 .012

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			-.000 -2.024 -2.922 -2.289 -1.681 -.886 -.663 -.331 -.060 -.084 -.060 .030
Lower			.470 .084 -.181 -.157 .000 -.042 .024

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.157 -1.898 -2.410 -2.199 -1.633 -.964 -.620 -.235	.645 -1.349 -2.054 -1.910 -1.259 -.801 -.518 -.446	.771 -.807 -1.253 -1.241 -.946 -.663 -.373 -.380
Lower	.0500 .1500 .4000 .7000	.470 .151 -.096 .024	.271 .024 -.054 .120	.066 -.090 -.090 -.006
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.325 -.235 -.211 -.199 -.036 -.024	-.392 -.331 -.090 -.175 -.096 -.060	-.349 -.355 -.500 -.343 -.199 -.151
Lower	.8100 .8340 .9250 .9750	.018 .169 .036 .024	.241 .319 .078 .018	.193 .163 -.030 -.114

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	-3.120	-3.410	-3.084	.084	.386
	.0100	-3.355	-3.867	-3.440	-2.518	-1.614
	.0250	-2.861	-3.313	-2.934	-3.139	-1.886
	.0500	-1.795	-1.946	-1.922	-2.819	-1.771
	.1000	-.831	-1.006	-.922	-1.657	-1.181
	.2250	-.548	-.566	-.458	-.970	-.753
	.4500	-.271	-.295	-.127	-.542	-.386
	.7500	.500	.873	.831	-.361	-.476
	.8350	-.078	.054			
	.8750	-.048	.072			
	.9250	.084	.145			
	.9750	.072	.145			
Lower	.0500	.536	.476	.416	.608	.325
	.1500	.440	.295	.127	.259	.090
	.4000	.199	.193	.139	.042	-.036
	.7000	.000	.139	-.259	.205	-.054
	.8530	-.048	.127			
	.9250	-.084	.078			
	.9750	-.030	.090			
Aileron						
Upper	.8050			-.223	-.325	-.464
	.8150			-.133	-.325	-.392
	.8350			-.078	-.108	-.602
	.8750			-.018	-.120	-.428
	.9250			.036	-.072	-.253
	.9750			.193		-.181
Lower	.8100			.247	.271	.127
	.8340			-.018	.331	.139
	.9250			.054	.102	-.127
	.9750			.006	.024	-.193

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.849	-1.753
	.0100	-4.747	-5.241
	.0250	-4.578	-5.217
	.0500	-3.970	-4.319
	.1000	-2.337	-2.542
	.2250	-1.277	-1.331
	.4500	-.663	-.639
	.7500	-.133	-.205
	.8350	-.181	.024
	.8750	-.078	-.108
	.9250	-.139	-.060
	.9750	-.084	.024
Lower	.0500	.783	1.006
	.1500	.488	.633
	.4000	.169	.277
	.7000	.157	.120
	.8530	.072	.024
	.9250	-.084	.054
	.9750	-.060	-.042

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.886	-.813	-.265
	.0100	-4.795	-3.964	-2.699
	.0250	-4.934	-4.361	-2.699
	.0500	-4.175	-4.036	-2.476
	.1000	-2.530	-2.120	-1.506
	.2250	-1.301	-1.169	-.880
	.4500	-.711	-.584	-.446
	.7500	-.253	-.301	-.608
Lower	.0500	.855	.807	.639
	.1500	.536	.434	.217
	.4000	.108	.151	-.012
	.7000	.139	.175	-.120
Aileron				
Upper	.8050	-.241	-.295	-.645
	.8150	-.175	-.319	-.584
	.8350	-.163	-.114	-.735
	.8750	-.120	-.114	-.614
	.9250	-.096	-.084	-.380
	.9750	-.096	-.078	-.289
Lower	.8100	.169	.301	.030
	.8340	.301	.265	.096
	.9250	.006	.078	-.120
	.9750	-.030	.012	-.229

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 8.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.470	-1.735
	.0100	-4.163	-4.970
	.0250	-4.030	-4.934
	.0500	-3.392	-4.090
	.1000	-1.970	-2.295
	.2250	-1.127	-1.175
	.4500	-.614	-.608
	.7500	-.428	-.223
	.8350	-.331	.018
	.8750	-.289	-.133
	.9250	-.247	-.096
	.9750	-.211	-.060
Lower	.0500	.873	.916
	.1500	.572	.560
	.4000	.199	.205
	.7000	.078	.096
	.8530	.000	.006
	.9250	-.277	-.042
	.9750	-.187	-.114

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.867	-.976	-.367
	.0100	-4.572	-4.084	-2.741
	.0250	-4.976	-4.380	-2.717
	.0500	-4.410	-4.102	-2.458
	.1000	-2.452	-2.066	-1.464
	.2250	-1.247	-1.127	-.825
	.4500	-.693	-.548	-.446
	.7500	-.223	-.259	-.524
Lower	.0500	.880	.831	.620
	.1500	.542	.494	.247
	.4000	.199	.169	.018
	.7000	.163	.175	-.133
Aileron				
Upper	.8050	-.223	-.205	-.596
	.8150	-.193	-.283	-.512
	.8350	-.151	-.048	-.657
	.8750	-.114	-.102	-.560
	.9250	-.072	-.078	-.361
	.9750	-.060	-.042	-.235
Lower	.8100	.193	.229	.018
	.8340	.277	.416	.090
	.9250	-.012	.054	-.157
	.9750	-.006	.024	-.205

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.584	-2.108
	.0100	-4.349	-5.476
	.0250	-4.157	-5.410
	.0500	-3.343	-4.343
	.1000	-1.904	-2.428
	.2250	-1.078	-1.199
	.4500	-.693	-.596
	.7500	-.488	-.205
	.8350	-.488	-.066
	.8750	-.512	-.193
	.9250	-.404	-.157
	.9750	-.373	-.096
Lower	.0500	.970	1.000
	.1500	.663	.693
	.4000	.163	.151
	.7000	.066	.090
	.8530	-.066	-.042
	.9250	-.440	-.066
	.9750	-.289	-.169

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.717	-1.271	-.524
	.0100	-5.783	-4.681	-3.048
	.0250	-5.729	-4.886	-2.940
	.0500	-4.657	-4.614	-2.675
	.1000	-2.651	-2.289	-1.602
	.2250	-1.367	-1.193	-.904
	.4500	-.651	-.560	-.476
	.7500	-.193	-.247	-.596
Lower	.0500	.922	.898	.687
	.1500	.614	.548	.283
	.4000	.163	.211	.006
	.7000	.163	.175	-.145
Aileron				
Upper	.8050	-.223	-.241	-.663
	.8150	-.169	-.114	-.548
	.8350	-.151	-.133	-.747
	.8750	-.120	-.114	-.645
	.9250	-.084	-.084	-.428
	.9750	-.096	-.066	-.283
Lower	.8100	.205	.247	-.006
	.8340	.175	.367	.054
	.9250	-.054	.084	-.175
	.9750	-.078	.000	-.223

TABLE II.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 11.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-2.807 -5.946 -6.006 -4.807 -2.693 -1.235 -.596 -.307	-1.548 -5.036 -5.145 -4.861 -2.337 -1.217 -.627 -.283	-.928 -3.518 -3.223 -2.801 -1.657 -.922 -.470 -.651
Lower	.0500 .1500 .4000 .7000	.982 .735 .307 .127	.898 .512 .187 .211	.717 .313 .006 -.151
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.181 -.108 -.127 -.114 -.078 -.036	-.229 -.114 -.120 -.133 -.102 -.096	-.735 -.494 -.729 -.681 -.404 -.271
Lower	.8100 .8340 .9250 .9750	-.120 .175 .000 -.036	.253 .355 .036 .000	-.030 .018 -.181 -.235

TABLE II.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(i) $\alpha_f = 12.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.608	-1.675
	.0100	-4.151	-4.657
	.0250	-3.964	-4.536
	.0500	-3.223	-3.687
	.1000	-1.771	-1.952
	.2250	-.952	-.982
	.4500	-.608	-.627
	.7500	-.584	-.602
	.8350	-.506	-.072
	.8750	-.494	-.398
	.9250	-.488	-.355
	.9750	-.422	-.301
Lower	.0500	.916	.910
	.1500	.578	.542
	.4000	.193	.187
	.7000	-.000	.000
	.8530	-.133	-.078
	.9250	-.446	-.187
	.9750	-.349	-.319

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.946	-.223	-.199
	.0100	-2.645	-2.289	-2.452
	.0250	-2.211	-2.331	-2.452
	.0500	-1.428	-2.018	-2.211
	.1000	-.801	-.807	-1.295
	.2250	-.789	-.801	-.705
	.4500	-.789	-.843	-.416
	.7500	-.699	-.639	-.596
Lower	.0500	.837	.795	.602
	.1500	.488	.416	.217
	.4000	.133	.102	-.024
	.7000	.030	.084	-.163
Aileron				
Upper	.8050	-.584	-.548	-.590
	.8150	-.554	-.494	-.452
	.8350	-.590	-.524	-.693
	.8750	-.542	-.506	-.602
	.9250	-.506	-.464	-.470
	.9750	-.512	-.422	-.349
Lower	.8100	.030	.151	-.018
	.8340	.145	.331	.127
	.9250	-.283	-.090	-.151
	.9750	-.380	-.205	-.229

TABLE III

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.752
	.0100		-.059
	.0250		-.353
	.0500		-.336
	.1000		-.328
	.2250		-.366
	.4500		-.420
	.7500		-.521
	.8350		.059
	.8750		-.067
	.9250		-.059
	.9750		.046
Lower	.0500		-.622
	.1500		-.424
	.4000		-.395
	.7000		-.155
	.8530		-.080
	.9250		-.063
	.9750		.008

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.933	.807	.790
	.0100	.538	.609	.647
	.0250	-.008	.025	.176
	.0500	-.092	-.055	.071
	.1000	-.416	-.277	-.227
	.2250	-.290	-.277	-.311
	.4500	-.340	-.269	-.294
	.7500	-.223	-.445	-.345
Lower	.0500	-.798	-1.059	-.983
	.1500	-.513	-.676	-.563
	.4000	-.282	-.340	-.223
	.7000	-.008	.050	.063
Aileron				
Upper	.8050	-.345	-.466	-.332
	.8150	-.324	-.487	-.328
	.8350	-.143	-.408	-.429
	.8750	-.088	-.294	-.235
	.9250	-.013	-.101	-.067
	.9750	.013	.029	.021
Lower	.8100	.080	.164	.185
	.8340	.080	.244	.147
	.9250	.017	.084	.076
	.9750	.097	.084	.038

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.756 -.277 -.958 -.992 -.937 -.672 -.559 -.324	.966 .013 -.723 -.756 -.685 -.517 -.374 -.492	.971 .176 -.349 -.416 -.525 -.471 -.324 -.340
Lower	.0500 .1500 .4000 .7000	-.210 -.269 -.319 -.092	-.458 -.374 -.239 .080	-.492 -.366 -.193 .034
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.336 -.336 -.231 -.126 -.034 -.063	-.475 -.475 -.399 -.277 -.101 -.021	-.361 -.332 -.466 -.261 -.109 -.029
Lower	.8100 .8340 .9250 .9750	.080 .080 -.025 .004	.185 .261 .088 .055	.210 .206 .034 .017

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.013
	.0100		-3.996
	.0250		-4.315
	.0500		-3.765
	.1000		-2.227
	.2250		-1.261
	.4500		-.702
	.7500		-.206
	.8350		-.071
	.8750		-.101
	.9250		-.080
	.9750		-.004
Lower	.0500		.681
	.1500		.391
	.4000		.071
	.7000		.008
	.8530		-.084
	.9250		-.067
	.9750		-.059

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.954	.008	.290
	.0100	-3.639	-2.693	-1.794
	.0250	-4.063	-3.328	-2.013
	.0500	-3.319	-3.017	-1.899
	.1000	-2.227	-1.773	-1.273
	.2250	-1.223	-1.008	-.769
	.4500	-.672	-.576	-.412
	.7500	-.244	-.387	-.471
Lower	.0500	.693	.576	.395
	.1500	.395	.277	.097
	.4000	.097	.038	-.046
	.7000	.071	.126	-.084
Aileron				
Upper	.8050	-.273	-.353	-.462
	.8150	-.172	-.353	-.441
	.8350	-.172	-.269	-.618
	.8750	-.147	-.168	-.479
	.9250	-.139	-.130	-.269
	.9750	-.101	-.101	-.210
Lower	.8100	.122	.235	.088
	.8340	.063	.218	.126
	.9250	.008	.067	-.139
	.9750	.004	.034	-.193

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = 4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.630	-.550	-.063
	.0100	-4.786	-3.731	-2.508
	.0250	-5.084	-4.244	-2.571
	.0500	-4.134	-3.929	-2.307
	.1000	-2.525	-2.113	-1.496
	.2250	-1.239	-1.176	-.861
	.4500	-.697	-.592	-.471
	.7500	-.118	-.336	-.542
Lower	.0500	.819	.798	.567
	.1500	.563	.403	.189
	.4000	.143	.122	-.008
	.7000	-.013	.151	-.139
Aileron				
Upper	.8050	-.210	-.290	-.542
	.8150	-.176	-.286	-.529
	.8350	-.139	-.210	-.689
	.8750	-.071	-.143	-.605
	.9250	-.042	-.088	-.374
	.9750	-.034	-.063	-.235
Lower	.8100	.151	.294	.021
	.8340	.038	.340	.088
	.9250	-.004	.101	-.139
	.9750	.017	.059	-.206

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = 5.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-2.113 -5.126 -5.235 -4.353 -2.576 -1.298 -.765 -.080	-.824 -4.118 -4.534 -4.210 -2.181 -1.168 -.618 -.315	-.315 -2.807 -2.761 -2.420 -1.496 -.861 -.454 -.546
Lower	.0500 .1500 .4000 .7000	.803 .563 .202 .067	.828 .462 .160 .168	.613 .218 -.038 -.143
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.223 -.155 -.118 -.067 -.071 -.071	-.294 -.244 -.193 -.118 -.067 -.046	-.571 -.521 -.634 -.550 -.366 -.239
Lower	.8100 .8340 .9250 .9750	.147 .025 -.059 .013	.261 .349 .092 .025	.013 .063 -.143 -.214

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 6.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.739	-.718	-.298
	.0100	-4.891	-3.891	-2.693
	.0250	-5.080	-4.298	-2.664
	.0500	-4.143	-3.996	-2.328
	.1000	-2.483	-2.080	-1.471
	.2250	-1.252	-1.113	-.857
	.4500	-.685	-.567	-.433
	.7500	-.059	-.324	-.492
Lower	.0500	.870	.828	.609
	.1500	.584	.433	.223
	.4000	.193	.143	-.021
	.7000	.160	.172	-.134
Aileron				
Upper	.8050	-.151	-.252	-.508
	.8150	-.151	-.223	-.466
	.8350	-.109	-.164	-.609
	.8750	-.105	-.101	-.508
	.9250	-.038	-.038	-.340
	.9750	-.038	-.017	-.206
Lower	.8100	.122	.231	.025
	.8340	.088	.357	.088
	.9250	.008	.067	-.130
	.9750	.008	.029	-.189

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 7.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.979
	.0100		-5.420
	.0250		-5.424
	.0500		-4.433
	.1000		-2.450
	.2250		-1.311
	.4500		-.731
	.7500		-.252
	.8350		-.046
	.8750		-.092
	.9250		-.105
	.9750		-.130
Lower	.0500		.857
	.1500		.529
	.4000		.269
	.7000		.168
	.8530		-.050
	.9250		-.050
	.9750		-.080

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.471	-1.256	-.634
	.0100	-5.605	-4.718	-3.122
	.0250	-5.630	-4.950	-2.962
	.0500	-4.559	-4.555	-2.546
	.1000	-2.660	-2.261	-1.555
	.2250	-1.315	-1.185	-.878
	.4500	-.689	-.563	-.429
	.7500	-.147	-.218	-.521
Lower	.0500	.887	.866	.651
	.1500	.651	.521	.273
	.4000	.218	.176	-.008
	.7000	.063	.143	-.147
Aileron				
Upper	.8050	-.218	-.197	-.521
	.8150	-.181	-.181	-.462
	.8350	-.105	-.143	-.664
	.8750	-.042	-.105	-.597
	.9250	-.046	-.076	-.408
	.9750	.008	-.034	-.248
Lower	.8100	.202	.206	-.038
	.8340	.231	.374	.021
	.9250	-.076	.059	-.185
	.9750	-.084	.017	-.244

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 8.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.412	-1.878
	.0100	-3.769	-5.361
	.0250	-4.118	-5.256
	.0500	-3.353	-4.218
	.1000	-1.303	-2.399
	.2250	-1.298	-1.197
	.4500	-.710	-.601
	.7500	-.366	-.282
	.8350	-.353	-.126
	.8750	-.332	-.109
	.9250	-.269	-.076
	.9750	-.336	-.063
Lower	.0500	.870	.979
	.1500	.601	.605
	.4000	.244	.256
	.7000	.143	.076
	.8530	-.063	.025
	.9250	-.332	-.025
	.9750	-.206	-.088

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.361	-1.063	-.571
	.0100	-5.420	-4.433	-3.025
	.0250	-5.424	-4.765	-2.924
	.0500	-4.441	-4.420	-2.487
	.1000	-2.542	-2.235	-1.571
	.2250	-1.298	-1.202	-.874
	.4500	-.693	-.588	-.466
	.7500	-.143	-.290	-.529
Lower	.0500	.975	.840	.618
	.1500	.601	.500	.244
	.4000	.319	.193	-.034
	.7000	.063	.151	-.164
Aileron				
Upper	.8050	-.168	-.235	-.555
	.8150	-.218	-.231	-.441
	.8350	-.139	-.176	-.605
	.8750	-.080	-.134	-.534
	.9250	-.021	-.092	-.328
	.9750	-.013	-.067	-.214
Lower	.8100	.189	.197	-.008
	.8340	.261	.345	.021
	.9250	-.038	.059	-.172
	.9750	.017	.034	-.206

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(i) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:					
		0.1621	0.2895	Surface	$\frac{x}{c}$	0.5000	0.6800	0.8920
Wing				Wing				
Upper				Upper	.0000	-2.765	-1.370	-.777
					.0100	-6.210	-4.870	-3.336
					.0250	-5.870	-5.080	-3.080
					.0500	-4.622	-4.706	-2.639
					.1000	-2.689	-2.294	-1.597
					.2250	-1.399	-1.185	-.870
					.4500	-.773	-.559	-.450
					.7500	-.324	-.261	-.492
Lower				Lower	.0500	.887	.882	.685
					.1500	.655	.546	.303
					.4000	.252	.176	-.021
					.7000	.034	.155	-.189
				Aileron				
				Upper	.8050	-.218	-.185	-.542
					.8150	-.210	-.155	-.433
					.8350	-.189	-.139	-.584
					.8750	-.059	-.092	-.567
					.9250	-.084	-.080	-.357
					.9750	-.042	-.046	-.214
				Lower	.8100	.004	.218	-.029
					.8340	.034	.387	.004
					.9250	-.122	.046	-.176
					.9750	-.025	.029	-.206

TABLE III.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(j) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.433	-2.193
	.0100	-4.034	-5.689
	.0250	-3.979	-5.521
	.0500	-3.134	-4.420
	.1000	-1.836	-2.370
	.2250	-1.050	-1.151
	.4500	-.689	-.655
	.7500	-.567	-.445
	.8350	-.521	-.248
	.8750	-.437	-.210
	.9250	-.437	-.210
	.9750	-.429	-.160
Lower	.0500	.929	.979
	.1500	.613	.697
	.4000	.244	.366
	.7000	.013	.118
	.8530	-.109	-.038
	.9250	-.437	-.105
	.9750	-.307	-.185

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-3.046	-1.706	-1.071
	.0100	-6.084	-5.366	-3.769
	.0250	-6.000	-5.521	-3.370
	.0500	-4.782	-5.008	-2.866
	.1000	-2.714	-2.429	-1.676
	.2250	-1.345	-1.218	-.887
	.4500	-.626	-.567	-.445
	.7500	-.185	-.235	-.601
Lower	.0500	.954	.882	.718
	.1500	.639	.576	.332
	.4000	.248	.189	.004
	.7000	.092	.118	-.227
Aileron				
Upper	.8050	-.176	-.181	-.651
	.8150	-.168	-.160	-.282
	.8350	-.143	-.147	-.618
	.8750	-.126	-.113	-.592
	.9250	-.105	-.080	-.366
	.9750	-.084	-.067	-.231
Lower	.8100	.105	.176	-.139
	.8340	.231	.319	-.071
	.9250	-.101	.034	-.218
	.9750	-.084	.000	-.252

TABLE III.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.2 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(k) $\alpha_f = 11.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.744	-2.445
	.0100	-4.416	-5.899
	.0250	-4.261	-5.697
	.0500	-3.319	-4.517
	.1000	-1.798	-2.454
	.2250	-1.130	-1.181
	.4500	-.727	-.689
	.7500	-.634	-.525
	.8350	-.542	-.475
	.8750	-.521	-.311
	.9250	-.571	-.218
	.9750	-.567	-.185
Lower	.0500	.945	.920
	.1500	.660	.626
	.4000	.357	.248
	.7000	.113	.088
	.8530	-.197	-.063
	.9250	-.517	-.155
	.9750	-.445	-.366

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-3.227	-1.702	-.945
	.0100	-6.273	-5.374	-3.601
	.0250	-6.063	-5.450	-3.273
	.0500	-4.870	-5.008	-2.786
	.1000	-2.748	-2.378	-1.630
	.2250	-1.357	-1.185	-.870
	.4500	-.622	-.513	-.408
	.7500	-.168	-.185	-.517
Lower	.0500	.916	.929	.748
	.1500	.588	.618	.353
	.4000	.197	.239	.038
	.7000	-.013	.160	-.168
Aileron				
Upper	.8050	-.223	-.134	-.542
	.8150	-.185	-.109	-.294
	.8350	-.185	-.113	-.626
	.8750	-.126	-.084	-.618
	.9250	-.139	-.059	-.403
	.9750	-.126	-.046	-.256
Lower	.8100	.088	.189	-.130
	.8340	.147	.311	-.071
	.9250	-.172	.055	-.248
	.9750	-.134	.017	-.273

TABLE IV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.846 .467 .039 -.182 -.442 -.274 -.267 -.267	.723 .751 .182 .081 -.175 -.228 -.253 -.470	.705 .733 .305 .168 -.165 -.284 -.298 -.372
Lower	.0500 .1500 .4000 .7000	-.944 -.698 -.449 -.109	-1.274 -.782 -.389 .046	-1.140 -.632 -.235 .074
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.393 -.316 -.151 -.116 -.053 -.053	-.467 -.491 -.432 -.326 -.116 .021	-.316 -.316 -.407 -.274 -.077 .007
Lower	.8100 .8340 .9250 .9750	.070 .070 -.039 .018	.130 .200 .088 .088	.165 .182 .105 .077

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.839
	.0100		-.291
	.0250		-1.102
	.0500		-.972
	.1000		-.993
	.2250		-.582
	.4500		-.407
	.7500		-.537
	.8350		-.130
	.8750		-.049
	.9250		-.032
	.9750		.053
Lower	.0500		-.126
	.1500		-.274
	.4000		-.277
	.7000		-.091
	.8530		-.070
	.9250		-.109
	.9750		-.025

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.947	1.000	.947
	.0100	-.028	.207	.344
	.0250	-.789	-.509	-.172
	.0500	-.723	-.575	-.281
	.1000	-.821	-.575	-.442
	.2250	-.554	-.460	-.432
	.4500	-.477	-.365	-.323
	.7500	-.302	-.477	-.319
Lower	.0500	-.323	-.611	-.660
	.1500	-.372	-.495	-.421
	.4000	-.260	-.288	-.168
	.7000	.046	.074	.056
Aileron				
Upper	.8050	-.351	-.484	-.323
	.8150	-.337	-.488	-.319
	.8350	-.270	-.396	-.435
	.8750	-.165	-.274	-.256
	.9250	-.077	-.112	-.074
	.9750	-.004	.035	-.007
Lower	.8100	.070	.189	.165
	.8340	.116	.267	.211
	.9250	-.035	.074	.053
	.9750	.011	.056	.028

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.786
	.0100		-1.028
	.0250		-1.691
	.0500		-1.533
	.1000		-1.228
	.2250		-.782
	.4500		-.596
	.7500		-.516
	.8350		-.042
	.8750		-.070
	.9250		-.011
	.9750		-.004
Lower	.0500		.021
	.1500		-.077
	.4000		-.256
	.7000		-.025
	.8530		-.049
	.9250		-.046
	.9750		.018

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.849	.986	.965
	.0100	-.625	-.242	.011
	.0250	-1.344	-.986	-.502
	.0500	-1.267	-.989	-.554
	.1000	-1.102	-.786	-.607
	.2250	-.723	-.547	-.519
	.4500	-.547	-.418	-.347
	.7500	-.288	-.470	-.354
Lower	.0500	-.056	-.256	-.400
	.1500	-.235	-.298	-.330
	.4000	-.204	-.225	-.175
	.7000	.042	.077	-.000
Aileron				
Upper	.8050	-.372	-.467	-.358
	.8150	-.323	-.470	-.354
	.8350	-.193	-.389	-.484
	.8750	-.098	-.263	-.302
	.9250	-.021	-.091	-.133
	.9750	-.014	-.018	-.053
Lower	.8100	.119	.196	.154
	.8340	.119	.284	.196
	.9250	-.077	.074	.007
	.9750	.018	.035	-.000

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.309
	.0100		-1.933
	.0250		-2.540
	.0500		-2.193
	.1000		-1.632
	.2250		-1.011
	.4500		-.677
	.7500		-.526
	.8350		-.179
	.8750		-.154
	.9250		-.039
	.9750		.011
Lower	.0500		.337
	.1500		.021
	.4000		-.130
	.7000		-.130
	.8530		-.095
	.9250		-.063
	.9750		-.032

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.491	.898	.940
	.0100	-1.449	-.828	-.404
	.0250	-2.196	-1.646	-.867
	.0500	-1.986	-1.568	-.895
	.1000	-1.565	-1.056	-.818
	.2250	-.916	-.681	-.596
	.4500	-.677	-.456	-.372
	.7500	-.361	-.435	-.365
Lower	.0500	.246	.102	-.119
	.1500	-.049	-.098	-.175
	.4000	-.123	-.102	-.130
	.7000	.014	.140	-.000
Aileron				
Upper	.8050	-.375	-.407	-.323
	.8150	-.319	-.414	-.319
	.8350	-.246	-.344	-.463
	.8750	-.147	-.214	-.323
	.9250	-.077	-.091	-.154
	.9750	-.035	-.004	-.067
Lower	.8100	.105	.232	.200
	.8340	.172	.330	.193
	.9250	-.046	.077	-.011
	.9750	-.028	.063	-.042

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; $q_\infty = 17.1$ lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = 0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	 -2.407 -2.961 -2.544 -1.747 -.968 -.593 -.270	.554 -1.726 -2.586 -2.305 -1.456 -.849 -.551 -.418	.754 -1.021 -1.439 -1.400 -1.056 -.758 -.407 -.446
Lower	.0500 .1500 .4000 .7000	.442 .168 .095 .004	.323 .088 -.112 .102	.116 -.091 -.123 -.067
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.179 -.232 -.204 -.133 -.074 -.042	-.418 -.393 -.305 -.207 -.140 -.004	-.404 -.418 -.646 -.411 -.228 -.147
Lower	.8100 .8340 .9250 .9750	-.049 -.109 -.039 -.035	.214 .319 .032 .000	.102 .088 -.105 -.154

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 1.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-.530 -3.502 -3.751 -3.358 -2.021 -1.147 -.712 -.418 -.144 -.088 -.007 .070
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.768 .344 .091 .053 .025 .021 -.053

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.411 -3.119 -3.544 -3.053 -1.972 -1.102 -.684 -.228	.358 -2.179 -2.940 -2.674 -1.586 -.916 -.530 -.375	.600 -1.288 -1.632 -1.565 -1.161 -.758 -.411 -.439
Lower	.0500 .1500 .4000 .7000	.667 .291 .042 .042	.519 .186 -.018 .151	.267 .014 -.088 -.081
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.277 -.232 -.168 -.109 -.053 .011	-.347 -.337 -.277 -.172 -.081 -.018	-.396 -.386 -.596 -.453 -.267 -.151
Lower	.8100 .8340 .9250 .9750	.179 .200 -.046 .007	.239 .337 .077 .028	.116 .144 -.140 -.154

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 2.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000			-1.000	-.119	.186
	.0100			-4.379	-3.067	-2.025
	.0250			-4.372	-3.695	-2.200
	.0500			-3.628	-3.372	-2.035
	.1000			-2.288	-1.860	-1.368
	.2250			-1.214	-1.021	-.818
	.4500			-.747	-.575	-.439
	.7500			-.225	-.358	-.460
Lower	.0500			.681	.681	.449
	.1500			.288	.302	.119
	.4000			.000	.046	-.049
	.7000			-.007	.123	-.126
Aileron						
Upper	.8050			-.274	-.302	-.449
	.8150			-.175	-.298	-.414
	.8350			-.126	-.232	-.611
	.8750			-.077	-.161	-.519
	.9250			-.028	-.067	-.323
	.9750			-.007	-.035	-.204
Lower	.8100			.056	.214	.053
	.8340			.154	.368	.091
	.9250			-.011	.060	-.154
	.9750			-.056	.035	-.182

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-1.260 -4.551 -4.558 -3.839 -2.358 -1.316 -.719 -.333 -.088 -.084 -.067 -.014
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.796 .396 .105 .039 .039 -.025 -.067

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.189 -4.263 -4.460 -3.765 -2.284 -1.246 -.754 -.182	-.126 -3.154 -3.782 -3.467 -1.909 -1.049 -.568 -.333	.207 -2.032 -2.200 -2.049 -1.368 -.828 -.453 -.460
Lower	.0500 .1500 .4000 .7000	.754 .347 .123 .035	.667 .326 .077 .119	.453 .144 -.053 -.130
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.200 -.193 -.147 -.112 -.049 -.046	-.305 -.277 -.218 -.172 -.074 -.032	-.421 -.382 -.575 -.498 -.298 -.168
Lower	.8100 .8340 .9250 .9750	.098 .095 -.053 -.007	.193 .393 .046 .039	.063 .067 -.154 -.161

TABLE IV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(i) $\alpha_f = 3.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-1.435 -4.786 -4.723 -3.846 -2.323 -1.256 -.660 -.232 -.035 .000 .063 .105
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.909 .509 .232 .130 .077 .077 -.032

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.523 -4.674 -4.863 -4.000 -2.281 -1.207 -.660 -.175	-.389 -3.653 -4.204 -3.863 -2.063 -1.123 -.596 -.309	.032 -2.319 -2.439 -2.193 -1.456 -.874 -.456 -.456
Lower	.0500 .1500 .4000 .7000	.905 .498 .196 .018	.754 .382 .091 .109	.509 .186 -.046 -.147
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.172 -.196 -.095 -.060 -.014 -.004	-.253 -.242 -.218 -.151 -.042 -.004	-.460 -.414 -.589 -.537 -.358 -.196
Lower	.8100 .8340 .9250 .9750	.175 .126 -.035 -.000	.214 .375 .049 .028	.007 .063 -.200 -.189

TABLE IV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.1 lb/sq ft; belly plate off; p = 7.0 lb/sq in.; forward guy cables, lightly tightened]

(j) $\alpha_f = 3.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.828
	.0100		-5.337
	.0250		-5.158
	.0500		-4.140
	.1000		-2.516
	.2250		-1.274
	.4500		-.621
	.7500		-.189
	.8350		.028
	.8750		.028
	.9250		.028
	.9750		.081
Lower	.0500		.944
	.1500		.565
	.4000		.326
	.7000		.077
	.8530		.060
	.9250		-.007
	.9750		-.007

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.014	-.674	-.235
	.0100	-5.368	-4.102	-2.716
	.0250	-5.288	-4.547	-2.716
	.0500	-4.256	-4.239	-2.375
	.1000	-2.523	-2.165	-1.547
	.2250	-1.256	-1.154	-.895
	.4500	-.660	-.611	-.470
	.7500	-.158	-.281	-.463
Lower	.0500	.842	.800	.600
	.1500	.537	.428	.211
	.4000	.295	.133	-.042
	.7000	.053	.116	-.189
Aileron				
Upper	.8050	-.140	-.239	-.439
	.8150	-.137	-.235	-.421
	.8350	-.095	-.186	-.561
	.8750	-.039	-.126	-.565
	.9250	.021	-.063	-.379
	.9750	.042	-.028	-.232
Lower	.8100	.144	.158	-.039
	.8340	.084	.361	.063
	.9250	.039	.035	-.218
	.9750	.032	.032	-.218

TABLE V

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -6.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.919 .219 -.435 -.466 -.678 -.498 -.470 -.290	.958 .442 -.286 -.304 -.406 -.350 -.304 -.382	.922 .491 -.004 -.110 -.343 -.371 -.300 -.300
Lower	.0500 .1500 .4000 .7000	-.611 -.512 -.389 -.064	-.837 -.597 -.311 .011	-.799 -.498 -.208 .011
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.375 -.237 -.134 -.071 -.000 .046	-.389 -.396 -.350 -.261 -.088 .042	-.304 -.261 -.371 -.219 -.042 .049
Lower	.8100 .8340 .9250 .9750	-.004 .117 .018 .067	.124 .198 .074 .081	.138 .127 .057 .057

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500			.806 -.127 -.972 -.979 -.926 -.640 -.534 -.512	1.014 .088 -.650 -.707 -.629 -.477 -.371 -.396	.993 .219 -.311 -.382 -.481 -.445 -.332 -.290
Lower	.0500 .1500 .4000 .7000			-.442 -.413 -.406 -.219	-.509 -.431 -.325 -.000	-.569 -.410 -.184 .004
Aileron						
Upper	.8050 .8150 .8350 .8750 .9250 .9750			-.325 -.304 -.208 -.163 -.042 -.060	-.389 -.410 -.346 -.290 -.085 .039	-.318 -.290 -.375 -.237 -.071 .018
Lower	.8100 .8340 .9250 .9750			.078 .067 -.046 -.018	.138 .226 .081 .085	.163 .110 .021 .035

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.830 -.883 -1.527 -1.403 -1.233 -.700 -.541 -.233	.951 -.477 -1.269 -1.219 -.901 -.611 -.417 -.385	.965 -.120 -.625 -.682 -.700 -.534 -.339 -.322
Lower	.0500 .1500 .4000 .7000	.102 -.148 -.120 -.011	-.127 -.233 -.180 .081	-.290 -.279 -.134 -.021
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.226 -.198 -.092 -.049 -.018 .053	-.360 -.357 -.322 -.212 -.053 .039	-.269 -.251 -.438 -.286 -.127 -.025
Lower	.8100 .8340 .9250 .9750	.099 .155 -.011 .053	.170 .279 .071 .071	.155 .092 .004 .011

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.544 -1.512 -2.145 -1.873 -1.537 -.894 -.629 -.329	.862 -.887 -1.710 -1.618 -1.124 -.710 -.459 -.403	.929 -.495 -.968 -.996 -.883 -.622 -.371 -.325
Lower	.0500 .1500 .4000 .7000	.216 .000 -.134 -.014	.099 -.113 -.155 .067	-.078 -.184 -.138 -.039
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.283 -.251 -.159 -.148 -.071 -.046	-.375 -.367 -.329 -.194 -.064 .021	-.336 -.290 -.445 -.336 -.155 -.049
Lower	.8100 .8340 .9250 .9750	.092 .099 -.021 -.049	.170 .276 .060 .057	.134 .064 -.064 -.053

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = 1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		C_p for values of $y/\frac{b}{2}$ of:		
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000		-.127	-.011	.569	.724
	.0100		-2.982	-2.484	-1.739	-1.057
	.0250		-3.495	-3.057	-2.523	-1.445
	.0500		-3.025	-2.604	-2.304	-1.392
	.1000		-1.961	-1.827	-1.424	-1.053
	.2250		-1.067	-1.021	-.848	-.686
	.4500		-.678	-.601	-.509	-.389
	.7500		-.459	-.254	-.357	-.329
	.8350		-.127			
	.8750		-.074			
	.9250		-.078			
	.9750		.035			
Lower	.0500		.530	.537	.378	.163
	.1500		.191	.177	.085	-.046
	.4000		-.004	-.011	-.060	-.088
	.7000		-.018	.046	.078	-.071
	.8530		-.042			
	.9250		-.067			
	.9750		-.049			
Aileron						
Upper	.8050			-.230	-.329	-.300
	.8150			-.216	-.325	-.300
	.8350			-.152	-.276	-.456
	.8750			-.067	-.163	-.353
	.9250			-.007	-.057	-.216
	.9750			.071	.042	-.092
Lower	.8100			.099	.180	.120
	.8340			.106	.307	.078
	.9250			-.042	.074	-.110
	.9750			-.007	.057	-.092

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 1.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.417	-.180
	.0100	-3.152	-3.272
	.0250	-3.488	-3.654
	.0500	-2.912	-3.230
	.1000	-1.830	-1.954
	.2250	-1.081	-1.088
	.4500	-.668	-.657
	.7500	-.198	-.459
	.8350	-.085	-.106
	.8750	-.071	-.035
	.9250	-.042	.007
	.9750	.067	.071
Lower	.0500	.664	.686
	.1500	.318	.272
	.4000	.049	.028
	.7000	.025	.032
	.8530	-.035	.028
	.9250	-.184	-.007
	.9750	.000	-.035

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.131	.463	.671
	.0100	-2.813	-1.940	-1.170
	.0250	-3.265	-2.703	-1.541
	.0500	-2.792	-2.456	-1.463
	.1000	-1.887	-1.509	-1.106
	.2250	-1.032	-.866	-.703
	.4500	-.686	-.505	-.392
	.7500	-.219	-.339	-.346
Lower	.0500	.611	.459	.219
	.1500	.194	.131	-.007
	.4000	-.011	-.025	-.067
	.7000	.039	.092	-.067
Aileron				
Upper	.8050	-.244	-.293	-.307
	.8150	-.205	-.279	-.297
	.8350	-.173	-.233	-.488
	.8750	-.095	-.141	-.396
	.9250	-.004	-.049	-.230
	.9750	.000	.046	-.106
Lower	.8100	.102	.201	.110
	.8340	.113	.339	.060
	.9250	-.060	.060	-.131
	.9750	.007	.064	-.110

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 1.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.417
	.0100		-3.382
	.0250		-3.781
	.0500		-3.279
	.1000		-2.110
	.2250		-1.177
	.4500		-.788
	.7500		-.495
	.8350		-.120
	.8750		-.124
	.9250		-.074
	.9750		.004
Lower	.0500		.590
	.1500		.226
	.4000		.081
	.7000		-.011
	.8530		-.021
	.9250		-.106
	.9750		-.106

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.240	.403	.604
	.0100	-2.912	-2.074	-1.307
	.0250	-3.382	-2.845	-1.654
	.0500	-2.883	-2.583	-1.590
	.1000	-1.965	-1.569	-1.155
	.2250	-1.113	-.905	-.756
	.4500	-.689	-.534	-.420
	.7500	-.247	-.346	-.367
Lower	.0500	.516	.466	.254
	.1500	.187	.138	-.007
	.4000	-.042	-.042	-.099
	.7000	-.028	.085	-.088
Aileron				
Upper	.8050	-.261	-.314	-.325
	.8150	-.201	-.314	-.325
	.8350	-.131	-.247	-.516
	.8750	-.117	-.170	-.438
	.9250	-.025	-.064	-.265
	.9750	-.053	.028	-.138
Lower	.8100	.067	.187	.110
	.8340	.106	.314	.060
	.9250	-.078	.049	-.134
	.9750	-.000	.046	-.134

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 2.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.777	-.601
	.0100	-3.615	-3.876
	.0250	-3.876	-4.106
	.0500	-3.191	-3.643
	.1000	-1.972	-2.113
	.2250	-1.240	-1.141
	.4500	-.647	-.664
	.7500	-.265	-.392
	.8350	-.071	-.095
	.8750	-.049	-.039
	.9250	-.014	.018
	.9750	.035	.049
Lower	.0500	.774	.802
	.1500	.357	.343
	.4000	.092	.124
	.7000	.064	.060
	.8530	-.046	.049
	.9250	-.134	-.011
	.9750	-.039	-.042

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.544	.177	.452
	.0100	-3.431	-2.608	-1.615
	.0250	-3.820	-3.307	-1.908
	.0500	-3.254	-3.004	-1.777
	.1000	-2.071	-1.710	-1.251
	.2250	-1.092	-.965	-.792
	.4500	-.700	-.530	-.410
	.7500	-.180	-.297	-.364
Lower	.0500	.731	.587	.360
	.1500	.329	.223	.057
	.4000	.067	.014	-.064
	.7000	.078	.088	-.117
Aileron				
Upper	.8050	-.173	-.261	-.332
	.8150	-.166	-.247	-.272
	.8350	-.117	-.198	-.541
	.8750	-.057	-.120	-.449
	.9250	.000	-.025	-.318
	.9750	.042	.035	-.166
Lower	.8100	.131	.205	.074
	.8340	.092	.367	.042
	.9250	-.035	.042	-.170
	.9750	.032	.035	-.141

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.636
	.0100		-3.943
	.0250		-4.339
	.0500		-3.753
	.1000		-2.152
	.2250		-1.187
	.4500		-.693
	.7500		-.353
	.8350		-.039
	.8750		.014
	.9250		.007
	.9750		.074
Lower	.0500		.827
	.1500		.392
	.4000		.170
	.7000		.092
	.8530		.032
	.9250		.057
	.9750		.011

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.777	.081	.382
	.0100	-3.689	-2.792	-1.767
	.0250	-3.763	-3.473	-2.035
	.0500	-3.459	-3.152	-1.898
	.1000	-2.177	-1.784	-1.307
	.2250	-1.173	-1.007	-.816
	.4500	-.703	-.555	-.417
	.7500	-.184	-.307	-.378
Lower	.0500	.777	.622	.382
	.1500	.516	.254	.088
	.4000	.095	.032	-.064
	.7000	.053	.088	-.127
Aileron				
Upper	.8050	-.198	-.251	-.339
	.8150	-.170	-.233	-.297
	.8350	-.113	-.191	-.534
	.8750	-.064	-.120	-.456
	.9250	-.018	-.025	-.300
	.9750	.021	.039	-.152
Lower	.8100	.120	.191	.078
	.8340	.102	.378	.057
	.9250	-.021	.039	-.170
	.9750	-.000	.049	-.145

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 3.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:					
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000			.011	.053
	.0100			-.011	.046
	.0250	.000		-.000	.014
	.0500	-.951		-.184	.219
	.1000	-4.074		-3.307	-2.117
	.2250	-4.166		-3.908	-2.297
	.4500	-3.728		-3.569	-2.120
	.7500	-2.290		-1.940	-1.413
Lower	.0500	-1.155		-1.060	-.855
	.1500	-.710		-.551	-.431
	.4000	-.240		-.276	-.353
	.7000	.763		.721	.473
Aileron						
Upper	.8050	.378		.311	.148
	.8150	.134		.057	-.071
	.8350	.088		.085	-.145
	.8750	-.173		-.216	-.336
	.9250	-.208		-.201	-.300
	.9750	-.159		-.166	-.537
Lower	.8100	-.074		-.106	-.495
	.8340	.007		-.018	-.346
	.9250	-.039		.046	-.180
	.9750	.018		.205	.025

TABLE V.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) $\alpha_f = 4.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500		-.689 -4.152 -4.608 -4.265 -2.180 -1.155 -.580 -.240	-.194 -2.731 -2.735 -2.435 -1.537 -.890 -.456 -.385
Lower	.0500 .1500 .4000 .7000		.806 .378 .102 .102	.622 .201 -.032 -.205
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750		-.194 -.180 -.117 -.067 -.025 .057	-.357 -.371 -.519 -.541 -.424 -.269
Lower	.8100 .8340 .9250 .9750		.159 .346 .021 .039	-.039 -.028 -.205 -.028

TABLE V.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 1-3-6-8; q_∞ = 17.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(*l*) $\alpha_f = 4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.011	-1.922
	.0100	-5.187	-5.827
	.0250	-5.099	-5.572
	.0500	-4.057	-4.435
	.1000	-2.495	-2.608
	.2250	-1.396	-1.385
	.4500	-.749	-.721
	.7500	-.230	-.244
	.8350	-.032	-.110
	.8750	-.067	-.067
	.9250	-.046	-.078
	.9750	-.042	-.032
Lower	.0500	.788	.919
	.1500	.541	.523
	.4000	.261	.201
	.7000	.166	.110
	.8530	.011	.057
	.9250	-.184	-.039
	.9750	-.110	-.102

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.102	-.700	-.187
	.0100	-5.431	-4.180	-2.693
	.0250	-5.325	-4.618	-2.686
	.0500	-4.403	-4.265	-2.403
	.1000	-2.608	-2.170	-1.534
	.2250	-1.322	-1.127	-.890
	.4500	-.724	-.544	-.449
	.7500	-.166	-.184	-.385
Lower	.0500	.795	.837	.604
	.1500	.516	.459	.244
	.4000	.138	.131	-.021
	.7000	.071	.110	-.177
Aileron				
Upper	.8050	-.124	-.148	-.343
	.8150	-.120	-.127	-.339
	.8350	-.110	-.113	-.452
	.8750	-.074	-.071	-.502
	.9250	-.025	-.007	-.378
	.9750	.018	.039	-.219
Lower	.8100	.085	.191	-.014
	.8340	.088	.367	.014
	.9250	-.053	.049	-.201
	.9750	.000	.057	-.170

TABLE VI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.974	.939
	.0100	-.122	-.052
	.0250	-.748	-.661
	.0500	-.748	-.696
	.1000	-.696	-.687
	.2250	-.565	-.557
	.4500	-.348	-.400
	.7500	-.296	-.539
	.8350	-.087	-.113
	.8750	-.070	-.087
	.9250	-.139	-.052
	.9750	-.096	-.000
Lower	.0500	-.487	-.348
	.1500	-.365	-.435
	.4000	-.322	-.313
	.7000	-.174	-.122
	.8530	-.104	-.087
	.9250	-.096	-.043
	.9750	-.096	-.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.965	1.035	.939
	.0100	.087	.226	.313
	.0250	-.548	-.391	-.209
	.0500	-.539	-.461	-.287
	.1000	-.635	-.487	-.409
	.2250	-.513	-.374	-.383
	.4500	-.487	-.278	-.270
	.7500	-.261	-.243	-.078
Lower	.0500	-.452	-.539	-.557
	.1500	-.391	-.417	-.400
	.4000	-.357	-.270	-.226
	.7000	-.104	-.052	-.052
Aileron				
Upper	.8050	-.243	-.139	.017
	.8150	-.226	-.139	.104
	.8350	-.139	-.191	-.191
	.8750	-.096	-.165	-.157
	.9250	.000	-.078	-.070
	.9750	.035	.035	.000
Lower	.8100	-.009	.122	-.035
	.8340	.035	.122	-.009
	.9250	-.052	.026	.035
	.9750	.052	.113	.017

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.843	.800
	.0100	-.878	-.913
	.0250	-1.513	-1.583
	.0500	-1.400	-1.461
	.1000	-1.061	-1.139
	.2250	-.713	-.713
	.4500	-.383	-.443
	.7500	-.261	-.487
	.8350	-.052	-.043
	.8750	-.009	-.017
	.9250	-.009	.009
	.9750	.052	.148
Lower	.0500	.035	.148
	.1500	-.061	-.096
	.4000	-.165	-.200
	.7000	-.035	-.052
	.8530	-.052	.009
	.9250	-.052	-.017
	.9750	.017	.043

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.835	.922	.913
	.0100	-.765	-.513	-.278
	.0250	-1.443	-1.226	-.791
	.0500	-1.322	-1.209	-.774
	.1000	-1.096	-.843	-.643
	.2250	-.678	-.583	-.487
	.4500	-.478	-.383	-.270
	.7500	-.252	-.226	-.078
Lower	.0500	.070	-.061	-.191
	.1500	-.148	-.183	-.270
	.4000	-.226	-.157	-.183
	.7000	-.026	-.009	-.096
Aileron				
Upper	.8050	-.226	-.157	-.035
	.8150	-.174	-.139	-.017
	.8350	-.104	-.165	-.096
	.8750	-.017	-.139	-.148
	.9250	.035	-.017	-.052
	.9750	.035	.113	-.052
Lower	.8100	-.035	.200	-.061
	.8340	.122	.183	-.078
	.9250	-.043	.043	-.078
	.9750	.043	.078	-.017

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.313	.278
	.0100	-1.965	-2.087
	.0250	-2.400	-2.661
	.0500	-2.122	-2.374
	.1000	-1.417	-1.548
	.2250	-.922	-.939
	.4500	-.487	-.539
	.7500	-.278	-.435
	.8350	-.070	-.087
	.8750	-.052	-.035
	.9250	-.035	.026
	.9750	.000	.070
Lower	.0500	.461	.530
	.1500	.157	.122
	.4000	-.043	-.070
	.7000	.000	.009
	.8530	-.052	-.017
	.9250	-.130	.000
	.9750	-.035	.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.278	.539	.678
	.0100	-1.896	-1.548	-1.026
	.0250	-2.374	-2.252	-1.383
	.0500	-2.139	-2.000	-1.339
	.1000	-1.591	-1.296	-.974
	.2250	-.922	-.783	-.609
	.4500	-.583	-.435	-.313
	.7500	-.200	-.270	-.104
Lower	.0500	.452	.365	.148
	.1500	.122	.052	-.070
	.4000	-.087	-.122	-.104
	.7000	.017	-.017	-.113
Aileron				
Upper	.8050	-.191	-.157	-.035
	.8150	-.157	-.130	-.017
	.8350	-.113	-.157	-.148
	.8750	-.043	-.122	-.165
	.9250	.000	-.026	-.087
	.9750	.035	.070	-.035
Lower	.8100	.087	.139	-.087
	.8340	.191	.209	-.139
	.9250	.000	.009	-.070
	.9750	.035	.035	-.017

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = 2.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.574
	.0100		-3.713
	.0250		-3.904
	.0500		-3.496
	.1000		-2.043
	.2250		-1.252
	.4500		-.652
	.7500		-.304
	.8350		-.096
	.8750		-.122
	.9250		-.043
	.9750		-.043
Lower	.0500		.600
	.1500		.148
	.4000		-.043
	.7000		-.070
	.8530		-.009
	.9250		-.009
	.9750		-.061

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.478	-.026	.252
	.0100	-3.226	-2.626	-1.852
	.0250	-3.478	-3.217	-2.087
	.0500	-3.096	-2.913	-1.922
	.1000	-2.017	-1.678	-1.287
	.2250	-1.070	-.913	-.765
	.4500	-.583	-.548	-.391
	.7500	-.348	-.313	-.235
Lower	.0500	.617	.530	.443
	.1500	.304	.217	.078
	.4000	-.009	-.017	-.087
	.7000	-.017	-.009	-.165
Aileron				
Upper	.8050	-.252	-.165	-.157
	.8150	-.270	-.165	-.087
	.8350	-.157	-.183	-.209
	.8750	-.113	-.148	-.261
	.9250	-.078	-.078	-.200
	.9750	-.078	-.017	-.148
Lower	.8100	.061	.035	-.104
	.8340	.130	.165	-.087
	.9250	-.043	.009	-.165
	.9750	-.043	.009	-.096

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.983	-1.722
	.0100	-4.965	-5.322
	.0250	-4.713	-5.287
	.0500	-3.957	-4.557
	.1000	-2.296	-2.443
	.2250	-1.304	-1.304
	.4500	-.574	-.643
	.7500	-.209	-.191
	.8350	-.087	-.096
	.8750	-.070	-.070
	.9250	-.035	-.043
	.9750	.000	-.017
Lower	.0500	.904	.930
	.1500	.600	.557
	.4000	.174	.209
	.7000	.096	.113
	.8530	.078	.017
	.9250	-.113	-.026
	.9750	-.035	-.096

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.835	-.809	-.209
	.0100	-4.852	-3.974	-2.643
	.0250	-4.965	-4.243	-2.670
	.0500	-4.174	-3.983	-2.426
	.1000	-2.443	-2.009	-1.400
	.2250	-1.296	-1.096	-.783
	.4500	-.661	-.487	-.391
	.7500	-.174	-.139	-.357
Lower	.0500	.870	.870	.661
	.1500	.496	.478	.261
	.4000	.165	.165	.009
	.7000	.113	.113	-.157
Aileron				
Upper	.8050	-.165	-.096	-.252
	.8150	-.139	-.070	.070
	.8350	-.104	-.078	-.209
	.8750	-.061	-.017	-.365
	.9250	-.043	.026	-.270
	.9750	-.035	.035	-.183
Lower	.8100	.157	.183	-.217
	.8340	.278	.243	.026
	.9250	-.009	.078	-.148
	.9750	-.017	.078	-.104

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 8.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.383	-1.496
	.0100	-3.983	-4.904
	.0250	-3.965	-4.922
	.0500	-3.417	-4.252
	.1000	-1.896	-2.339
	.2250	-1.043	-1.217
	.4500	-.600	-.661
	.7500	-.496	-.313
	.8350	-.417	-.200
	.8750	-.383	-.157
	.9250	-.339	-.130
	.9750	-.270	-.070
Lower	.0500	.826	.887
	.1500	.504	.504
	.4000	.104	.157
	.7000	.009	.070
	.8530	.070	-.052
	.9250	-.322	-.104
	.9750	-.278	-.139

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.043	-1.183	-.539
	.0100	-5.061	-4.443	-3.122
	.0250	-5.200	-4.661	-3.009
	.0500	-4.261	-4.357	-2.704
	.1000	-2.496	-2.139	-1.539
	.2250	-1.261	-1.148	-.843
	.4500	-.678	-.539	-.426
	.7500	-.226	-.191	-.530
Lower	.0500	.843	.878	.652
	.1500	.522	.496	.226
	.4000	.148	.139	-.017
	.7000	.104	.078	-.235
Aileron				
Upper	.8050	-.183	-.148	-.357
	.8150	-.174	-.139	.165
	.8350	-.122	-.130	-.365
	.8750	-.096	-.061	-.496
	.9250	-.026	-.035	-.383
	.9750	-.070	.009	-.243
Lower	.8100	.122	.130	-.330
	.8340	.183	.235	.017
	.9250	-.052	.017	-.226
	.9750	-.035	.026	-.191

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.530	-1.765
	.0100	-4.157	-5.235
	.0250	-3.991	-5.139
	.0500	-3.357	-4.296
	.1000	-1.835	-2.313
	.2250	-.957	-1.148
	.4500	-.591	-.557
	.7500	-.513	-.296
	.8350	-.426	-.183
	.8750	-.374	-.139
	.9250	-.374	-.122
	.9750	-.348	-.104
Lower	.0500	.930	.957
	.1500	.617	.591
	.4000	.191	.226
	.7000	.070	.104
	.8530	.052	-.026
	.9250	-.365	-.070
	.9750	-.313	-.139

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.313	-1.217	-.487
	.0100	-5.322	-4.496	-3.052
	.0250	-5.270	-4.678	-2.930
	.0500	-4.417	-4.374	-2.704
	.1000	-2.487	-2.148	-1.522
	.2250	-1.252	-1.122	-.843
	.4500	-.626	-.539	-.400
	.7500	-.174	-.191	-.452
Lower	.0500	.922	.913	.661
	.1500	.583	.548	.304
	.4000	.226	.165	.035
	.7000	.157	.104	-.183
Aileron				
Upper	.8050	-.148	-.104	-.322
	.8150	-.139	-.104	.130
	.8350	-.122	-.096	-.287
	.8750	-.070	-.043	-.443
	.9250	-.052	-.026	-.348
	.9750	-.035	.035	-.252
Lower	.8100	.148	.148	-.278
	.8340	.235	.226	.035
	.9250	.017	.026	-.165
	.9750	-.017	.061	-.139

TABLE VI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.687	-2.017
	.0100	-4.270	-5.461
	.0250	-4.087	-5.330
	.0500	-3.357	-4.383
	.1000	-1.826	-2.348
	.2250	-1.009	-1.148
	.4500	-.626	-.591
	.7500	-.522	-.339
	.8350	-.452	-.226
	.8750	-.435	-.191
	.9250	-.426	-.165
	.9750	-.417	-.122
Lower	.0500	.913	.948
	.1500	.609	.591
	.4000	.217	.261
	.7000	.052	.078
	.8530	.035	-.052
	.9250	-.426	-.113
	.9750	-.348	-.183

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.722	-1.565	-.739
	.0100	-5.748	-5.000	-3.435
	.0250	-5.704	-5.026	-3.261
	.0500	-4.617	-4.809	-2.843
	.1000	-2.574	-2.278	-1.635
	.2250	-1.270	-1.157	-.861
	.4500	-.609	-.530	-.426
	.7500	-.148	-.183	-.565
Lower	.0500	.904	.922	.722
	.1500	.600	.557	.322
	.4000	.252	.200	-.009
	.7000	.148	.113	-.217
Aileron				
Upper	.8050	-.157	-.165	-.374
	.8150	-.139	-.139	.217
	.8350	-.122	-.130	-.339
	.8750	-.096	-.078	-.513
	.9250	-.070	-.052	-.409
	.9750	-.078	.009	-.243
Lower	.8100	.157	.157	-.365
	.8340	.243	.235	.026
	.9250	-.052	.026	-.235
	.9750	-.043	.026	-.217

TABLE VI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 11.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.739	-2.278
	.0100	-4.365	-5.730
	.0250	-4.070	-5.548
	.0500	-3.574	-4.539
	.1000	-2.043	-2.287
	.2250	-1.278	-1.183
	.4500	-.904	-.713
	.7500	-.470	-.435
	.8350	-.522	-.383
	.8750	-.643	-.296
	.9250	-.530	-.270
	.9750	-.530	-.096
Lower	.0500	.617	.800
	.1500	.470	.626
	.4000	.087	.183
	.7000	-.122	.122
	.8530	-.026	-.096
	.9250	-.504	-.104
	.9750	-.383	-.104

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-3.043	-1.965	-1.122
	.0100	-6.139	-5.322	-3.774
	.0250	-5.922	-5.313	-3.443
	.0500	-4.817	-5.087	-2.965
	.1000	-2.609	-2.330	-1.704
	.2250	-1.348	-1.165	-.904
	.4500	-.617	-.557	-.470
	.7500	-.226	-.200	-.609
Lower	.0500	.870	.896	.722
	.1500	.504	.565	.348
	.4000	.226	.209	-.009
	.7000	.113	.078	-.243
Aileron				
Upper	.8050	-.130	-.174	-.374
	.8150	-.139	-.122	.157
	.8350	-.139	-.113	-.383
	.8750	-.087	-.078	-.557
	.9250	-.104	-.052	-.443
	.9750	-.157	-.035	-.278
Lower	.8100	.035	.104	-.383
	.8340	.235	.217	-.009
	.9250	-.052	-.017	-.226
	.9750	-.017	-.017	-.226

TABLE VII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	1.018	.982
	.0100	-.095	.107
	.0250	-.710	-.580
	.0500	-.698	-.592
	.1000	-.663	-.669
	.2250	-.538	-.497
	.4500	-.325	-.385
	.7500	-.272	-.538
	.8350	-.047	-.041
	.8750	-.071	-.036
	.9250	-.124	-.018
	.9750	-.083	.024
Lower	.0500	-.485	-.391
	.1500	-.355	-.438
	.4000	-.284	-.308
	.7000	-.160	-.101
	.8530	-.089	-.053
	.9250	-.071	-.024
	.9750	-.047	.041

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.935	.970	.929
	.0100	.260	.349	.361
	.0250	-.391	-.314	-.154
	.0500	-.450	-.373	-.237
	.1000	-.592	-.462	-.391
	.2250	-.473	-.343	-.379
	.4500	-.450	-.260	-.254
	.7500	-.219	-.189	-.071
Lower	.0500	-.556	-.657	-.633
	.1500	-.473	-.485	-.444
	.4000	-.373	-.296	-.213
	.7000	-.089	-.059	-.047
Aileron				
Upper	.8050	-.249	-.154	.041
	.8150	-.237	-.142	.154
	.8350	-.166	-.178	-.219
	.8750	-.089	-.154	-.130
	.9250	-.030	-.071	-.041
	.9750	.018	.047	.036
Lower	.8100	-.036	.071	.000
	.8340	.065	.071	-.024
	.9250	-.053	.024	-.006
	.9750	.041	.071	.047

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.686	.734
	.0100	-1.065	-1.041
	.0250	-1.615	-1.669
	.0500	-1.473	-1.538
	.1000	-1.124	-1.183
	.2250	-.775	-.716
	.4500	-.432	-.473
	.7500	-.314	-.515
	.8350	-.071	-.095
	.8750	-.071	-.071
	.9250	-.077	-.030
	.9750	-.006	.047
Lower	.0500	.018	.148
	.1500	-.083	-.148
	.4000	-.201	-.172
	.7000	-.083	-.083
	.8530	-.101	-.065
	.9250	-.112	-.053
	.9750	-.024	.000

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.781	.882	.923
	.0100	-.805	-.592	-.278
	.0250	-1.450	-1.325	-.746
	.0500	-1.385	-1.225	-.763
	.1000	-1.178	-.893	-.669
	.2250	-.710	-.598	-.521
	.4500	-.544	-.379	-.302
	.7500	-.260	-.237	-.095
Lower	.0500	.041	-.053	-.183
	.1500	-.130	-.172	-.249
	.4000	-.207	-.183	-.166
	.7000	-.036	-.053	-.071
Aileron				
Upper	.8050	-.237	-.160	-.000
	.8150	-.213	-.178	.012
	.8350	-.178	-.183	-.112
	.8750	-.071	-.142	-.136
	.9250	-.036	-.053	-.065
	.9750	.030	.036	-.024
Lower	.8100	.024	.089	-.041
	.8340	.112	.112	-.065
	.9250	-.012	.012	-.095
	.9750	.030	.065	-.024

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.124	.183
	.0100	-2.178	-2.314
	.0250	-2.574	-2.876
	.0500	-2.290	-2.550
	.1000	-1.497	-1.627
	.2250	-.929	-.905
	.4500	-.503	-.568
	.7500	-.243	-.414
	.8350	-.065	-.077
	.8750	-.036	-.030
	.9250	-.012	.030
	.9750	.041	.077
Lower	.0500	.479	.592
	.1500	.207	.142
	.4000	-.030	-.006
	.7000	.024	.024
	.8530	.024	.006
	.9250	-.112	.006
	.9750	-.024	.006

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.207	.485	.645
	.0100	-2.101	-1.663	-1.124
	.0250	-2.598	-2.385	-1.467
	.0500	-2.325	-2.178	-1.408
	.1000	-1.651	-1.367	-1.030
	.2250	-.929	-.799	-.651
	.4500	-.592	-.450	-.325
	.7500	-.219	-.260	-.148
Lower	.0500	.467	.355	.195
	.1500	.154	.071	-.041
	.4000	-.036	-.089	-.107
	.7000	.041	-.018	-.118
Aileron				
Upper	.8050	-.189	-.178	-.071
	.8150	-.201	-.166	-.036
	.8350	-.130	-.172	-.148
	.8750	-.065	-.130	-.201
	.9250	-.024	-.047	-.124
	.9750	.041	.036	-.083
Lower	.8100	.083	.071	-.107
	.8340	.183	.130	-.148
	.9250	-.006	.024	-.118
	.9750	.047	.047	-.071

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.8350 .8750 .9250 .9750		-.077 -.053 -.006 .012
Lower	.8530 .9250 .9750		-.000 -.036 -.053

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.811 -3.663 -3.911 -3.420 -2.124 -1.154 -.645 -.201	-.195 -3.024 -3.509 -3.201 -1.775 -.970 -.503 -.225	.172 -2.059 -2.213 -2.036 -1.320 -.793 -.391 -.254
Lower	.0500 .1500 .4000 .7000	.746 .367 .053 .077	.680 .284 .065 .047	.450 .130 -.065 -.148
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.178 -.166 -.130 -.101 -.036 .006	-.142 -.148 -.142 -.089 -.024 .012	-.189 -.142 -.219 -.308 -.237 -.189
Lower	.8100 .8340 .9250 .9750	.112 .148 -.024 .006	.071 .166 .030 .036	-.160 -.041 -.178 -.142

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.136	-1.947
	.0100	-5.142	-5.686
	.0250	-4.876	-5.574
	.0500	-3.988	-4.556
	.1000	-2.325	-2.462
	.2250	-1.284	-1.260
	.4500	-.550	-.609
	.7500	-.172	-.107
	.8350	-.041	-.077
	.8750	-.018	-.065
	.9250	.000	-.036
	.9750	.030	-.018
Lower	.0500	.976	.953
	.1500	.657	.598
	.4000	.278	.266
	.7000	.178	.166
	.8530	.142	.059
	.9250	-.089	.018
	.9750	-.024	-.053

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.047	-.964	-.402
	.0100	-5.189	-4.225	-2.905
	.0250	-5.154	-4.479	-2.864
	.0500	-4.349	-4.231	-2.533
	.1000	-2.515	-2.130	-1.515
	.2250	-1.260	-1.107	-.846
	.4500	-.615	-.544	-.414
	.7500	-.148	-.219	-.420
Lower	.0500	.911	.840	.651
	.1500	.574	.467	.249
	.4000	.219	.154	-.024
	.7000	.148	.083	-.201
Aileron				
Upper	.8050	-.112	-.160	-.284
	.8150	-.107	-.148	.124
	.8350	-.101	-.142	-.302
	.8750	-.065	-.107	-.462
	.9250	-.030	-.059	-.367
	.9750	-.024	.012	-.249
Lower	.8100	.160	.089	-.314
	.8340	.249	.189	.041
	.9250	-.012	-.018	-.237
	.9750	.012	.012	-.172

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^\circ$; cable configuration 2-5-8; $q_\infty = 10.1$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 7.9^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.391	-1.604
	.0100	-4.000	-5.065
	.0250	-3.899	-5.053
	.0500	-3.266	-4.160
	.1000	-1.846	-2.249
	.2250	-1.047	-1.148
	.4500	-.615	-.604
	.7500	-.485	-.284
	.8350	-.426	-.178
	.8750	-.391	-.142
	.9250	-.367	-.112
	.9750	-.308	-.077
Lower	.0500	.893	.911
	.1500	.515	.527
	.4000	.166	.213
	.7000	.041	.077
	.8530	.036	-.065
	.9250	-.373	-.107
	.9750	-.290	-.148

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.077	-1.077	-.456
	.0100	-5.112	-4.367	-2.964
	.0250	-5.213	-4.580	-2.858
	.0500	-4.249	-4.325	-2.544
	.1000	-2.485	-2.130	-1.515
	.2250	-1.266	-1.118	-.893
	.4500	-.663	-.533	-.438
	.7500	-.189	-.195	-.438
Lower	.0500	.876	.852	.657
	.1500	.544	.479	.231
	.4000	.172	.136	-.041
	.7000	.118	.071	-.225
Aileron				
Upper	.8050	-.178	-.142	-.320
	.8150	-.154	-.130	.047
	.8350	-.142	-.112	-.278
	.8750	-.095	-.071	-.426
	.9250	-.071	-.030	-.331
	.9750	-.059	.018	-.213
Lower	.8100	.142	.118	-.290
	.8340	.154	.189	.065
	.9250	-.089	-.012	-.237
	.9750	-.024	.012	-.172

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 8.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.414	-1.757
	.0100	-3.959	-5.213
	.0250	-3.787	-5.154
	.0500	-3.172	-4.260
	.1000	-1.757	-2.296
	.2250	-1.024	-1.154
	.4500	-.657	-.604
	.7500	-.491	-.379
	.8350	-.485	-.260
	.8750	-.420	-.178
	.9250	-.426	-.166
	.9750	-.402	-.130
Lower	.0500	.905	.923
	.1500	.544	.544
	.4000	.201	.225
	.7000	.018	.095
	.8530	-.059	-.071
	.9250	-.420	-.118
	.9750	-.343	-.178

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.343	-1.367	-.675
	.0100	-5.485	-4.751	-3.272
	.0250	-5.556	-4.888	-3.071
	.0500	-4.509	-4.633	-2.651
	.1000	-2.604	-2.201	-1.556
	.2250	-1.284	-1.112	-.846
	.4500	-.633	-.491	-.414
	.7500	-.160	-.160	-.438
Lower	.0500	.899	.935	.692
	.1500	.580	.574	.296
	.4000	.189	.195	-.012
	.7000	.101	.118	-.213
Aileron				
Upper	.8050	-.189	-.112	-.320
	.8150	-.183	-.101	.166
	.8350	-.154	-.089	-.314
	.8750	-.130	-.059	-.444
	.9250	-.071	-.006	-.349
	.9750	-.065	.030	-.254
Lower	.8100	.112	.136	-.325
	.8340	.178	.207	.059
	.9250	-.059	.006	-.237
	.9750	-.047	.018	-.172

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.633	-1.953
	.0100	-4.225	-5.485
	.0250	-4.030	-5.314
	.0500	-3.278	-4.343
	.1000	-1.822	-2.308
	.2250	-.988	-1.160
	.4500	-.651	-.633
	.7500	-.544	-.414
	.8350	-.533	-.243
	.8750	-.497	-.213
	.9250	-.473	-.178
	.9750	-.438	-.154
Lower	.0500	.876	.911
	.1500	.586	.568
	.4000	.201	.213
	.7000	.047	.071
	.8530	-.130	-.065
	.9250	-.462	-.112
	.9750	-.367	-.189

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.556	-1.509	-.822
	.0100	-5.651	-4.917	-3.432
	.0250	-5.657	-5.053	-3.160
	.0500	-4.592	-4.728	-2.728
	.1000	-2.639	-2.272	-1.592
	.2250	-1.302	-1.172	-.876
	.4500	-.663	-.544	-.414
	.7500	-.213	-.219	-.485
Lower	.0500	.870	.888	.692
	.1500	.574	.538	.296
	.4000	.207	.148	-.006
	.7000	.095	.071	-.272
Aileron				
Upper	.8050	-.195	-.154	-.343
	.8150	-.213	-.136	.148
	.8350	-.166	-.124	-.343
	.8750	-.124	-.101	-.491
	.9250	-.118	-.047	-.385
	.9750	-.101	.012	-.231
Lower	.8100	.118	.101	-.361
	.8340	.154	.201	.018
	.9250	-.071	.000	-.278
	.9750	-.083	.000	-.219

TABLE VII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.846	-2.260
	.0100	-4.402	-5.769
	.0250	-4.178	-5.562
	.0500	-3.349	-4.462
	.1000	-1.799	-2.325
	.2250	-.947	-1.124
	.4500	-.627	-.604
	.7500	-.533	-.379
	.8350	-.544	-.237
	.8750	-.521	-.237
	.9250	-.467	-.213
	.9750	-.456	-.172
Lower	.0500	.935	.899
	.1500	.604	.621
	.4000	.243	.266
	.7000	.053	.083
	.8530	-.089	-.077
	.9250	-.473	-.118
	.9750	-.367	-.178

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.970	-1.704	-.846
	.0100	-6.012	-5.130	-3.426
	.0250	-5.870	-5.160	-3.178
	.0500	-4.728	-4.899	-2.740
	.1000	-2.675	-2.314	-1.633
	.2250	-1.308	-1.213	-.893
	.4500	-.609	-.538	-.444
	.7500	-.207	-.189	-.550
Lower	.0500	.870	.876	.663
	.1500	.639	.568	.272
	.4000	.219	.201	-.000
	.7000	.107	.101	-.260
Aileron				
Upper	.8050	-.189	-.142	-.373
	.8150	-.166	-.148	.136
	.8350	-.166	-.124	-.355
	.8750	-.148	-.101	-.497
	.9250	-.124	-.053	-.385
	.9750	-.124	-.030	-.266
Lower	.8100	.112	.101	-.355
	.8340	.178	.213	-.012
	.9250	-.089	-.018	-.249
	.9750	-.118	.000	-.213

TABLE VII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 11.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.314	-1.408
	.0100	-3.882	-4.414
	.0250	-3.704	-4.314
	.0500	-2.994	-3.479
	.1000	-1.639	-1.905
	.2250	-.728	-.947
	.4500	-.568	-.556
	.7500	-.633	-.627
	.8350	-.373	-.379
	.8750	-.426	-.396
	.9250	-.414	-.314
	.9750	-.343	-.278
Lower	.0500	.793	.905
	.1500	.444	.533
	.4000	.201	.142
	.7000	.036	.065
	.8530	-.059	-.065
	.9250	-.491	-.231
	.9750	-.402	-.302

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.805	-.160	-.237
	.0100	-2.491	-2.166	-2.485
	.0250	-2.142	-2.254	-2.444
	.0500	-1.473	-1.994	-2.166
	.1000	-.698	-.740	-1.254
	.2250	-.627	-.663	-.651
	.4500	-.746	-.716	-.379
	.7500	-.787	-.633	-.467
Lower	.0500	.846	.746	.568
	.1500	.479	.391	.189
	.4000	.124	.059	-.077
	.7000	-.012	-.012	-.249
Aileron				
Upper	.8050	-.509	-.544	-.420
	.8150	-.568	-.521	-.272
	.8350	-.609	-.574	-.414
	.8750	-.615	-.538	-.479
	.9250	-.521	-.491	-.391
	.9750	-.533	-.462	-.290
Lower	.8100	.047	-.006	-.231
	.8340	.107	.083	.000
	.9250	-.290	-.243	-.243
	.9750	-.391	-.308	-.219

TABLE VIII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	1.021	.987
	.0100	-.174	.009
	.0250	-.817	-.706
	.0500	-.804	-.706
	.1000	-.766	-.762
	.2250	-.587	-.549
	.4500	-.362	-.400
	.7500	-.281	-.562
	.8350	-.055	-.085
	.8750	-.089	-.055
	.9250	-.089	-.043
	.9750	-.085	.038
Lower	.0500	-.481	-.366
	.1500	-.366	-.443
	.4000	-.298	-.315
	.7000	-.170	-.106
	.8530	-.085	-.068
	.9250	-.077	-.034
	.9750	-.047	.030

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.970	.974	.928
	.0100	.213	.328	.357
	.0250	-.477	-.374	-.128
	.0500	-.502	-.451	-.226
	.1000	-.723	-.498	-.379
	.2250	-.511	-.396	-.404
	.4500	-.460	-.315	-.277
	.7500	-.247	-.272	-.085
Lower	.0500	-.553	-.728	-.694
	.1500	-.485	-.532	-.455
	.4000	-.374	-.353	-.251
	.7000	-.094	-.094	-.072
Aileron				
Upper	.8050	-.247	-.191	.038
	.8150	-.243	-.191	.140
	.8350	-.162	-.217	-.243
	.8750	-.132	-.200	-.162
	.9250	-.055	-.081	-.051
	.9750	.009	.013	.013
Lower	.8100	-.043	.000	-.017
	.8340	.089	.047	-.021
	.9250	-.051	-.004	-.004
	.9750	.034	.047	.064

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.702	.715
	.0100	-1.191	-1.209
	.0250	-1.745	-1.881
	.0500	-1.600	-1.719
	.1000	-1.187	-1.281
	.2250	-.800	-.757
	.4500	-.451	-.498
	.7500	-.289	-.494
	.8350	-.072	-.081
	.8750	-.060	-.051
	.9250	-.051	-.017
	.9750	.009	.060
Lower	.0500	.068	.221
	.1500	-.038	-.106
	.4000	-.157	-.132
	.7000	-.068	-.030
	.8530	-.098	-.026
	.9250	-.106	-.017
	.9750	-.017	.004

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.770	.885	.923
	.0100	-.996	-.660	-.349
	.0250	-1.681	-1.438	-.830
	.0500	-1.481	-1.336	-.830
	.1000	-1.285	-.945	-.736
	.2250	-.736	-.638	-.536
	.4500	-.536	-.396	-.302
	.7500	-.238	-.264	-.085
Lower	.0500	.094	-.026	-.149
	.1500	-.119	-.174	-.213
	.4000	-.200	-.200	-.162
	.7000	-.021	-.064	-.072
Aileron				
Upper	.8050	-.209	-.204	.000
	.8150	-.204	-.187	-.017
	.8350	-.145	-.183	-.128
	.8750	-.128	-.170	-.162
	.9250	-.021	-.064	-.077
	.9750	.030	.030	-.017
Lower	.8100	.034	.077	-.034
	.8340	.085	.098	-.051
	.9250	-.030	-.017	-.106
	.9750	.030	.055	-.047

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.102	.009
	.0100	-2.613	-2.868
	.0250	-2.906	-3.319
	.0500	-2.566	-2.970
	.1000	-1.664	-1.779
	.2250	-1.055	-1.000
	.4500	-.528	-.604
	.7500	-.277	-.434
	.8350	-.068	-.081
	.8750	-.034	-.034
	.9250	-.004	.030
	.9750	.068	.077
Lower	.0500	.591	.655
	.1500	.272	.179
	.4000	.017	.034
	.7000	.026	.047
	.8530	-.004	.009
	.9250	-.111	-.017
	.9750	-.009	-.017

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.009	.447	.626
	.0100	-2.536	-1.864	-1.213
	.0250	-2.974	-2.591	-1.545
	.0500	-2.634	-2.345	-1.485
	.1000	-1.821	-1.464	-1.068
	.2250	-1.013	-.838	-.677
	.4500	-.638	-.460	-.362
	.7500	-.217	-.243	-.140
Lower	.0500	.562	.413	.213
	.1500	.213	.115	-.021
	.4000	-.034	-.068	-.102
	.7000	.030	.004	-.128
Aileron				
Upper	.8050	-.191	-.200	-.111
	.8150	-.187	-.166	-.064
	.8350	-.145	-.166	-.174
	.8750	-.094	-.140	-.226
	.9250	-.030	-.030	-.115
	.9750	.034	.038	-.072
Lower	.8100	.081	.089	-.068
	.8340	.123	.149	-.128
	.9250	-.034	.017	-.170
	.9750	.004	.064	-.077

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.370	-1.230
	.0100	-4.306	-4.864
	.0250	-4.187	-4.949
	.0500	-3.583	-4.153
	.1000	-2.179	-2.281
	.2250	-1.260	-1.187
	.4500	-.604	-.651
	.7500	-.238	-.272
	.8350	-.077	-.077
	.8750	-.034	-.030
	.9250	-.004	.009
	.9750	.051	.047
Lower	.0500	.894	.949
	.1500	.523	.494
	.4000	.204	.221
	.7000	.128	.119
	.8530	.094	.047
	.9250	-.102	.004
	.9750	-.017	-.034

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.315	-.443	.038
	.0100	-4.468	-3.545	-2.336
	.0250	-4.523	-4.026	-2.417
	.0500	-3.949	-3.715	-2.209
	.1000	-2.383	-1.974	-1.383
	.2250	-1.209	-1.055	-.804
	.4500	-.643	-.506	-.387
	.7500	-.162	-.209	-.277
Lower	.0500	.881	.774	.545
	.1500	.489	.387	.196
	.4000	.153	.089	-.034
	.7000	.136	.064	-.166
Aileron				
Upper	.8050	-.136	-.166	-.217
	.8150	-.119	-.140	-.085
	.8350	-.102	-.123	-.209
	.8750	-.072	-.072	-.336
	.9250	-.009	-.004	-.260
	.9750	.009	.051	-.174
Lower	.8100	.136	.106	-.157
	.8340	.157	.200	.038
	.9250	-.034	.017	-.213
	.9750	.009	.055	-.140

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = 4.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.149	-1.302
	.0100	-3.881	-4.889
	.0250	-3.902	-4.945
	.0500	-3.285	-4.132
	.1000	-1.928	-2.268
	.2250	-1.170	-1.166
	.4500	-.587	-.638
	.7500	-.357	-.289
	.8350	-.294	-.149
	.8750	-.268	-.102
	.9250	-.226	-.072
	.9750	-.187	-.038
Lower	.0500	.834	.911
	.1500	.485	.460
	.4000	.174	.209
	.7000	.051	.081
	.8530	.026	-.026
	.9250	-.281	-.072
	.9750	-.187	-.094

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.655	-.762	-.268
	.0100	-4.817	-4.026	-2.753
	.0250	-5.017	-4.413	-2.698
	.0500	-4.166	-4.111	-2.434
	.1000	-2.498	-2.111	-1.502
	.2250	-1.234	-1.111	-.851
	.4500	-.668	-.523	-.426
	.7500	-.191	-.213	-.357
Lower	.0500	.860	.821	.604
	.1500	.502	.443	.221
	.4000	.149	.081	-.043
	.7000	.098	.055	-.217
Aileron				
Upper	.8050	-.187	-.170	-.272
	.8150	-.170	-.149	.009
	.8350	-.123	-.132	-.255
	.8750	-.102	-.081	-.383
	.9250	-.060	-.026	-.306
	.9750	-.030	.030	-.204
Lower	.8100	.106	.111	-.243
	.8340	.111	.174	.047
	.9250	-.051	.000	-.255
	.9750	-.038	.034	-.153

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 5.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.272	-1.481
	.0100	-3.949	-5.153
	.0250	-3.966	-5.140
	.0500	-3.268	-4.264
	.1000	-1.940	-2.302
	.2250	-1.149	-1.187
	.4500	-.621	-.660
	.7500	-.434	-.315
	.8350	-.366	-.187
	.8750	-.328	-.145
	.9250	-.302	-.102
	.9750	-.277	-.081
Lower	.0500	.860	.911
	.1500	.511	.494
	.4000	.191	.221
	.7000	.017	.072
	.8530	-.026	-.064
	.9250	-.328	-.094
	.9750	-.234	-.132

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.021	-1.017	-.477
	.0100	-5.204	-4.404	-3.034
	.0250	-5.328	-4.698	-2.881
	.0500	-4.370	-4.379	-2.519
	.1000	-2.604	-2.191	-1.545
	.2250	-1.268	-1.145	-.851
	.4500	-.664	-.523	-.430
	.7500	-.196	-.183	-.426
Lower	.0500	.889	.877	.651
	.1500	.540	.464	.260
	.4000	.170	.140	-.034
	.7000	.111	.072	-.238
Aileron				
Upper	.8050	-.162	-.145	-.302
	.8150	-.162	-.153	.060
	.8350	-.145	-.136	-.302
	.8750	-.111	-.089	-.430
	.9250	-.068	-.021	-.340
	.9750	-.060	.013	-.221
Lower	.8100	.102	.077	-.311
	.8340	.128	.196	.047
	.9250	-.060	-.000	-.255
	.9750	-.043	.026	-.187

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 6.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.196	-1.430
	.0100	-3.762	-4.962
	.0250	-3.745	-4.962
	.0500	-3.068	-4.102
	.1000	-1.774	-2.187
	.2250	-1.000	-1.098
	.4500	-.596	-.604
	.7500	-.460	-.345
	.8350	-.409	-.179
	.8750	-.379	-.149
	.9250	-.362	-.123
	.9750	-.328	-.094
Lower	.0500	.851	.932
	.1500	.511	.540
	.4000	.209	.230
	.7000	.043	.094
	.8530	-.051	-.060
	.9250	-.362	-.077
	.9750	-.272	-.111

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.940	-1.034	-.523
	.0100	-5.055	-4.336	-3.043
	.0250	-5.191	-4.647	-2.902
	.0500	-4.255	-4.349	-2.502
	.1000	-2.523	-2.166	-1.523
	.2250	-1.243	-1.115	-.847
	.4500	-.634	-.523	-.417
	.7500	-.200	-.217	-.421
Lower	.0500	.898	.864	.647
	.1500	.557	.455	.243
	.4000	.179	.140	-.043
	.7000	.115	.060	-.255
Aileron				
Upper	.8050	-.153	-.166	-.302
	.8150	-.157	-.162	.106
	.8350	-.136	-.145	-.281
	.8750	-.106	-.106	-.421
	.9250	-.060	-.055	-.319
	.9750	-.047	.004	-.213
Lower	.8100	.098	.064	-.311
	.8340	.132	.183	.081
	.9250	-.043	.004	-.243
	.9750	-.030	.013	-.170

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 7.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.298	-1.719
	.0100	-3.940	-5.387
	.0250	-3.906	-5.328
	.0500	-3.174	-4.340
	.1000	-1.809	-2.319
	.2250	-1.026	-1.136
	.4500	-.591	-.634
	.7500	-.481	-.391
	.8350	-.426	-.183
	.8750	-.438	-.170
	.9250	-.409	-.136
	.9750	-.391	-.115
Lower	.0500	.902	.970
	.1500	.579	.553
	.4000	.221	.264
	.7000	.068	.098
	.8530	-.094	-.060
	.9250	-.400	-.089
	.9750	-.306	-.136

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.494	-1.447	-.791
	.0100	-5.698	-4.966	-3.370
	.0250	-5.672	-5.132	-3.132
	.0500	-4.626	-4.757	-2.689
	.1000	-2.719	-2.285	-1.604
	.2250	-1.302	-1.140	-.872
	.4500	-.634	-.489	-.421
	.7500	-.187	-.170	-.447
Lower	.0500	.928	.936	.689
	.1500	.621	.587	.340
	.4000	.209	.204	-.017
	.7000	.136	.119	-.226
Aileron				
Upper	.8050	-.149	-.102	-.298
	.8150	-.149	-.102	.081
	.8350	-.136	-.094	-.294
	.8750	-.115	-.064	-.426
	.9250	-.077	-.034	-.374
	.9750	-.055	.004	-.234
Lower	.8100	.140	.115	-.315
	.8340	.140	.196	.004
	.9250	-.047	.030	-.247
	.9750	-.047	.030	-.204

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 8.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.506	-1.877
	.0100	-4.038	-5.498
	.0250	-3.928	-5.336
	.0500	-3.166	-4.340
	.1000	-1.817	-2.315
	.2250	-.987	-1.149
	.4500	-.630	-.634
	.7500	-.540	-.438
	.8350	-.502	-.243
	.8750	-.485	-.217
	.9250	-.477	-.179
	.9750	-.464	-.128
Lower	.0500	.855	.932
	.1500	.570	.557
	.4000	.243	.264
	.7000	.051	.089
	.8530	-.157	-.072
	.9250	-.451	-.115
	.9750	-.366	-.162

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.574	-1.498	-.970
	.0100	-5.791	-5.038	-3.634
	.0250	-5.719	-5.200	-3.302
	.0500	-4.638	-4.787	-2.779
	.1000	-2.715	-2.345	-1.643
	.2250	-1.294	-1.213	-.872
	.4500	-.626	-.545	-.434
	.7500	-.179	-.209	-.498
Lower	.0500	.936	.877	.719
	.1500	.634	.553	.315
	.4000	.230	.191	-.043
	.7000	.123	.089	-.264
Aileron				
Upper	.8050	-.166	-.157	-.328
	.8150	-.166	-.153	.102
	.8350	-.145	-.140	-.340
	.8750	-.119	-.102	-.489
	.9250	-.098	-.060	-.362
	.9750	-.089	-.021	-.209
Lower	.8100	.098	.077	-.379
	.8340	.119	.166	-.021
	.9250	-.068	-.021	-.268
	.9750	-.047	.000	-.238

TABLE VIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 9.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.000	-.617
	.0100	-3.340	-2.698
	.0250	-3.285	-2.187
	.0500	-2.694	-1.451
	.1000	-1.494	-1.068
	.2250	-.851	-.953
	.4500	-.638	-.843
	.7500	-.540	-.570
	.8350	-.438	-.489
	.8750	-.434	-.477
	.9250	-.421	-.443
	.9750	-.409	-.409
Lower	.0500	.817	.864
	.1500	.494	.438
	.4000	.157	.162
	.7000	-.004	.017
	.8530	-.102	-.123
	.9250	-.404	-.213
	.9750	-.315	-.357

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.681	-.638	-.434
	.0100	-2.421	-3.574	-2.872
	.0250	-1.906	-3.915	-2.749
	.0500	-1.285	-3.536	-2.387
	.1000	-.885	-1.711	-1.443
	.2250	-.902	-.911	-.791
	.4500	-.813	-.574	-.379
	.7500	-.549	-.438	-.374
Lower	.0500	.809	.766	.583
	.1500	.455	.417	.217
	.4000	.106	.077	-.072
	.7000	.021	.009	-.226
Aileron				
Upper	.8050	-.404	-.349	-.272
	.8150	-.404	-.349	-.060
	.8350	-.464	-.374	-.238
	.8750	-.396	-.349	-.345
	.9250	-.374	-.302	-.247
	.9750	-.353	-.264	-.153
Lower	.8100	-.026	.060	-.230
	.8340	.017	.102	.072
	.9250	-.226	-.136	-.196
	.9750	-.268	-.149	-.157

TABLE VIII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) $\alpha_f = 10.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.136	-.638
	.0100	-3.515	-2.689
	.0250	-3.345	-2.145
	.0500	-2.732	-1.430
	.1000	-1.485	-1.017
	.2250	-.821	-.940
	.4500	-.634	-.847
	.7500	-.511	-.587
	.8350	-.455	-.506
	.8750	-.451	-.502
	.9250	-.434	-.460
	.9750	-.400	-.413
Lower	.0500	.872	.885
	.1500	.511	.472
	.4000	.174	.191
	.7000	.021	.021
	.8530	-.145	-.128
	.9250	-.417	-.226
	.9750	-.336	-.379

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.613	-.566	-.502
	.0100	-2.187	-3.443	-2.945
	.0250	-1.540	-3.813	-2.787
	.0500	-1.047	-3.455	-2.438
	.1000	-.779	-1.702	-1.464
	.2250	-.834	-.919	-.787
	.4500	-.813	-.638	-.379
	.7500	-.600	-.489	-.413
Lower	.0500	.804	.770	.583
	.1500	.464	.417	.226
	.4000	.115	.094	-.085
	.7000	.034	.004	-.251
Aileron				
Upper	.8050	-.417	-.391	-.294
	.8150	-.455	-.391	-.055
	.8350	-.485	-.430	-.277
	.8750	-.426	-.396	-.374
	.9250	-.396	-.332	-.277
	.9750	-.349	-.281	-.183
Lower	.8100	-.017	.030	-.281
	.8340	.000	.089	.055
	.9250	-.226	-.140	-.247
	.9750	-.272	-.132	-.204

TABLE IX

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -6.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.968
	.0100		-.192
	.0250		-.915
	.0500		-.822
	.1000		-.964
	.2250		-.623
	.4500		-.445
	.7500		-.612
	.8350		-.114
	.8750		-.071
	.9250		-.032
	.9750		.050
Lower	.0500		-.310
	.1500		-.480
	.4000		-.292
	.7000		-.139
	.8530		-.064
	.9250		-.032
	.9750		.039

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.900	.947	.893
	.0100	.064	.342	.470
	.0250	-.601	-.359	-.039
	.0500	-.609	-.406	-.139
	.1000	-.776	-.466	-.374
	.2250	-.544	-.406	-.395
	.4500	-.509	-.342	-.317
	.7500	-.295	-.406	-.306
Lower	.0500	-.516	-.762	-.769
	.1500	-.484	-.548	-.470
	.4000	-.349	-.324	-.196
	.7000	-.082	.032	.032
Aileron				
Upper	.8050	-.313	-.413	-.302
	.8150	-.299	-.423	-.281
	.8350	-.160	-.370	-.395
	.8750	-.121	-.292	-.231
	.9250	-.025	-.103	-.053
	.9750	.011	.036	.025
Lower	.8100	.050	.142	.125
	.8340	.103	.199	.100
	.9250	-.093	.068	.064
	.9750	.028	.071	.057

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 16.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.875	.883
	.0100	-.762	-.623
	.0250	-1.402	-1.359
	.0500	-1.260	-1.224
	.1000	-1.064	-1.093
	.2250	-.765	-.705
	.4500	-.459	-.512
	.7500	-.224	-.498
	.8350	-.053	-.117
	.8750	-.053	-.082
	.9250	-.053	-.057
	.9750	-.053	.028
Lower	.0500	-.185	-.046
	.1500	-.246	-.267
	.4000	-.181	-.246
	.7000	-.125	-.089
	.8530	-.057	-.043
	.9250	-.110	-.057
	.9750	-.000	-.018

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.890	1.004	.957
	.0100	-.217	.014	.178
	.0250	-.986	-.715	-.342
	.0500	-.936	-.751	-.416
	.1000	-.986	-.669	-.534
	.2250	-.623	-.505	-.473
	.4500	-.552	-.363	-.335
	.7500	-.302	-.434	-.320
Lower	.0500	-.210	-.459	-.509
	.1500	-.324	-.367	-.374
	.4000	-.267	-.249	-.171
	.7000	-.032	.028	.004
Aileron				
Upper	.8050	-.317	-.416	-.306
	.8150	-.302	-.416	-.281
	.8350	-.181	-.367	-.416
	.8750	-.117	-.256	-.249
	.9250	-.039	-.096	-.093
	.9750	.021	.018	-.007
Lower	.8100	.075	.153	.139
	.8340	.157	.221	.149
	.9250	-.039	.057	.018
	.9750	.025	.060	-.007

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.648	.637
	.0100	-1.370	-1.416
	.0250	-1.972	-2.085
	.0500	-1.754	-1.875
	.1000	-1.267	-1.381
	.2250	-.897	-.833
	.4500	-.473	-.584
	.7500	-.285	-.516
	.8350	-.014	-.057
	.8750	-.018	-.032
	.9250	-.028	.014
	.9750	.043	.071
Lower	.0500	.238	.320
	.1500	.064	-.057
	.4000	-.100	-.121
	.7000	-.014	-.036
	.8530	-.135	-.014
	.9250	-.085	-.004
	.9750	-.000	-.004

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.722	.936	.954
	.0100	-1.028	-.580	-.249
	.0250	-1.719	-1.395	-.740
	.0500	-1.552	-1.327	-.776
	.1000	-1.310	-.957	-.751
	.2250	-.776	-.648	-.555
	.4500	-.569	-.452	-.338
	.7500	-.274	-.427	-.327
Lower	.0500	.160	-.064	-.217
	.1500	-.075	-.189	-.238
	.4000	-.164	-.181	-.132
	.7000	.000	.068	-.011
Aileron				
Upper	.8050	-.313	-.406	-.313
	.8150	-.263	-.388	-.281
	.8350	-.192	-.345	-.466
	.8750	-.117	-.228	-.299
	.9250	-.025	-.071	-.153
	.9750	.025	.032	-.050
Lower	.8100	.075	.189	.164
	.8340	.149	.274	.128
	.9250	-.050	.078	-.028
	.9750	.032	.060	-.028

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.185	.263
	.0100	-2.192	-2.299
	.0250	-2.616	-2.868
	.0500	-2.320	-2.569
	.1000	-1.548	-1.687
	.2250	-1.021	-.986
	.4500	-.651	-.651
	.7500	-.281	-.512
	.8350	-.053	-.082
	.8750	-.046	-.025
	.9250	-.078	-.007
	.9750	.050	.060
Lower	.0500	.388	.527
	.1500	.174	.060
	.4000	-.004	-.060
	.7000	-.078	-.014
	.8530	-.100	-.021
	.9250	-.128	-.032
	.9750	-.018	-.028

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.356	.769	.875
	.0100	-1.836	-1.199	-.683
	.0250	-2.452	-1.975	-1.128
	.0500	-2.171	-1.829	-1.114
	.1000	-1.637	-1.224	-.907
	.2250	-.918	-.751	-.626
	.4500	-.655	-.448	-.363
	.7500	-.288	-.367	-.324
Lower	.0500	.395	.210	.025
	.1500	.078	-.011	-.096
	.4000	-.093	-.075	-.082
	.7000	.028	.117	-.011
Aileron				
Upper	.8050	-.302	-.356	-.285
	.8150	-.249	-.342	-.295
	.8350	-.189	-.285	-.459
	.8750	-.110	-.185	-.327
	.9250	-.039	-.053	-.174
	.9750	.004	.039	-.064
Lower	.8100	.125	.210	.157
	.8340	.157	.288	.125
	.9250	-.057	.082	-.025
	.9750	.007	.064	-.064

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 16.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = 0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.256	.459	.648
	.0100	-2.843	-1.993	-1.267
	.0250	-3.402	-2.744	-1.616
	.0500	-2.929	-2.509	-1.537
	.1000	-1.964	-1.541	-1.146
	.2250	-1.046	-.904	-.726
	.4500	-.719	-.527	-.402
	.7500	-.178	-.363	-.374
Lower	.0500	.598	.441	.235
	.1500	.249	.125	.000
	.4000	.000	-.039	-.082
	.7000	.032	.096	-.064
Aileron				
Upper	.8050	-.285	-.342	-.335
	.8150	-.206	-.335	-.324
	.8350	-.157	-.278	-.548
	.8750	-.085	-.164	-.406
	.9250	-.021	-.060	-.238
	.9750	-.028	.028	-.117
Lower	.8100	.064	.199	.121
	.8340	.114	.292	.078
	.9250	-.028	.068	-.132
	.9750	-.046	.050	-.135

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 1.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.459	-.356
	.0100	-3.117	-3.484
	.0250	-3.452	-3.851
	.0500	-2.911	-3.381
	.1000	-1.815	-1.982
	.2250	-1.096	-1.075
	.4500	-.552	-.665
	.7500	-.221	-.406
	.8350	.021	-.046
	.8750	.021	-.014
	.9250	.068	.036
	.9750	.117	.093
Lower	.0500	.836	.836
	.1500	.423	.313
	.4000	.217	.103
	.7000	.132	.060
	.8530	-.050	.036
	.9250	-.146	.000
	.9750	-.014	-.036

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.356	.313	.530
	.0100	-3.039	-2.246	-1.434
	.0250	-3.445	-2.972	-1.754
	.0500	-2.982	-2.708	-1.648
	.1000	-1.996	-1.619	-1.185
	.2250	-1.050	-.943	-.747
	.4500	-.662	-.527	-.391
	.7500	-.221	-.345	-.384
Lower	.0500	.687	.523	.302
	.1500	.299	.189	.036
	.4000	.032	-.007	-.057
	.7000	.082	.103	-.085
Aileron				
Upper	.8050	-.242	-.306	-.338
	.8150	-.181	-.302	-.313
	.8350	-.146	-.238	-.541
	.8750	-.064	-.149	-.413
	.9250	-.032	-.046	-.242
	.9750	.007	.025	-.110
Lower	.8100	.135	.203	.142
	.8340	.153	.295	.107
	.9250	-.043	.064	-.125
	.9750	.014	.039	-.117

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 2.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.943
	.0100		-4.548
	.0250		-4.829
	.0500		-3.904
	.1000		-2.263
	.2250		-1.149
	.4500		-.690
	.7500		-.342
	.8350		-.036
	.8750		.014
	.9250		.060
	.9750		.103
Lower	.0500		.961
	.1500		.488
	.4000		.256
	.7000		.146
	.8530		.093
	.9250		.078
	.9750		.014

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.982	.078	.381
	.0100	-3.954	-2.769	-1.801
	.0250	-4.164	-3.438	-2.004
	.0500	-3.580	-3.167	-1.872
	.1000	-2.306	-1.776	-1.278
	.2250	-1.171	-1.007	-.790
	.4500	-.662	-.527	-.409
	.7500	-.214	-.278	-.356
Lower	.0500	.772	.676	.409
	.1500	.384	.317	.121
	.4000	.157	.085	-.032
	.7000	.071	.149	-.096
Aileron				
Upper	.8050	-.174	-.292	-.359
	.8150	-.128	-.246	-.313
	.8350	-.093	-.192	-.544
	.8750	-.057	-.117	-.434
	.9250	.007	-.018	-.281
	.9750	.014	.036	-.135
Lower	.8100	.078	.246	.096
	.8340	.117	.420	.093
	.9250	-.064	.103	-.146
	.9750	.039	.068	-.139

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.189	-1.075
	.0100	-4.064	-4.648
	.0250	-4.228	-4.772
	.0500	-3.402	-4.011
	.1000	-2.107	-2.249
	.2250	-1.242	-1.196
	.4500	-.587	-.669
	.7500	-.199	-.342
	.8350	-.018	-.064
	.8750	-.036	-.032
	.9250	-.007	.007
	.9750	.100	.050
Lower	.0500	.911	.833
	.1500	.559	.470
	.4000	.288	.217
	.7000	.085	.125
	.8530	.032	.100
	.9250	-.128	.007
	.9750	.025	-.032

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.224	-.199	.153
	.0100	-4.242	-3.242	-2.114
	.0250	-4.491	-3.836	-2.281
	.0500	-3.744	-3.509	-2.085
	.1000	-2.327	-1.936	-1.388
	.2250	-1.185	-1.075	-.826
	.4500	-.673	-.559	-.441
	.7500	-.210	-.295	-.420
Lower	.0500	.822	.726	.477
	.1500	.434	.363	.142
	.4000	.128	.075	-.057
	.7000	.107	.125	-.142
Aileron				
Upper	.8050	-.149	-.242	-.399
	.8150	-.157	-.228	-.367
	.8350	-.117	-.185	-.555
	.8750	-.060	-.110	-.491
	.9250	-.007	-.036	-.313
	.9750	.000	.032	-.171
Lower	.8100	.128	.228	.036
	.8340	.128	.420	.053
	.9250	-.028	.064	-.160
	.9750	.000	.032	-.160

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 3.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.459	-1.299
	.0100	-4.484	-5.064
	.0250	-4.552	-5.028
	.0500	-3.559	-4.178
	.1000	-2.285	-2.352
	.2250	-1.370	-1.267
	.4500	-.765	-.712
	.7500	-.231	-.320
	.8350	-.014	-.110
	.8750	-.050	-.075
	.9250	-.046	-.036
	.9750	.046	.014
Lower	.0500	.441	.865
	.1500	.544	.452
	.4000	.199	.206
	.7000	.096	.160
	.8530	-.043	.025
	.9250	-.164	-.018
	.9750	-.075	-.071

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.448	-.324	.032
	.0100	-4.555	-3.523	-2.367
	.0250	-4.779	-4.103	-2.480
	.0500	-3.968	-3.765	-2.242
	.1000	-2.463	-2.046	-1.473
	.2250	-1.281	-1.114	-.872
	.4500	-.712	-.566	-.463
	.7500	-.217	-.288	-.423
Lower	.0500	.794	.758	.537
	.1500	.438	.370	.171
	.4000	.135	.096	-.050
	.7000	.093	.114	-.160
Aileron				
Upper	.8050	-.181	-.238	-.413
	.8150	-.146	-.224	-.388
	.8350	-.114	-.178	-.559
	.8750	-.075	-.117	-.523
	.9250	-.036	-.039	-.359
	.9750	.011	.018	-.196
Lower	.8100	.110	.221	.007
	.8340	.082	.409	.032
	.9250	-.043	.046	-.178
	.9750	-.000	.036	-.178

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 3.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.794	-1.769
	.0100	-4.758	-5.626
	.0250	-4.786	-5.505
	.0500	-3.815	-4.544
	.1000	-2.441	-2.512
	.2250	-1.537	-1.388
	.4500	-.779	-.769
	.7500	-.374	-.278
	.8350	-.196	-.121
	.8750	-.114	-.075
	.9250	-.071	-.036
	.9750	-.011	-.050
Lower	.0500	.808	.879
	.1500	.495	.448
	.4000	.270	.174
	.7000	.021	.014
	.8530	.021	-.078
	.9250	-.192	-.114
	.9750	-.149	-.043

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-5.288	-.754	-.221
	.0100	-5.374	-4.214	-2.737
	.0250	-4.395	-4.623	-2.726
	.0500	-2.641	-4.299	-2.427
	.1000	-1.331	-2.199	-1.548
	.2250	-.726	-1.178	-.897
	.4500	-.167	-.569	-.448
	.7500	.879	-.221	-.423
Lower	.0500	.484	.819	.623
	.1500	.142	.438	.242
	.4000	.071	.128	-.021
	.7000	-.139	.096	-.157
Aileron				
Upper	.8050	-.121	-.185	-.395
	.8150	-.110	-.178	-.381
	.8350	-.057	-.142	-.555
	.8750	-.021	-.089	-.548
	.9250	.004	-.021	-.399
	.9750	.103	.014	-.221
Lower	.8100	.071	.206	-.018
	.8340	-.053	.384	.004
	.9250	-.018	.032	-.210
	.9750	.021	.021	-.192

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) $\alpha_f = 4.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.893	-1.922
	.0100	-4.890	-5.886
	.0250	-5.025	-5.786
	.0500	-3.922	-4.726
	.1000	-2.391	-2.630
	.2250	-1.388	-1.292
	.4500	-.523	-.687
	.7500	-.164	-.199
	.8350	-.142	-.107
	.8750	-.075	-.071
	.9250	-.025	-.043
	.9750	.046	-.021
Lower	.0500	.957	.936
	.1500	.690	.544
	.4000	.313	.274
	.7000	.125	.132
	.8530	.050	.057
	.9250	-.135	.011
	.9750	-.018	-.082

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.434	-.968	-.441
	.0100	-5.673	-4.548	-3.053
	.0250	-5.644	-4.886	-2.922
	.0500	-4.601	-4.562	-2.544
	.1000	-2.715	-2.278	-1.584
	.2250	-1.363	-1.174	-.890
	.4500	-.690	-.544	-.434
	.7500	-.164	-.178	-.402
Lower	.0500	.943	.883	.648
	.1500	.562	.505	.278
	.4000	.214	.157	-.014
	.7000	.125	.121	-.171
Aileron				
Upper	.8050	-.135	-.132	-.377
	.8150	-.121	-.125	-.352
	.8350	-.093	-.107	-.498
	.8750	-.075	-.057	-.527
	.9250	-.025	-.000	-.370
	.9750	-.021	.039	-.196
Lower	.8100	.078	.203	-.028
	.8340	.107	.391	.060
	.9250	-.025	.053	-.189
	.9750	-.011	.036	-.196

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(*l*) $\alpha_f = 4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.057	-2.089
	.0100	-5.135	-6.128
	.0250	-5.064	-5.879
	.0500	-4.025	-4.762
	.1000	-2.438	-2.658
	.2250	-1.406	-1.384
	.4500	-.665	-.680
	.7500	-.295	-.224
	.8350	-.174	-.139
	.8750	-.157	-.139
	.9250	-.114	-.107
	.9750	-.089	-.071
Lower	.0500	.900	.918
	.1500	.605	.562
	.4000	.242	.267
	.7000	.100	.096
	.8530	.060	.011
	.9250	-.189	-.053
	.9750	-.117	-.132

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.730	-1.335	-.779
	.0100	-5.982	-5.057	-3.505
	.0250	-5.900	-5.299	-3.235
	.0500	-4.808	-4.868	-2.786
	.1000	-2.826	-2.438	-1.694
	.2250	-1.434	-1.270	-.943
	.4500	-.701	-.609	-.466
	.7500	-.181	-.217	-.445
Lower	.0500	.886	.854	.694
	.1500	.573	.488	.267
	.4000	.203	.153	-.036
	.7000	.082	.068	-.221
Aileron				
Upper	.8050	-.146	-.167	-.434
	.8150	-.149	-.149	-.327
	.8350	-.139	-.132	-.512
	.8750	-.110	-.096	-.552
	.9250	-.075	-.050	-.416
	.9750	-.039	-.007	-.246
Lower	.8100	.043	.167	-.103
	.8340	.057	.352	-.021
	.9250	-.085	-.004	-.270
	.9750	-.053	.004	-.242

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(m) $\alpha_f = 5.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.552	-1.797
	.0100	-4.480	-5.577
	.0250	-4.480	-5.345
	.0500	-3.584	-4.281
	.1000	-2.110	-2.452
	.2250	-1.167	-1.335
	.4500	-.559	-.676
	.7500	-.302	-.238
	.8350	-.263	-.160
	.8750	-.246	-.132
	.9250	-.164	-.093
	.9750	-.103	-.071
Lower	.0500	.950	.972
	.1500	.623	.559
	.4000	.320	.253
	.7000	.121	.135
	.8530	.039	.007
	.9250	-.242	-.046
	.9750	-.139	-.114

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.591	-1.359	-.751
	.0100	-5.698	-4.968	-3.391
	.0250	-5.722	-5.174	-3.135
	.0500	-4.598	-4.701	-2.680
	.1000	-2.609	-2.338	-1.609
	.2250	-1.367	-1.235	-.883
	.4500	-.680	-.562	-.420
	.7500	-.174	-.192	-.384
Lower	.0500	.893	.843	.655
	.1500	.559	.502	.278
	.4000	.203	.160	-.014
	.7000	.093	.078	-.185
Aileron				
Upper	.8050	-.153	-.160	-.381
	.8150	-.146	-.142	-.292
	.8350	-.135	-.121	-.473
	.8750	-.107	-.100	-.523
	.9250	-.068	-.057	-.391
	.9750	-.043	-.018	-.228
Lower	.8100	.060	.164	-.082
	.8340	.078	.345	-.018
	.9250	-.071	.000	-.249
	.9750	-.039	.000	-.228

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(n) $\alpha_f = 5.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.630
	.0100		-5.388
	.0250		-5.292
	.0500		-4.345
	.1000		-2.406
	.2250		-1.171
	.4500		-.562
	.7500		-.270
	.8350		-.132
	.8750		-.089
	.9250		-.117
	.9750		-.100
Lower	.0500		.918
	.1500		.616
	.4000		.263
	.7000		.135
	.8530		.000
	.9250		-.032
	.9750		-.078

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.491	-1.249	-.701
	.0100	-5.644	-4.883	-3.359
	.0250	-5.619	-5.128	-3.078
	.0500	-4.577	-4.708	-2.708
	.1000	-2.616	-2.345	-1.637
	.2250	-1.302	-1.210	-.893
	.4500	-.676	-.559	-.434
	.7500	-.203	-.203	-.416
Lower	.0500	.847	.858	.662
	.1500	.562	.488	.278
	.4000	.192	.139	-.025
	.7000	.085	.068	-.189
Aileron				
Upper	.8050	-.171	-.157	-.399
	.8150	-.135	-.142	-.302
	.8350	-.100	-.121	-.480
	.8750	-.082	-.089	-.516
	.9250	-.064	-.050	-.363
	.9750	-.007	-.018	-.206
Lower	.8100	.050	.153	-.057
	.8340	.046	.349	.043
	.9250	-.032	-.004	-.253
	.9750	-.043	.018	-.224

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(o) $\alpha_f = 6.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.577	-1.950
	.0100	-4.431	-5.762
	.0250	-4.441	-5.559
	.0500	-3.470	-4.463
	.1000	-2.110	-2.544
	.2250	-1.206	-1.388
	.4500	-.765	-.708
	.7500	-.491	-.349
	.8350	-.470	-.164
	.8750	-.409	-.178
	.9250	-.399	-.174
	.9750	-.327	-.135
Lower	.0500	.747	.854
	.1500	.520	.619
	.4000	.238	.128
	.7000	.007	-.011
	.8530	.018	-.075
	.9250	-.359	-.125
	.9750	-.256	-.199

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.836	-1.477	-.872
	.0100	-5.975	-5.199	-3.566
	.0250	-5.861	-5.356	-3.228
	.0500	-4.715	-4.836	-2.794
	.1000	-2.747	-2.388	-1.651
	.2250	-1.402	-1.214	-.886
	.4500	-.694	-.562	-.423
	.7500	-.231	-.178	-.416
Lower	.0500	.868	.861	.683
	.1500	.552	.537	.299
	.4000	.181	.171	-.007
	.7000	.028	.068	-.196
Aileron				
Upper	.8050	-.196	-.142	-.391
	.8150	-.189	-.114	-.260
	.8350	-.160	-.096	-.459
	.8750	-.121	-.068	-.516
	.9250	-.107	-.032	-.370
	.9750	-.053	-.000	-.210
Lower	.8100	.007	.164	-.078
	.8340	.046	.363	.004
	.9250	-.110	-.007	-.249
	.9750	-.078	.007	-.224

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(p) $\alpha_f = 6.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-2.865 -6.053 -5.932 -4.836 -2.769 -1.416 -.765 -.253	-1.302 -4.986 -5.235 -4.794 -2.370 -1.203 -.544 -.178	-.680 -3.352 -3.093 -2.715 -1.633 -.893 -.448 -.402
Lower	.0500 .1500 .4000 .7000	.893 .619 .238 .028	.883 .544 .185 .103	.665 .281 -.025 -.203
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.078 -.142 -.128 -.117 -.093 -.093	-.139 -.125 -.103 -.068 -.032 .011	-.391 -.288 -.463 -.512 -.363 -.219
Lower	.8100 .8340 .9250 .9750	-.004 .078 -.093 -.139	.149 .377 .021 .021	-.093 -.039 -.267 -.221

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(q) $\alpha_f = 7.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-2.018
	.0100		-5.815
	.0250		-5.544
	.0500		-4.477
	.1000		-2.448
	.2250		-1.181
	.4500		-.619
	.7500		-.406
	.8350		-.157
	.8750		-.121
	.9250		-.174
	.9750		-.053
Lower	.0500		.975
	.1500		.662
	.4000		.270
	.7000		.214
	.8530		.028
	.9250		-.110
	.9750		-.149

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-3.178	-1.694	-1.082
	.0100	-6.171	-5.413	-3.819
	.0250	-5.922	-5.530	-3.370
	.0500	-4.819	-4.932	-2.922
	.1000	-2.783	-2.416	-1.683
	.2250	-1.352	-1.221	-.868
	.4500	-.580	-.534	-.409
	.7500	-.228	-.174	-.452
Lower	.0500	.922	.865	.712
	.1500	.662	.559	.317
	.4000	.246	.199	-.028
	.7000	.107	.075	-.253
Aileron				
Upper	.8050	-.157	-.139	-.431
	.8150	-.149	-.121	-.210
	.8350	-.135	-.110	-.420
	.8750	-.103	-.089	-.473
	.9250	-.068	-.071	-.327
	.9750	-.050	-.043	-.206
Lower	.8100	.057	.132	-.142
	.8340	.110	.324	-.032
	.9250	-.078	-.025	-.256
	.9750	-.028	-.007	-.206

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(r) $\alpha_f = 7.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.270	-1.313
	.0100	-3.897	-5.530
	.0250	-3.865	-5.217
	.0500	-3.050	-4.100
	.1000	-1.797	-2.224
	.2250	-1.046	-1.078
	.4500	-.562	-.580
	.7500	-.409	-.306
	.8350	-.342	-.189
	.8750	-.260	-.231
	.9250	-.274	-.196
	.9750	-.285	-.128
Lower	.0500	.900	1.014
	.1500	.609	.683
	.4000	.238	.295
	.7000	.025	.174
	.8530	.068	.050
	.9250	-.253	-.004
	.9750	-.185	-.146

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.769	-.925	-.559
	.0100	-4.338	-4.174	-3.078
	.0250	-5.039	-4.544	-2.858
	.0500	-2.957	-4.181	-2.505
	.1000	-1.769	-2.025	-1.498
	.2250	-1.043	-1.011	-.815
	.4500	-.594	-.456	-.374
	.7500	-.256	-.238	-.391
Lower	.0500	.943	.872	.630
	.1500	.623	.534	.285
	.4000	.263	.189	-.011
	.7000	.174	.121	-.160
Aileron				
Upper	.8050	-.210	-.189	-.384
	.8150	-.139	-.164	-.292
	.8350	-.192	-.164	-.356
	.8750	-.082	-.142	-.381
	.9250	-.039	-.096	-.249
	.9750	-.053	-.082	-.110
Lower	.8100	.203	.203	.021
	.8340	.203	.370	.114
	.9250	-.025	.014	-.117
	.9750	.025	-.014	-.103

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(s) $\alpha_f = 8.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.000	-1.349
	.0100	-3.021	-4.167
	.0250	-3.018	-4.068
	.0500	-2.441	-3.295
	.1000	-1.388	-1.851
	.2250	-.815	-.911
	.4500	-.520	-.498
	.7500	-.399	-.335
	.8350	-.370	-.203
	.8750	-.349	-.189
	.9250	-.338	-.142
	.9750	-.327	-.107
Lower	.0500	.676	.779
	.1500	.423	.427
	.4000	.167	.196
	.7000	.011	.071
	.8530	.043	-.057
	.9250	-.356	-.103
	.9750	-.274	-.146

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.968	-1.085	-.722
	.0100	-4.477	-3.833	-2.751
	.0250	-4.431	-3.996	-2.523
	.0500	-3.566	-3.698	-2.199
	.1000	-2.078	-1.826	-1.292
	.2250	-1.046	-.968	-.708
	.4500	-.527	-.456	-.359
	.7500	-.103	-.178	-.395
Lower	.0500	.712	.662	.495
	.1500	.456	.388	.196
	.4000	.171	.121	-.050
	.7000	.064	.068	-.181
Aileron				
Upper	.8050	-.121	-.142	-.409
	.8150	-.114	-.132	-.288
	.8350	-.103	-.114	-.370
	.8750	-.082	-.082	-.377
	.9250	-.057	-.050	-.242
	.9750	-.039	-.032	-.117
Lower	.8100	.057	.096	-.028
	.8340	.071	.224	.028
	.9250	-.064	.000	-.132
	.9750	-.028	-.014	-.135

TABLE IX.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 16.9 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(t) $\alpha_f = 8.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-1.737 -5.253 -4.804 -3.836 -2.210 -1.274 -.790 -.598 -.356 -.306 -.306 -.278
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.890 .516 .206 -.057 -.107 -.174 -.274

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-2.548 -5.295 -5.046 -3.943 -2.214 -1.352 -.783 -.399	-1.320 -4.886 -5.028 -4.438 -2.214 -1.206 -.609 -.288	-.786 -3.438 -3.153 -2.730 -1.626 -.893 -.423 -.441
Lower	.0500 .1500 .4000 .7000	.836 .505 .128 -.014	.840 .484 .142 .046	.644 .270 -.028 -.217
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.320 -.317 -.320 -.310 -.292 -.246	-.231 -.217 -.210 -.181 -.146 -.114	-.416 -.281 -.498 -.530 -.384 -.260
Lower	.8100 .8340 .9250 .9750	-.043 .011 -.210 -.217	.096 .295 -.028 -.028	-.135 -.046 -.270 -.238

TABLE IX.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 16.9$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(u) $\alpha_f = 9.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.217	-1.370	-.833
	.0100	-6.149	-4.915	-3.498
	.0250	-5.922	-5.064	-3.171
	.0500	-3.544	-4.512	-2.744
	.1000	-2.117	-2.253	-1.616
	.2250	-1.335	-1.149	-.879
	.4500	-.797	-.580	-.420
	.7500	-.548	-.299	-.431
Lower	.0500	.968	.836	.676
	.1500	.541	.495	.281
	.4000	.249	.174	-.018
	.7000	.046	.085	-.224
Aileron				
Upper	.8050	-.249	-.231	-.423
	.8150	-.342	-.224	-.281
	.8350	-.292	-.217	-.505
	.8750	-.228	-.196	-.523
	.9250	-.246	-.142	-.388
	.9750	-.299	-.142	-.270
Lower	.8100	.014	.064	-.153
	.8340	-.046	.274	-.021
	.9250	-.164	-.021	-.267
	.9750	-.203	-.039	-.263

TABLE X

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -6.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.881 .215 -.478 -.409 -.716 -.472 -.525 -.313	.907 .475 -.203 -.269 -.379 -.328 -.290 -.415	.878 .504 .012 -.090 -.340 -.358 -.293 -.266
Lower	.0500 .1500 .4000 .7000	-.615 -.510 -.370 -.113	-.842 -.594 -.328 .036	-.845 -.501 -.221 .039
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.310 -.281 -.197 -.152 -.054 -.000	-.382 -.388 -.346 -.269 -.096 .033	-.197 -.173 -.364 -.218 -.054 .042
Lower	.8100 .8340 .9250 .9750	.024 .084 -.054 .048	.146 .227 .072 .018	.096 .110 .087 .069

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^\circ$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.887	.901
	.0100	-.818	-.675
	.0250	-1.451	-1.427
	.0500	-1.325	-1.284
	.1000	-1.048	-1.069
	.2250	-.687	-.639
	.4500	-.445	-.454
	.7500	-.299	-.531
	.8350	-.060	-.039
	.8750	-.066	-.048
	.9250	-.104	.006
	.9750	-.087	.093
Lower	.0500	-.101	.048
	.1500	-.158	-.176
	.4000	-.096	-.269
	.7000	-.042	-.054
	.8530	-.087	-.000
	.9250	-.122	-.033
	.9750	-.018	.006

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.916	.997	.955
	.0100	-.346	.030	.197
	.0250	-1.030	-.704	-.304
	.0500	-.958	-.728	-.382
	.1000	-1.039	-.639	-.513
	.2250	-.630	-.457	-.436
	.4500	-.528	-.352	-.310
	.7500	-.284	-.394	-.257
Lower	.0500	-.203	-.436	-.499
	.1500	-.307	-.394	-.358
	.4000	-.281	-.254	-.170
	.7000	-.036	.048	.027
Aileron				
Upper	.8050	-.287	-.376	-.191
	.8150	-.269	-.364	-.191
	.8350	-.173	-.325	-.349
	.8750	-.101	-.248	-.209
	.9250	-.036	-.078	-.066
	.9750	.015	.042	.012
Lower	.8100	.051	.176	.155
	.8340	.054	.299	.054
	.9250	-.039	.063	.084
	.9750	.012	-.024	.027

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.782	.925	.946
	.0100	-.922	-.537	-.236
	.0250	-1.660	-1.337	-.716
	.0500	-1.555	-1.287	-.764
	.1000	-1.313	-.931	-.743
	.2250	-.681	-.621	-.534
	.4500	-.490	-.397	-.325
	.7500	-.197	-.415	-.284
Lower	.0500	.125	-.075	-.236
	.1500	-.033	-.194	-.224
	.4000	-.149	-.176	-.164
	.7000	-.012	.072	.000
Aileron				
Upper	.8050	-.200	-.349	-.212
	.8150	-.191	-.361	-.212
	.8350	-.146	-.328	-.379
	.8750	-.018	-.248	-.260
	.9250	.000	-.069	-.122
	.9750	.060	.039	-.015
Lower	.8100	.119	.194	.155
	.8340	.119	.319	.072
	.9250	.015	.051	-.003
	.9750	.096	-.069	-.009

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.236
	.0100		-2.397
	.0250		-2.964
	.0500		-2.657
	.1000		-1.704
	.2250		-.931
	.4500		-.603
	.7500		-.316
	.8350		.030
	.8750		.030
	.9250		.039
	.9750		.054
Lower	.0500		.525
	.1500		.146
	.4000		.081
	.7000		.057
	.8530		.009
	.9250		-.033
	.9750		-.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.307	.743	.830
	.0100	-1.937	-1.236	-.728
	.0250	-2.487	-2.054	-1.158
	.0500	-2.239	-1.940	-1.158
	.1000	-1.767	-1.251	-.949
	.2250	-.952	-.776	-.651
	.4500	-.621	-.478	-.382
	.7500	-.260	-.376	-.322
Lower	.0500	.379	.230	.024
	.1500	.113	-.006	-.110
	.4000	-.090	-.101	-.116
	.7000	.069	.081	.003
Aileron				
Upper	.8050	-.290	-.346	-.239
	.8150	-.275	-.337	-.287
	.8350	-.179	-.287	-.448
	.8750	-.104	-.182	-.310
	.9250	-.057	-.066	-.167
	.9750	.027	.021	-.078
Lower	.8100	.087	.206	.140
	.8340	.021	.343	.069
	.9250	-.075	.042	-.030
	.9750	.009	-.101	-.075

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = -0.1^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.257	-.039
	.0100	-2.722	-2.958
	.0250	-3.006	-3.424
	.0500	-2.666	-3.015
	.1000	-1.734	-1.803
	.2250	-1.107	-.976
	.4500	-.582	-.627
	.7500	-.281	-.406
	.8350	-.057	-.066
	.8750	-.033	-.036
	.9250	.006	.009
	.9750	.024	.081
Lower	.0500	.573	.701
	.1500	.290	.218
	.4000	.045	.078
	.7000	.048	.060
	.8530	-.066	.012
	.9250	-.152	-.006
	.9750	-.042	-.018

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.018	.564	.728
	.0100	-2.481	-1.693	-1.042
	.0250	-2.976	-2.501	-1.442
	.0500	-2.564	-2.284	-1.394
	.1000	-1.881	-1.433	-1.060
	.2250	-.964	-.827	-.696
	.4500	-.645	-.490	-.397
	.7500	-.242	-.373	-.361
Lower	.0500	.525	.367	.167
	.1500	.206	.081	-.036
	.4000	-.012	-.057	-.096
	.7000	.075	.081	-.012
Aileron				
Upper	.8050	-.206	-.328	-.290
	.8150	-.212	-.322	-.313
	.8350	-.161	-.284	-.475
	.8750	-.075	-.176	-.367
	.9250	-.012	-.066	-.212
	.9750	.024	.006	-.093
Lower	.8100	.134	.200	.119
	.8340	.107	.322	.045
	.9250	-.012	.048	-.057
	.9750	-.006	-.084	-.131

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = 0.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.472	-.355
	.0100	-3.096	-3.531
	.0250	-3.296	-3.412
	.0500	-2.878	-2.006
	.1000	-1.767	-1.075
	.2250	-1.110	-.690
	.4500	-.603	-.433
	.7500	-.245	.690
	.8350	-.084	-.033
	.8750	-.069	.006
	.9250	.006	.045
	.9750	.015	-.003
Lower	.0500	.367	.275
	.1500	.284	.084
	.4000	.143	.039
	.7000	.027	-.039
	.8530	-.075	-.021
	.9250	-.122	-.042
	.9750	-.063	.012

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.866	.394	.603
	.0100	-3.310	-1.991	-1.203
	.0250	-2.833	-2.728	-1.558
	.0500	-1.973	-2.484	-1.493
	.1000	-1.036	-1.504	-1.099
	.2250	-.657	-.881	-.710
	.4500	-.248	-.510	-.394
	.7500	.606	-.364	-.388
Lower	.0500	.236	.436	.200
	.1500	-.021	.119	-.006
	.4000	.069	-.048	-.087
	.7000	-.218	.084	-.033
Aileron				
Upper	.8050	-.215	-.322	-.287
	.8150	-.173	-.319	-.299
	.8350	-.093	-.269	-.478
	.8750	-.039	-.179	-.388
	.9250	-.000	-.075	-.239
	.9750	.122	.003	-.113
Lower	.8100	.099	.173	.090
	.8340	-.024	.290	.042
	.9250	-.018	.042	-.048
	.9750	-.009	-.093	-.110

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.319
	.0100		-3.609
	.0250		-3.940
	.0500		-3.388
	.1000		-1.976
	.2250		-1.063
	.4500		-.687
	.7500		-.385
	.8350		-.030
	.8750		.018
	.9250		.057
	.9750		.042
Lower	.0500		.776
	.1500		.358
	.4000		.185
	.7000		.116
	.8530		.033
	.9250		-.006
	.9750		-.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.352	.343	.546
	.0100	-3.090	-2.206	-1.472
	.0250	-3.612	-2.970	-1.764
	.0500	-3.009	-2.722	-1.672
	.1000	-2.128	-1.624	-1.191
	.2250	-1.024	-.910	-.737
	.4500	-.690	-.525	-.403
	.7500	-.197	-.358	-.349
Lower	.0500	.621	.510	.301
	.1500	.328	.164	.033
	.4000	-.003	-.024	-.060
	.7000	.090	.110	-.042
Aileron				
Upper	.8050	-.251	-.325	-.343
	.8150	-.203	-.316	-.257
	.8350	-.149	-.257	-.525
	.8750	-.119	-.170	-.427
	.9250	-.042	-.057	-.272
	.9750	.030	.009	-.125
Lower	.8100	.090	.209	.072
	.8340	.036	.334	.042
	.9250	-.042	.036	-.066
	.9750	-.045	-.078	-.134

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) $\alpha_f = 1.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.907	-.731
	.0100	-3.704	-4.230
	.0250	-3.824	-4.436
	.0500	-3.236	-3.749
	.1000	-2.030	-2.125
	.2250	-1.218	-1.146
	.4500	-.639	-.704
	.7500	-.284	-.346
	.8350	-.057	-.090
	.8750	-.042	-.060
	.9250	.018	-.003
	.9750	.024	.048
Lower	.0500	-.084	.842
	.1500	.409	.352
	.4000	.128	.161
	.7000	.096	.078
	.8530	-.107	.042
	.9250	-.158	-.018
	.9750	-.072	-.060

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.776	.143	.418
	.0100	-3.704	-2.725	-1.707
	.0250	-4.090	-3.454	-1.979
	.0500	-3.403	-3.143	-1.836
	.1000	-2.287	-1.776	-1.263
	.2250	-1.140	-.982	-.770
	.4500	-.687	-.534	-.409
	.7500	-.221	-.328	-.409
Lower	.0500	.734	.630	.391
	.1500	.358	.272	.104
	.4000	.048	.039	-.030
	.7000	.101	.140	-.033
Aileron				
Upper	.8050	-.203	-.290	-.382
	.8150	-.203	-.293	-.331
	.8350	-.158	-.221	-.573
	.8750	-.104	-.128	-.463
	.9250	-.060	-.045	-.296
	.9750	-.030	.027	-.143
Lower	.8100	.140	.242	.045
	.8340	.096	.367	.048
			.075	-.057
	.9750	-.045	-.051	-.143

TABLE X.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 1.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.125	-.967
	.0100	-4.033	-4.645
	.0250	-3.964	-4.716
	.0500	-3.310	-3.979
	.1000	-2.128	-2.245
	.2250	-1.203	-1.290
	.4500	-.558	-.696
	.7500	-.179	-.301
	.8350	.024	-.090
	.8750	.027	.000
	.9250	.018	.036
	.9750	.042	.078
Lower	.0500	.806	.910
	.1500	.499	.484
	.4000	.215	.209
	.7000	.122	.176
	.8530	.030	.027
	.9250	-.101	.087
	.9750	-.042	.000

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.116		
	.0100	-4.149		
	.0250	-4.427		
	.0500	-3.669		
	.1000	-2.373		
	.2250	-1.170		
	.4500	-.672		
	.7500	-.185		
Lower	.0500	.824		
	.1500	.412		
	.4000	.096		
	.7000	.137		
Aileron				
Upper	.8050	-.182		
	.8150	-.149		
	.8350	-.101		
	.8750	-.060		
	.9250	-.030		
	.9750	-.003		
Lower	.8100	.167		
	.8340	.107		
	.9250	-.000		
	.9750	-.009		

TABLE X.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 20.1 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 2.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.430	-1.307
	.0100	-3.621	-4.063
	.0250	-3.773	-4.096
	.0500	-3.540	-3.982
	.1000	-2.090	-2.158
	.2250	-1.104	-1.134
	.4500	-.481	-.612
	.7500	-.128	-.206
	.8350	-.090	-.125
	.8750	-.036	-.122
	.9250	.000	-.078
	.9750	-.006	-.042
Lower	.0500	.836	.675
	.1500	.636	.454
	.4000	.191	.173
	.7000	.221	.101
	.8530	.131	.027
	.9250	-.081	-.078
	.9750	-.021	-.155

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.600	-.060	.296
	.0100	-4.212	-2.949	-1.815
	.0250	-4.084	-3.654	-2.009
	.0500	-3.382	-3.367	-1.851
	.1000	-2.251	-1.827	-1.260
	.2250	-1.158	-.973	-.752
	.4500	-.642	-.501	-.397
	.7500	-.251	-.299	-.427
Lower	.0500	.636	.585	.358
	.1500	.343	.272	.119
	.4000	.081	.054	.012
	.7000	.093	.161	-.018
Aileron				
Upper	.8050	-.197	-.263	-.412
	.8150	-.188	-.251	-.257
	.8350	-.173	-.197	-.621
	.8750	-.131	-.119	-.549
	.9250	-.093	-.027	-.379
	.9750	-.084	.024	-.215
Lower	.8100	.125	.257	-.039
	.8340	.087	.376	-.009
	.9250	-.054	.096	-.081
	.9750	-.072	-.027	-.158

TABLE XI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.5^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500		.945 .255 -.455 -.491 -.497 -.430 -.321 -.333	.939 .345 -.188 -.248 -.418 -.406 -.297 -.230
Lower	.0500 .1500 .4000 .7000		-.624 -.485 -.321 -.006	-.624 -.442 -.176 -.006
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750		-.370 -.309 -.273 -.236 -.085 .024	-.194 -.182 -.273 -.200 -.036 .055
Lower	.8100 .8340 .9250 .9750		.091 .212 .042 .067	.103 .079 .042 .048

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.636
	.0100		-1.170
	.0250		-1.752
	.0500		-1.752
	.1000		-1.364
	.2250		-.800
	.4500		-.661
	.7500		-.309
	.8350		-.127
	.8750		-.115
	.9250		-.139
	.9750		-.109
Lower	.0500		.158
	.1500		-.067
	.4000		-.133
	.7000		-.164
	.8530		-.079
	.9250		-.085
	.9750		-.061

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.770	.861	.945
	.0100	-.921	-.630	-.339
	.0250	-1.552	-1.412	-.800
	.0500	-1.491	-1.315	-.812
	.1000	-1.358	-.952	-.715
	.2250	-.727	-.642	-.570
	.4500	-.564	-.400	-.339
	.7500	-.382	-.388	-.230
Lower	.0500	.152	-.048	-.158
	.1500	-.085	-.158	-.242
	.4000	-.133	-.158	-.133
	.7000	.018	.018	-.024
Aileron				
Upper	.8050	-.206	-.321	-.176
	.8150	-.182	-.291	-.200
	.8350	-.115	-.273	-.339
	.8750	-.170	-.218	-.242
	.9250	-.042	-.085	-.115
	.9750	.067	.061	-.024
Lower	.8100	.067	.109	.109
	.8340	.085	.255	.091
	.9250	-.067	.055	-.030
	.9750	.030	.085	-.012

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.085
	.0100		-2.339
	.0250		-3.127
	.0500		-2.430
	.1000		-1.588
	.2250		-.867
	.4500		-.503
	.7500		-.333
	.8350		.012
	.8750		.006
	.9250		.079
	.9750		.115
Lower	.0500		.564
	.1500		-.006
	.4000		.067
	.7000		.109
	.8530		.091
	.9250		-.018
	.9750		.055

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.261	.545	.636
	.0100	-2.139	-1.509	-1.006
	.0250	-2.612	-2.267	-1.400
	.0500	-2.321	-2.115	-1.358
	.1000	-1.703	-1.345	-1.030
	.2250	-.873	-.830	-.709
	.4500	-.491	-.503	-.418
	.7500	-.109	-.388	-.345
Lower	.0500	.430	.321	.091
	.1500	.200	.042	-.091
	.4000	.006	-.085	-.109
	.7000	.024	.055	-.127
Aileron				
Upper	.8050	-.188	-.321	-.285
	.8150	-.139	-.315	-.285
	.8350	-.079	-.285	-.388
	.8750	-.042	-.200	-.327
	.9250	.030	-.103	-.212
	.9750	-.024	-.006	-.103
Lower	.8100	.200	.073	.085
	.8340	.194	.267	.030
	.9250	-.006	.048	-.109
	.9750	-.018	.036	-.097

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = 4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.624	-1.648
	.0100	-4.545	-5.145
	.0250	-4.418	-5.188
	.0500	-3.788	-4.333
	.1000	-2.230	-2.430
	.2250	-1.406	-1.224
	.4500	-.642	-.679
	.7500	-.242	-.206
	.8350	-.048	-.085
	.8750	-.018	-.079
	.9250	-.024	-.061
	.9750	.012	-.012
Lower	.0500	.879	.927
	.1500	.594	.545
	.4000	.194	.248
	.7000	.127	.097
	.8530	-.012	.073
	.9250	-.127	.012
	.9750	-.042	-.061

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.988	-.927	
	.0100	-4.988	-4.170	
	.0250	-5.097	-4.467	
	.0500	-4.200	-4.188	
	.1000	-2.485	-2.121	
	.2250	-1.248	-1.158	
	.4500	-.630	-.576	
	.7500	-.164	-.309	
Lower	.0500	.891	.800	
	.1500	.539		
	.4000	.200		
	.7000	.145		
Aileron				
Upper	.8050	-.121		
	.8150	-.133		
	.8350	-.103		
	.8750	-.048		
	.9250	-.024		
	.9750	-.012		
Lower	.8100	.139		
	.8340	.236		
	.9250	-.012		
	.9750	.030		

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = 6.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.733
	.0100		-5.236
	.0250		-5.339
	.0500		-4.358
	.1000		-2.370
	.2250		-1.152
	.4500		-.515
	.7500		-.242
	.8350		.006
	.8750		-.115
	.9250		-.085
	.9750		-.085
Lower	.0500		1.000
	.1500		.497
	.4000		.261
	.7000		.152
	.8530		-.091
	.9250		-.103
	.9750		-.152

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.224	-1.127	-.576
	.0100	-5.224	-4.455	-3.097
	.0250	-5.345	-4.745	-2.970
	.0500	-4.515	-4.436	-2.594
	.1000	-2.552	-2.188	-1.552
	.2250	-1.224	-1.152	-.879
	.4500	-.497	-.521	-.412
	.7500	-.061	-.212	-.370
Lower	.0500	.873	.873	.697
	.1500	.606	.515	.273
	.4000	.224	.200	.012
	.7000	.115	.121	-.158
Aileron				
Upper	.8050	-.224	-.158	-.352
	.8150	-.309	-.109	-.309
	.8350	-.085	-.097	-.388
	.8750	-.073	-.067	-.455
	.9250	.030	-.006	-.315
	.9750	-.061	.036	-.188
Lower	.8100	.115	.121	-.036
	.8340	.067	.285	.061
	.9250	-.055	.055	-.188
	.9750	-.127	.048	-.176

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 7.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.509	-2.048
	.0100	-4.212	-5.521
	.0250	-4.285	-5.630
	.0500	-2.921	-4.467
	.1000	-1.988	-2.588
	.2250	-.709	-1.267
			-.721
	.7500	-.212	-.339
	.8350	-.515	-.121
	.8750	-.479	-.188
	.9250	-.479	-.230
	.9750	-.515	-.242
Lower	.0500	.642	.897
	.1500	.139	.600
	.4000	.212	.206
	.7000	.115	.073
	.8530	-.230	-.042
	.9250	-.533	-.030
	.9750	-.491	-.188

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.776	-1.503	
	.0100	-5.788	-4.885	
	.0250	-5.800	-5.091	
	.0500	-4.709	-4.812	
	.1000	-2.661	-2.364	
	.2250	-1.412	-1.176	
	.4500	-.703	-.642	
	.7500	-.212	-.303	
Lower	.0500	.818	.836	
	.1500	.533	.521	
	.4000	.176	.073	
	.7000	.085	.061	
Aileron				
Upper	.8050	-.273	-.309	
	.8150	-.139		
	.8350	-.176		
	.8750	-.176		
	.9250	-.158		
	.9750	-.103		
Lower	.8100	.024		
	.8340	.127		
	.9250	-.097		
	.9750	-.024		

TABLE XI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 8.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.952
	.0100		-5.339
	.0250		-5.309
	.0500		-4.261
	.1000		-2.303
	.2250		-1.212
	.4500		-.642
	.7500		-.309
	.8350		-.194
	.8750		-.224
	.9250		-.279
	.9750		-.267
Lower	.0500		.909
	.1500		.648
	.4000		.127
	.7000		.036
	.8530		-.042
	.9250		-.103
	.9750		-.097

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.679	-1.436	-.800
	.0100	-5.745	-4.806	-3.303
	.0250	-5.636	-4.982	-3.073
	.0500	-4.582	-4.655	-2.655
	.1000	-2.576	-2.212	-1.527
	.2250	-1.309	-1.133	-.855
	.4500	-.582	-.491	-.424
	.7500	-.127	-.158	-.412
Lower	.0500	.909	.885	.691
	.1500	.588	.552	.291
	.4000	.121	.230	-.012
	.7000	.055	.145	-.176
Aileron				
Upper	.8050	-.176	-.127	-.406
	.8150	-.121	-.103	-.291
	.8350	-.133	-.079	-.400
	.8750	-.164	-.061	-.461
	.9250	-.121	-.006	-.333
	.9750	-.067	.012	-.182
Lower	.8100	.073	.097	-.085
	.8340	.152	.291	.030
	.9250	-.091		-.188
	.9750	-.061		-.182

TABLE XI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 9.9 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 9.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-2.145
	.0100		-5.564
	.0250		-5.455
	.0500		-4.485
	.1000		-2.497
	.2250		-1.073
	.4500		-.667
	.7500		-.206
	.8350		-.188
	.8750		-.315
	.9250		-.436
	.9750		-.218
Lower	.0500		.861
	.1500		.564
	.4000		.121
	.7000		.073
	.8530		-.085
	.9250		-.133
	.9750		-.200

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.848	-1.564	-1.006
	.0100	-5.897	-5.018	-3.527
	.0250	-6.024	-5.176	-3.291
	.0500	-4.703	-4.873	-2.764
	.1000	-2.570	-2.303	-1.648
	.2250	-1.321	-1.218	-.909
	.4500	-.715	-.558	-.442
	.7500	-.127	-.236	-.497
Lower	.0500	.958	.879	.667
	.1500	.606	.545	.309
	.4000	.206	.200	-.036
	.7000	.109	.085	-.236
Aileron				
Upper	.8050	-.164	-.152	-.455
	.8150	-.170	-.145	-.236
	.8350	-.133	-.145	-.406
	.8750	-.121	-.109	-.485
	.9250	.024	-.085	-.315
	.9750	-.079	-.048	-.188
Lower	.8100	.073	.055	-.176
	.8340	.164	.230	-.055
	.9250	-.109	.000	-.206
	.9750	-.012	.000	-.200

TABLE XII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.948
	.0100		-.104
	.0250		-.860
	.0500		-.762
	.1000		-.819
	.2250		-.617
	.4500		-.352
	.7500		-.425
	.8350		-.026
	.8750		-.166
	.9250		-.010
	.9750		.073
Lower	.0500		-.280
	.1500		-.321
	.4000		-.311
	.7000		-.088
	.8530		-.067
	.9250		-.073
	.9750		-.005

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.938	.984	.974
	.0100	.031	.228	.311
	.0250	-.611	-.508	-.218
	.0500	-.570	-.565	-.301
	.1000	-.746	-.570	-.446
	.2250	-.497	-.425	-.425
	.4500	-.497	-.290	-.295
	.7500	-.280	-.301	-.192
Lower	.0500	-.534	-.611	-.565
	.1500	-.368	-.466	-.404
	.4000	-.290	-.285	-.166
	.7000	-.031	-.057	-.041
Aileron				
Upper	.8050	-.347	-.264	-.104
	.8150	-.259	-.259	-.181
	.8350	-.124	-.249	-.269
	.8750	-.047	-.212	-.155
	.9250	.010	-.104	-.041
	.9750	.016	.036	.073
Lower	.8100	-.021	.057	.104
	.8340	.073	.197	.078
	.9250	.016	.036	.036
	.9750	.104	.088	.062

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.699
	.0100		-1.150
	.0250		-1.917
	.0500		-1.756
	.1000		-1.332
	.2250		-.772
	.4500		-.503
	.7500		-.430
	.8350		-.062
	.8750		-.057
	.9250		-.021
	.9750		.104
Lower	.0500		.171
	.1500		-.083
	.4000		-.119
	.7000		-.067
	.8530		.041
	.9250		.010
	.9750		-.005

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.756	.943	.974
	.0100	-.938	-.637	-.290
	.0250	-1.534	-1.425	-.777
	.0500	-1.560	-1.358	-.829
	.1000	-1.181	-.964	-.736
	.2250	-.767	-.674	-.591
	.4500	-.513	-.466	-.352
	.7500	-.280	-.399	-.285
Lower	.0500	.109	-.078	-.218
	.1500	-.062	-.181	-.280
	.4000	-.161	-.181	-.155
	.7000	-.021	-.000	-.057
Aileron				
Upper	.8050	-.249	-.363	-.223
	.8150	-.249	-.326	-.218
	.8350	-.171	-.301	-.383
	.8750	-.109	-.233	-.259
	.9250	.010	-.083	-.145
	.9750	.026	.031	-.031
Lower	.8100	.078	.098	.098
	.8340	.078	.254	.036
	.9250	-.021	.036	-.052
	.9750	.041	.062	-.021

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -2.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.285
	.0100		-2.093
	.0250		-2.803
	.0500		-2.368
	.1000		-1.570
	.2250		-.886
	.4500		-.513
	.7500		-.233
	.8350		-.093
	.8750		-.016
	.9250		.036
	.9750		.088
Lower	.0500		.601
	.1500		.207
	.4000		.104
	.7000		.041
	.8530		-.036
	.9250		-.005
	.9750		-.005

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.399	.689	.813
	.0100	-1.886	-1.404	-.824
	.0250	-2.539	-2.244	-1.244
	.0500	-2.218	-2.047	-1.223
	.1000	-1.637	-1.321	-.990
	.2250	-.865	-.793	-.684
	.4500	-.585	-.477	-.378
	.7500	-.150	-.373	-.306
Lower	.0500	.456	.259	.036
	.1500	.207	-.000	-.140
	.4000	.104	-.098	-.104
	.7000	.026	.016	-.067
Aileron				
Upper	.8050	-.181	-.358	-.238
	.8150	-.192	-.316	-.249
	.8350	-.150	-.280	-.373
	.8750	-.098	-.223	-.311
	.9250	-.036	-.093	-.155
	.9750	.062	.021	-.047
Lower	.8100	.062	.083	.093
	.8340	.057	.306	.067
	.9250	.057	.047	-.078
	.9750	.057	.047	-.052

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.171	-.109
	.0100	-2.736	-3.098
	.0250	-3.109	-3.642
	.0500	-2.720	-3.223
	.1000	-1.736	-1.938
	.2250	-1.176	-1.036
	.4500	-.663	-.741
	.7500	-.301	-.440
	.8350	-.088	-.171
	.8750	-.073	-.031
	.9250	-.078	-.031
	.9750	-.161	.098
Lower	.0500	.446	.684
	.1500	.264	.228
	.4000	.171	.078
	.7000	.041	.067
	.8530	-.073	.041
	.9250	-.114	-.047
	.9750	-.062	-.062

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.135	.409	.585
	.0100	-2.684	-2.000	-1.326
	.0250	-3.140	-2.803	-1.694
	.0500	-2.788	-2.549	-1.575
	.1000	-1.938	-1.539	-1.135
	.2250	-1.036	-.896	-.736
	.4500	-.684	-.503	-.378
	.7500	-.259	-.337	-.332
Lower	.0500	.611	.482	.238
	.1500	.207	.161	.000
	.4000	.005	-.010	-.078
	.7000	.052	.057	-.109
Aileron				
Upper	.8050	-.207	-.264	-.264
	.8150	-.187	-.238	-.269
	.8350	-.135	-.228	-.399
	.8750	-.073	-.176	-.332
	.9250	-.005	-.057	-.207
	.9750	.026	.047	-.098
Lower	.8100	.114	.114	.062
	.8340	.181	.280	.052
	.9250	-.021	.041	-.130
	.9750	.041	.057	-.093

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(e) $\alpha_f = -0.1^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.580
	.0100		-3.674
	.0250		-4.155
	.0500		-3.637
	.1000		-2.135
	.2250		-1.223
	.4500		-.819
	.7500		-.549
	.8350		-.228
	.8750		-.233
	.9250		-.135
	.9750		-.098
Lower	.0500		.627
	.1500		.181
	.4000		.047
	.7000		-.041
	.8530		-.088
	.9250		-.114
	.9750		-.114

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.570	.166	.363
	.0100	-3.269	-2.518	-1.725
	.0250	-3.715	-3.218	-1.995
	.0500	-3.207	-2.938	-1.865
	.1000	-2.088	-1.710	-1.264
	.2250	-1.135	-.995	-.788
	.4500	-.710	-.539	-.394
	.7500	-.269	-.306	-.373
Lower	.0500	.684	.601	.383
	.1500	.306	.238	.078
	.4000	.047	.000	-.067
	.7000	.016	.073	-.135
Aileron				
Upper	.8050	-.187	-.290	-.280
	.8150	-.223	-.264	-.290
	.8350	-.161	-.238	-.440
	.8750	-.073	-.166	-.389
	.9250	-.036	-.047	-.254
	.9750	.041	.026	-.124
Lower	.8100	.130	.135	.026
	.8340	.119	.337	.021
	.9250	-.021	.031	-.166
	.9750	.067	.041	-.145

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(f) $\alpha_f = 0.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.575	-.658
	.0100	-3.342	-3.933
	.0250	-3.617	-4.218
	.0500	-3.259	-3.648
	.1000	-1.984	-2.176
	.2250	-1.259	-1.124
	.4500	-.699	-.653
	.7500	-.383	-.244
	.8350	-.041	-.078
	.8750	-.041	-.016
	.9250	-.047	-.031
	.9750	-.016	.062
Lower	.0500	.731	.876
	.1500	.451	.399
	.4000	.223	.135
	.7000	.067	.119
	.8530	-.036	.073
	.9250	-.093	.073
	.9750	-.067	.047

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.839	.000	.233
	.0100	-3.668	-2.850	-1.979
	.0250	-4.010	-3.544	-2.166
	.0500	-3.394	-3.192	-2.000
	.1000	-2.161	-1.803	-1.316
	.2250	-1.155	-1.021	-.803
	.4500	-.689	-.565	-.415
	.7500	-.259	-.311	-.352
Lower	.0500	.793	.658	.461
	.1500	.394	.321	.104
	.4000	.104	.041	-.057
	.7000	.088	.073	-.135
Aileron				
Upper	.8050	-.181	-.238	-.295
	.8150	-.171	-.218	-.285
	.8350	-.140	-.202	-.415
	.8750	-.083	-.150	-.394
	.9250	-.041	-.041	-.249
	.9750	-.010	.010	-.124
Lower	.8100	.078	.104	.016
	.8340	.150	.311	.031
	.9250	-.041	.041	-.166
	.9750	-.021	.041	-.119

TABLE XII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(g) $\alpha_f = 1.3^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.031	-1.202
	.0100	-4.031	-4.699
	.0250	-4.155	-4.829
	.0500	-3.715	-4.109
	.1000	-2.389	-2.358
	.2250	-1.373	-1.259
	.4500	-.699	-.710
	.7500	-.295	-.192
	.8350	-.109	-.088
	.8750	-.052	-.062
	.9250	-.067	.005
	.9750	-.031	.031
Lower	.0500	.850	.860
	.1500	.425	.508
	.4000	.212	.098
	.7000	.083	.062
	.8530	-.021	.067
	.9250	-.145	.021
	.9750	-.104	-.047

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.363	-.383	-.067
	.0100	-4.352	-3.435	-2.451
	.0250	-4.575	-4.047	-2.544
	.0500	-3.907	-3.710	-2.311
	.1000	-2.363	-2.021	-1.477
	.2250	-1.244	-1.093	-.876
	.4500	-.725	-.580	-.466
	.7500	-.187	-.275	-.389
Lower	.0500	.829	.720	.534
	.1500	.492	.337	.130
	.4000	.140	.083	-.078
	.7000	.073	.078	-.192
Aileron				
Upper	.8050	-.166	-.244	-.342
	.8150	-.145	-.238	-.342
	.8350	-.114	-.228	-.394
	.8750	-.067	-.171	-.435
	.9250	-.031	-.067	-.342
	.9750	.021	-.016	-.171
Lower	.8100	.098	.078	-.057
	.8340	.161	.275	.005
	.9250	-.021	-.021	-.233
	.9750	-.000	-.010	-.176

TABLE XII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 11.6 lb/sq ft; belly plate off; p = 4.8 lb/sq in.; forward guy cables, lightly tightened]

(h) $\alpha_f = 1.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.373	-1.534
	.0100	-4.580	-5.218
	.0250	-4.420	-5.275
	.0500	-3.902	-4.404
	.1000	-2.404	-2.477
	.2250	-1.352	-1.337
	.4500	-.689	-.627
	.7500	-.280	-.166
	.8350	-.093	-.130
	.8750	-.057	-.093
	.9250	-.031	-.031
	.9750	-.010	-.031
Lower	.0500	.741	.896
	.1500	.596	.565
	.4000	.218	.228
	.7000	.166	.057
	.8530	.016	.083
	.9250	-.166	.010
	.9750	-.135	-.031

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.161	-.876	-.342
	.0100	-5.212	-4.301	-2.860
	.0250	-5.358	-4.705	-2.834
	.0500	-4.316	-4.383	-2.528
	.1000	-2.601	-2.202	-1.539
	.2250	-1.337	-1.150	-.881
	.4500	-.674	-.549	-.435
	.7500	-.181	-.212	-.332
Lower	.0500	.886	.855	.637
	.1500	.549	.482	.228
	.4000	.187	.135	-.021
	.7000	.062	.073	-.202
Aileron				
Upper	.8050	-.166	-.145	-.259
	.8150	-.155	-.114	-.264
	.8350	-.114	-.104	-.358
	.8750	-.062	-.073	-.420
	.9250	-.005	-.021	-.342
	.9750	.016	.021	-.218
Lower	.8100	.052	.078	-.083
	.8340	.114	.269	-.036
	.9250	-.052	.005	-.228
	.9750	-.026	.021	-.161

TABLE XIII

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate off; p = 2.0 lb/sq in.; forward guy cables, lightly tightened]

(a) $\alpha_f = -8.6^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:					
		0.1621	0.2895	0.5000	0.6800	0.8920	
Wing							
Upper	.0000		.868	.553	.825	.904	
	.0100		-1.000	-1.123	-.842	-.579	
	.0250		-1.912	-1.816	-1.684	-1.000	
	.0500		-1.807	-1.632	-1.579	-.982	
	.1000		-1.368	-1.219	-1.044	-.789	
	.2250		-.877	-.789	-.658	-.553	
	.4500		-.526	-.614	-.351	-.289	
	.7500		-.175	-.053	-.044	.096	
	.8350		-.088				
	.8750		-.123				
	.9250		-.123				
	.9750		-.044				
Lower	.0500		.132	.070	.088	-.044	
	.1500		-.246	-.175	-.140	-.193	
	.4000		-.193	-.289	-.149	-.149	
	.7000		-.114	-.281	-.254	-.184	
	.8530		-.053				
	.9250		-.053				
	.9750		-.053				
Aileron							
Upper	.8050			-.228	.088	.114	
	.8150			-.123	.079	.140	
	.8350			-.079	.079	.132	
	.8750			-.035	.018	.009	
	.9250			-.000	.018	.070	
	.9750			-.123	.053	.061	
Lower	.8100			-.289	-.246	-.316	
	.8340			-.149	-.123	-.175	
	.9250			.009	-.053	-.061	
	.9750			-.018	.018	.018	

TABLE XIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate off; p = 2.0 lb/sq in.; forward guy cables, lightly tightened]

(b) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.009
	.0100		-2.298
	.0250		-3.044
	.0500		-2.640
	.1000		-1.860
	.2250		-.912
	.4500		-.702
	.7500		-.518
	.8350		-.158
	.8750		-.132
	.9250		-.035
	.9750		.009
Lower	.0500		.395
	.1500		.026
	.4000		-.114
	.7000		-.105
	.8530		-.167
	.9250		-.096
	.9750		-.114

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.123	.509	.667
	.0100	-2.202	-1.807	-1.184
	.0250	-2.789	-2.544	-1.544
	.0500	-2.395	-2.351	-1.439
	.1000	-1.658	-1.439	-1.018
	.2250	-1.044	-.807	-.658
	.4500	-.649	-.412	-.298
	.7500	-.219	-.070	.018
Lower	.0500	.465	.395	.246
	.1500	.009	.096	-.035
	.4000	-.184	-.044	-.105
	.7000	-.184	-.140	-.175
Aileron				
Upper	.8050	-.237	.026	.088
	.8150	-.254	.009	.096
	.8350	-.132	.009	.061
	.8750	-.105	-.053	-.044
	.9250	-.061	-.026	-.018
	.9750	-.035	.070	-.018
Lower	.8100	-.237	-.202	-.246
	.8340	-.123	-.018	-.219
	.9250	-.026	-.070	-.114
	.9750	-.009	.053	-.035

TABLE XIII.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; $q_\infty = 6.9$ lb/sq ft; belly plate off; p = 2.0 lb/sq in.; forward guy cables, lightly tightened]

(c) $\alpha_f = -3.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.123	-.263
	.0100	-2.439	-3.026
	.0250	-2.807	-3.570
	.0500	-2.553	-3.175
	.1000	-1.772	-1.781
	.2250	-1.351	-.956
	.4500	-.614	-.640
	.7500	-.123	-.456
	.8350	-.061	-.061
	.8750	-.035	.026
	.9250	-.132	.070
	.9750	-.026	.026
Lower	.0500	.377	.632
	.1500	.211	.325
	.4000	.149	.096
	.7000	.044	.053
	.8530	-.175	-.018
	.9250	-.219	-.009
	.9750	-.167	-.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.570	.061	.360
	.0100	-2.956	-2.535	-1.640
	.0250	-3.482	-3.149	-1.886
	.0500	-2.947	-2.868	-1.728
	.1000	-1.939	-1.623	-1.184
	.2250	-1.061	-.851	-.746
	.4500	-.518	-.421	-.333
	.7500	-.149	-.079	-.053
Lower	.0500	.675	.561	.351
	.1500	.237	.254	.070
	.4000	.009	.035	-.088
	.7000	-.018	-.114	-.228
Aileron				
Upper	.8050	-.026	-.009	.026
	.8150	-.070	.018	.096
	.8350	.044	.026	.053
	.8750	.044	-.018	-.061
	.9250	.053	.026	-.044
	.9750	.088	.061	-.053
Lower	.8100	-.193	-.140	-.219
	.8340	.018	.035	-.289
	.9250	.018	-.061	-.184
	.9750	.026	.053	-.096

TABLE XIII.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 6.9 lb/sq ft; belly plate off; p = 2.0 lb/sq in.; forward guy cables, lightly tightened]

(d) $\alpha_f = -1.1^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper			
Lower			

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.316 -4.289 -4.561 -3.675 -2.482 -1.114 -.561 -.132	-.719 -3.991 -4.342 -4.070 -2.044 -1.009 -.465 -.061	-.333 -2.772 -2.711 -2.500 -1.465 -.825 -.377 -.061
Lower	.0500 .1500 .4000 .7000	.895 .377 .228 .132	.833 .447 .140 -.070	.640 .219 -.035 -.254
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	.018 -.088 -.044 .026 -.009 .061	.026 .035 .044 .053 .044 .114	.026 .009 .061 -.140 -.193 -.184
Lower	.8100 .8340 .9250 .9750	.000 -.088 -.044 .053	-.140 .088 -.044 .079	-.325 -.105 -.254 -.123

TABLE XIV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.692	.769
	.0100	-.889	-.855
	.0250	-1.556	-1.581
	.0500	-1.487	-1.504
	.1000	-1.393	-1.231
	.2250	-.821	-.795
	.4500	-.479	-.427
	.7500	-.368	-.581
	.8350	-.171	-.077
	.8750	-.043	-.060
	.9250	-.043	-.085
	.9750	-.094	.034
Lower	.0500	-.043	.009
	.1500	-.291	-.214
	.4000	-.214	-.316
	.7000	-.103	-.137
	.8530	-.248	-.068
	.9250	-.137	-.034
	.9750	-.145	.017

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.769	.932	.940
	.0100	-.838	-.564	-.274
	.0250	-1.513	-1.265	-.718
	.0500	-1.308	-1.214	-.761
	.1000	-1.197	-.846	-.641
	.2250	-.752	-.556	-.496
	.4500	-.538	-.342	-.256
	.7500	-.231	-.145	-.068
Lower	.0500	-.009	-.051	-.179
	.1500	-.214	-.188	-.248
	.4000	-.256	-.188	-.154
	.7000	-.128	-.085	-.068
Aileron				
Upper	.8050	-.162	-.094	-.017
	.8150	-.154	-.085	.017
	.8350	-.085	-.103	-.085
	.8750	-.085	-.094	-.103
	.9250	-.000	-.034	-.026
	.9750	-.009	.051	.026
Lower	.8100	-.137	-.034	.026
	.8340	-.026	.248	.026
	.9250	.051	.017	-.043
	.9750	.043	.077	.043

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = -15^o$; cable configuration 2-5-8; $q_\infty = 7.0$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.547	.402
	.0100	-1.094	-1.222
	.0250	-1.641	-1.966
	.0500	-1.436	-1.761
	.1000	-1.154	-1.308
	.2250	-.769	-.821
	.4500	-.530	-.556
	.7500	-.359	-.556
	.8350	-.188	-.188
	.8750	-.197	-.103
	.9250	-.239	-.103
	.9750	-.103	-.026
Lower	.0500	-.034	.068
	.1500	.034	-.308
	.4000	-.060	-.308
	.7000	-.085	-.325
	.8530	-.299	-.094
	.9250	-.325	-.162
	.9750	-.179	-.137

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.607	.915	.915
	.0100	-.940	-.632	-.376
	.0250	-1.641	-1.350	-.846
	.0500	-1.410	-1.316	-.829
	.1000	-1.214	-.863	-.692
	.2250	-.675	-.547	-.470
	.4500	-.530	-.274	-.205
	.7500	-.009	.145	.197
Lower	.0500	-.000	-.026	-.120
	.1500	-.214	-.179	-.248
	.4000	-.274	-.248	-.162
	.7000	-.291	-.299	-.222
Aileron				
Upper	.8050	-.282	.145	.248
	.8150	.017	.179	.282
	.8350	.162	.197	.316
	.8750	.094	.111	.162
	.9250	-.034	.068	.094
	.9750	-.171	-.017	.051
Lower	.8100	-.470	-.410	-.385
	.8340	-.162	-.222	-.274
	.9250	-.171	-.103	-.085
	.9750	-.077	-.034	.043

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 7.0$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.581	.667
	.0100	-.966	-.829
	.0250	-1.709	-1.692
	.0500	-1.547	-1.513
	.1000	-1.154	-1.248
	.2250	-.692	-.863
	.4500	-.504	-.556
	.7500	-.376	-.462
	.8350	-.154	-.179
	.8750	-.179	-.026
	.9250	-.034	-.051
	.9750	-.291	-.017
Lower	.0500	-.009	.154
	.1500	-.197	-.214
	.4000	-.308	-.239
	.7000	-.308	-.068
	.8530	-.282	-.051
	.9250	-.256	-.051
	.9750	-.137	-.145

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.735	.932	.906
	.0100	-.761	-.444	-.137
	.0250	-1.444	-1.154	-.675
	.0500	-1.342	-1.137	-.709
	.1000	-1.179	-.855	-.658
	.2250	-.769	-.598	-.530
	.4500	-.615	-.427	-.342
	.7500	-.419	-.453	-.427
Lower	.0500	.009	-.120	-.256
	.1500	-.179	-.248	-.248
	.4000	-.128	-.120	-.120
	.7000	.077	.154	.068
Aileron				
Upper	.8050	-.427	-.479	-.410
	.8150	-.376	-.419	-.427
	.8350	-.299	-.291	-.462
	.8750	-.214	-.188	-.256
	.9250	-.162	-.154	-.188
	.9750	-.111	-.171	-.188
Lower	.8100	.162	.308	.274
	.8340	.188	.308	.239
	.9250	-.043	.111	.085
	.9750	-.043	-.034	-.060

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.333	.376
	.0100	-1.932	-2.120
	.0250	-2.308	-2.735
	.0500	-2.085	-2.299
	.1000	-1.436	-1.624
	.2250	-.991	-.889
	.4500	-.598	-.504
	.7500	-.325	-.359
	.8350	-.051	-.068
	.8750	-.060	-.051
	.9250	-.060	-.051
	.9750	-.051	.103
Lower	.0500	.359	.427
	.1500	.111	-.009
	.4000	-.026	-.103
	.7000	.068	-.154
	.8530	-.205	-.009
	.9250	-.145	-.000
	.9750	.017	-.017

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.239	.573	.675
	.0100	-2.026	-1.470	-1.000
	.0250	-2.410	-2.162	-1.393
	.0500	-2.239	-1.983	-1.308
	.1000	-1.581	-1.188	-.923
	.2250	-.897	-.709	-.590
	.4500	-.538	-.385	-.291
	.7500	-.239	-.111	-.111
Lower	.0500	.376	.333	.179
	.1500	.085	.060	-.043
	.4000	-.077	-.068	-.103
	.7000	-.103	-.051	-.094
Aileron				
Upper	.8050	-.145	-.077	-.034
	.8150	-.154	-.077	-.043
	.8350	-.068	-.094	-.085
	.8750	-.060	-.077	-.128
	.9250	-.017	.000	-.051
	.9750	-.017	.051	-.009
Lower	.8100	-.103	-.009	-.043
	.8340	.000	.222	-.000
	.9250	-.060	.000	-.085
	.9750	.017	.103	-.017

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = -15^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.188	.111	-2.205	.444	.607
	.0100	-2.026	-2.274	-2.581	-1.709	-1.145
	.0250	-2.479	-2.795	-2.342	-2.316	-1.453
	.0500	-2.103	-2.547	-1.658	-2.120	-1.385
	.1000	-1.573	-1.564	-.880	-1.248	-.932
	.2250	-1.068	-.940	-.470	-.709	-.556
	.4500	-.556	-.462	-.043	-.291	-.205
	.7500	-.248	-.342	.453	.137	.214
	.8350	-.145	-.043			
	.8750	-.017	-.017			
	.9250	.060	.000			
	.9750	.000	.085			
Lower	.0500	.291	.547	.085	.427	.239
	.1500	.034	.111	-.154	.103	-.034
	.4000	-.154	-.051	-.171	-.103	-.120
	.7000	-.043	-.051	-.248	-.282	-.214
	.8530	-.000	-.026			
	.9250	-.137	-.017			
	.9750	-.068	.000			
Aileron						
Upper	.8050	-.017	.154	.214		
	.8150	.077	.188	.205		
	.8350	.043	.197	.239		
	.8750	-.017	.137	.188		
	.9250	-.085	.094	.111		
	.9750	-.291	.051	.034		
Lower	.8100	-.120	-.419	-.393		
	.8340	-.085	-.205	-.282		
	.9250	-.034	-.060	-.094		
	.9750	-.009	-.009	-.017		

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.291	.051
	.0100	-1.880	-2.145
	.0250	-2.368	-2.786
	.0500	-2.077	-2.444
	.1000	-1.504	-1.607
	.2250	-.957	-.991
	.4500	-.513	-.624
	.7500	-.274	-.427
	.8350	-.000	-.034
	.8750	-.085	.009
	.9250	-.026	.009
	.9750	-.034	.017
Lower	.0500	.274	.496
	.1500	.154	.051
	.4000	-.085	-.026
	.7000	-.017	-.034
	.8530	.000	-.026
	.9250	-.103	.026
	.9750	-.068	-.009

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.128	.530	.709
	.0100	-2.120	-1.547	-.966
	.0250	-2.487	-2.265	-1.385
	.0500	-2.359	-2.043	-1.325
	.1000	-1.615	-1.308	-.966
	.2250	-1.000	-.769	-.675
	.4500	-.675	-.453	-.376
	.7500	-.436	-.385	-.487
Lower	.0500	.427	.342	.179
	.1500	.068	.051	-.043
	.4000	.017	-.034	-.077
	.7000	.120	.222	.043
Aileron				
Upper	.8050	-.402	-.350	-.453
	.8150	-.350	-.333	-.436
	.8350	-.333	-.231	-.444
	.8750	-.274	-.162	-.299
	.9250	-.239	-.137	-.214
	.9750	-.137	-.137	-.239
Lower	.8100	.205	.385	.188
	.8340	.231	.333	.205
	.9250	-.034	.145	-.026
	.9750	-.034	.017	-.154

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 2.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.632
	.0100		-3.880
	.0250		-4.068
	.0500		-3.556
	.1000		-2.205
	.2250		-1.239
	.4500		-.667
	.7500		-.436
	.8350		-.034
	.8750		.034
	.9250		.111
	.9750		.051
Lower	.0500		.821
	.1500		.564
	.4000		.274
	.7000		.103
	.8530		-.017
	.9250		-.068
	.9750		-.026

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.718	-.051	.256
	.0100	-3.410	-2.692	-1.838
	.0250	-3.812	-3.231	-2.034
	.0500	-3.282	-2.906	-1.846
	.1000	-2.077	-1.632	-1.248
	.2250	-1.231	-.923	-.735
	.4500	-.667	-.453	-.359
	.7500	-.171	-.145	-.179
Lower	.0500	.624	.641	.419
	.1500	.239	.239	.077
	.4000	.060	.026	-.068
	.7000	-.051	-.051	-.154
Aileron				
Upper	.8050	-.111	-.077	-.137
	.8150	-.085	-.085	-.111
	.8350	-.103	-.051	-.205
	.8750	-.060	-.051	-.265
	.9250	.000	-.017	-.205
	.9750	.077	.043	-.094
Lower	.8100	-.017	.026	-.111
	.8340	-.009	.239	-.145
	.9250	-.026	.034	-.137
	.9750	.094	.077	-.111

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) α_f = 2.9°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.778	-.735
	.0100	-3.205	-3.632
	.0250	-3.573	-4.017
	.0500	-3.085	-3.615
	.1000	-2.077	-1.983
	.2250	-1.179	-1.026
	.4500	-.564	-.564
	.7500	-.231	-.274
	.8350	-.060	-.085
	.8750	-.145	-.009
	.9250	-.171	.000
	.9750	-.128	.043
Lower	.0500	.778	.829
	.1500	.256	.385
	.4000	.000	.145
	.7000	.085	.043
	.8530	.009	.000
	.9250	-.171	-.026
	.9750	-.077	-.034

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.838	-.248	.111
	.0100	-3.513	-2.991	-2.103
	.0250	-3.829	-3.419	-2.239
	.0500	-3.274	-3.060	-2.017
	.1000	-2.034	-1.641	-1.231
	.2250	-1.017	-.863	-.675
	.4500	-.521	-.325	-.222
	.7500	-.077	.103	.171
Lower	.0500	.718	.718	.513
	.1500	.385	.350	.137
	.4000	.026	-.009	-.077
	.7000	-.103	-.231	-.282
Aileron				
Upper	.8050	-.162	.120	.162
	.8150	-.009	.137	.179
	.8350	.034	.137	.248
	.8750	.060	.120	.162
	.9250	.060	.094	.077
	.9750	.034	.068	-.026
Lower	.8100	-.256	-.479	-.462
	.8340	-.034	-.162	-.188
	.9250	-.034	-.077	-.145
	.9750	.000	.017	-.077

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 7.0$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-.786 -3.684 -4.094 -3.521 -2.188 -1.470 -.838 -.658 -.197 -.188 -.077 -.077
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.564 .179 -.103 -.256 -.239 -.051 .017

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-.530 -3.350 -3.624 -3.128 -2.162 -1.197 -.624 -.410	.068 -2.419 -3.034 -2.675 -1.573 -.957 -.496 -.359	.393 -1.573 -1.863 -1.718 -1.171 -.735 -.410 -.564
Lower	.0500 .1500 .4000 .7000	.709 .239 .043 .171	.573 .265 .103 .248	.359 .043 -.060 .000
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.350 -.265 -.265 -.205 -.128 -.051	-.308 -.265 -.188 -.145 -.120 -.120	-.538 -.496 -.573 -.359 -.291 -.316
Lower	.8100 .8340 .9250 .9750	.410 .350 .137 .009	.385 .333 .154 .017	.154 .120 -.077 -.205

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 7.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.915
	.0100		-5.479
	.0250		-5.299
	.0500		-4.530
	.1000		-2.650
	.2250		-1.393
	.4500		-.692
	.7500		-.154
	.8350		-.043
	.8750		-.051
	.9250		-.103
	.9750		-.000
Lower	.0500		.915
	.1500		.615
	.4000		.239
	.7000		.051
	.8530		.009
	.9250		-.103
	.9750		-.103

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.103	-.949	-.393
	.0100	-5.162	-4.179	-2.915
	.0250	-5.248	-4.342	-2.821
	.0500	-4.188	-4.060	-2.538
	.1000	-2.504	-1.940	-1.376
	.2250	-1.231	-.949	-.744
	.4500	-.632	-.359	-.265
	.7500	-.120	-.026	-.162
Lower	.0500	.821	1.017	.726
	.1500	.496	.658	.359
	.4000	.214	.299	.068
	.7000	.026	.197	-.111
Aileron				
Upper	.8050	.068	.077	-.094
	.8150	.000	.094	-.103
	.8350	-.111	.077	-.179
	.8750	-.068	.094	-.308
	.9250	-.103	.137	-.256
	.9750	-.026	.162	-.111
Lower	.8100	-.026	.162	-.026
	.8340	-.026	.436	.017
	.9250	-.145	.128	-.120
	.9750	-.034	.154	-.077

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = -15^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) $\alpha_f = 7.2^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.137	-1.932
	.0100	-5.060	-5.692
	.0250	-4.855	-5.444
	.0500	-4.103	-4.624
	.1000	-2.342	-2.530
	.2250	-1.299	-1.282
	.4500	-.624	-.547
	.7500	-.188	-.145
	.8350	.017	-.103
	.8750	.060	-.068
	.9250	.034	.000
	.9750	.034	.017
Lower	.0500	.991	.983
	.1500	.564	.632
	.4000	.265	.316
	.7000	.256	.205
	.8530	.077	.034
	.9250	-.085	.017
	.9750	.103	-.085

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.214	-1.316	-.692
	.0100	-5.376	-4.538	-3.222
	.0250	-5.419	-4.615	-3.060
	.0500	-4.470	-4.359	-2.701
	.1000	-2.641	-2.051	-1.496
	.2250	-1.299	-.983	-.752
	.4500	-.504	-.308	-.256
	.7500	-.145	.060	.137
Lower	.0500	.829	.923	.735
	.1500	.547	.547	.325
	.4000	.188	.171	-.009
	.7000	-.068	-.145	-.316
Aileron				
Upper	.8050	-.188	-.000	.231
	.8150	-.085	.051	.179
	.8350	-.060	.051	.402
	.8750	.000	.068	.171
	.9250	.017	.103	.051
	.9750	-.017	.120	-.068
Lower	.8100	-.222	-.410	-.427
	.8340	-.017	-.094	-.197
	.9250	-.017	-.034	-.231
	.9750	.017	.000	-.085

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 7.0$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(*l*) $\alpha_f = 7.1^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-2.017	-1.675
	.0100	-4.795	-5.137
	.0250	-4.658	-5.265
	.0500	-4.000	-4.615
	.1000	-2.316	-2.521
	.2250	-1.530	-1.402
	.4500	-.581	-.718
	.7500	-.316	-.333
	.8350	-.094	-.137
	.8750	-.094	-.077
	.9250	-.068	-.051
	.9750	-.068	-.017
Lower	.0500	1.017	.923
	.1500	.598	.513
	.4000	.188	.197
	.7000	.145	.085
	.8530	.094	.060
	.9250	-.111	.000
	.9750	-.017	-.068

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.932	-.966	-.368
	.0100	-4.940	-4.171	-2.872
	.0250	-5.068	-4.479	-2.846
	.0500	-4.274	-4.162	-2.564
	.1000	-2.504	-2.137	-1.504
	.2250	-1.325	-1.162	-.872
	.4500	-.726	-.615	-.462
	.7500	-.325	-.402	-.735
Lower	.0500	.812	.803	.624
	.1500	.504	.479	.222
	.4000	.239	.188	-.009
	.7000	.256	.299	-.068
Aileron				
Upper	.8050	-.291	-.316	-.932
	.8150	-.214	-.274	-.718
	.8350	-.214	-.197	-.906
	.8750	-.171	-.171	-.632
	.9250	-.128	-.145	-.444
	.9750	-.111	-.171	-.393
Lower	.8100	.342	.376	.017
	.8340	.402	.316	.034
	.9250	.103	.154	-.128
	.9750	.026	.043	-.239

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(m) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.769	-2.282
	.0100	-4.316	-5.547
	.0250	-4.085	-5.376
	.0500	-3.393	-4.496
	.1000	-1.974	-2.513
	.2250	-.966	-1.274
	.4500	-.709	-.675
	.7500	-.538	-.359
	.8350	-.462	-.248
	.8750	-.470	-.299
	.9250	-.470	-.214
	.9750	-.470	-.145
Lower	.0500	.932	.897
	.1500	.521	.590
	.4000	.094	.299
	.7000	-.077	.077
	.8530	-.103	-.085
	.9250	-.513	-.154
	.9750	-.487	-.214

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.769	-1.598	-.872
	.0100	-5.872	-4.915	-3.462
	.0250	-5.675	-4.957	-3.248
	.0500	-4.641	-4.769	-2.838
	.1000	-2.650	-2.239	-1.598
	.2250	-1.350	-1.137	-.846
	.4500	-.658	-.462	-.402
	.7500	-.188	-.111	-.265
Lower	.0500	.778	.897	.701
	.1500	.564	.554	.282
	.4000	.162	.179	-.043
	.7000	.034	.034	-.222
Aileron				
Upper	.8050	-.145	-.068	-.197
	.8150	-.154	-.060	-.154
	.8350	-.145	-.060	-.205
	.8750	-.094	-.043	-.368
	.9250	-.009	-.017	-.316
	.9750	-.000	.026	-.248
Lower	.8100	.017	.017	-.111
	.8340	.017	.205	-.068
	.9250	-.085	-.060	-.222
	.9750	-.043	-.026	-.162

TABLE XIV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 7.0 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(n) α_f = 10.9°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.530	-1.342
	.0100	-3.949	-4.333
	.0250	-3.838	-4.265
	.0500	-3.222	-3.778
	.1000	-1.855	-1.983
	.2250	-.769	-1.026
	.4500	-.564	-.641
	.7500	-.470	-.632
	.8350	-.368	-.410
	.8750	-.214	-.402
	.9250	-.368	-.385
	.9750	-.239	-.333
Lower	.0500	.778	.855
	.1500	.547	.385
	.4000	.231	.111
	.7000	.060	-.034
	.8530	.026	-.103
	.9250	-.282	-.231
	.9750	-.137	-.350

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.692	-.333	-.419
	.0100	-2.265	-2.393	-2.752
	.0250	-1.880	-2.402	-2.658
	.0500	-1.154	-2.145	-2.333
	.1000	-.692	-.744	-1.291
	.2250	-.650	-.632	-.624
	.4500	-.667	-.709	-.385
	.7500	-.632	-.658	.026
Lower	.0500	.744	.761	.658
	.1500	.393	.393	.222
	.4000	.026	.009	-.043
	.7000	-.299	-.376	-.385
Aileron				
Upper	.8050	-.641	-.530	.051
	.8150	-.521	-.521	.017
	.8350	-.521	-.590	.205
	.8750	-.521	-.530	.094
	.9250	-.521	-.470	.017
	.9750	-.487	-.376	-.111
Lower	.8100	-.359	-.316	-.556
	.8340	-.419	-.333	-.368
	.9250	-.513	-.376	-.265
	.9750	-.462	-.291	-.179

TABLE XIV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 7.0$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(o) $\alpha_f = 10.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-2.077
	.0100		-5.350
	.0250		-5.410
	.0500		-4.359
	.1000		-2.222
	.2250		-1.094
	.4500		-.667
	.7500		-.444
	.8350		-.197
	.8750		-.274
	.9250		-.222
	.9750		-.137
Lower	.0500		.872
	.1500		.658
	.4000		.325
	.7000		.145
	.8530		-.060
	.9250		-.103
	.9750		-.111

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.487	-1.444	-.709
	.0100	-5.598	-4.735	-3.248
	.0250	-5.658	-4.863	-3.077
	.0500	-4.624	-4.590	-2.761
	.1000	-2.692	-2.308	-1.624
	.2250	-1.308	-1.274	-.897
	.4500	-.692	-.684	-.521
	.7500	-.368	-.385	-.863
Lower	.0500	.855	.821	.624
	.1500	.556	.462	.188
	.4000	.256	.137	-.043
	.7000	.256	.214	-.179
Aileron				
Upper	.8050	-.171	-.325	-1.043
	.8150	-.256	-.308	-.744
	.8350	-.231	-.274	-.991
	.8750	-.222	-.256	-.735
	.9250	-.188	-.239	-.487
	.9750	-.222	-.222	-.444
Lower	.8100	.350	.333	-.077
	.8340	.368	.222	-.068
	.9250	.009	.120	-.179
	.9750	-.068	-.026	-.274

TABLE XV

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.669	-.947
	.0100	-1.071	-1.680
	.0250	-1.586	-1.645
	.0500	-1.467	-1.195
	.1000	-.994	-.852
	.2250	-.746	-.509
	.4500	-.396	-.355
	.7500	-.237	.178
	.8350	-.077	-.112
	.8750	-.107	-.107
	.9250	-.077	.089
	.9750	-.024	-.012
Lower	.0500	-.000	-.142
	.1500	-.089	-.183
	.4000	-.095	-.118
	.7000	-.041	.012
	.8530	-.118	.024
	.9250	-.077	.036
	.9750	.053	.065

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.006	.817	.852
	.0100	-1.627	-.675	-.420
	.0250	-1.456	-1.432	-.911
	.0500	-1.349	-1.385	-.882
	.1000	-.805	-.923	-.722
	.2250	-.538	-.615	-.527
	.4500	-.154	-.367	-.308
	.7500	.107	-.154	-.083
Lower	.0500	-.130	-.047	-.136
	.1500	-.178	-.166	-.225
	.4000	-.083	-.207	-.166
	.7000	-.083	-.130	-.107
Aileron				
Upper	.8050	-.095	-.083	-.006
	.8150	-.012	-.083	
	.8350	-.024	-.089	-.107
	.8750	.012	-.112	-.107
	.9250	.036	-.047	-.041
	.9750	-.095	.012	-.000
Lower	.8100	.024	-.059	-.059
	.8340	.000	.142	-.041
	.9250	.077	-.018	-.047
	.9750	.012	.053	.018

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) α_f = -4.7

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:				
		0.1621	0.2895	0.5000	0.6800	0.8920
Wing						
Upper	.0000	.521	.598	.604	.799	.834
	.0100	-1.266	-1.320	-1.024	-.870	-.621
	.0250	-1.817	-1.935	-1.722	-1.592	-1.053
	.0500	-1.627	-1.787	-1.604	-1.485	-1.036
	.1000	-1.207	-1.284	-1.284	-.964	-.817
	.2250	-.811	-.828	-.728	-.586	-.533
	.4500	-.527	-.456	-.491	-.266	-.219
	.7500	-.302	-.396	-.077	.142	.189
	.8350	-.036	-.101			
	.8750	-.065	-.047			
	.9250	-.047	-.053			
	.9750	-.077	.006			
Lower	.0500	.041	.183	.077	.107	-.006
	.1500	-.077	-.160	-.118	-.101	-.166
	.4000	-.160	-.195	-.237	-.207	-.183
	.7000	-.095	-.136	-.207	-.302	-.243
	.8530	-.118	-.065			
	.9250	-.095	-.036			
	.9750	-.065	-.018			
Aileron						
Upper	.8050	-.201	.201			
	.8150	-.036	.243			
	.8350	.136	.254			
	.8750	.053	.160			
	.9250	.006	.107			
	.9750	-.006	.041			
Lower	.8100	-.254	-.385			
	.8340	-.065	-.278			
	.9250	-.041	-.089			
	.9750	-.036	-.012			

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -4.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.645	.734
	.0100	-.976	-.888
	.0250	-1.527	-1.680
	.0500	-1.491	-1.485
	.1000	-1.041	-1.059
	.2250	-.769	-.751
	.4500	-.426	-.473
	.7500	-.201	-.485
	.8350	.018	-.024
	.8750	.012	-.012
	.9250	-.107	.000
	.9750	-.036	.101
Lower	.0500	.065	.189
	.1500	-.166	-.124
	.4000	-.237	-.201
	.7000	-.059	-.047
	.8530	-.095	-.012
	.9250	-.101	.024
	.9750	-.030	.018

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.899	.953	.941
	.0100	-.627	-.325	-.053
	.0250	-1.331	-1.071	-.538
	.0500	-1.231	-1.059	-.598
	.1000	-1.077	-.822	-.609
	.2250	-.710	-.568	-.497
	.4500	-.556	-.420	-.325
	.7500	-.361	-.497	-.420
Lower	.0500	.000	-.178	-.302
	.1500	-.178	-.249	-.266
	.4000	-.118	-.136	-.118
	.7000	.095	.189	.071
Aileron				
Upper	.8050	-.391	-.491	-.420
	.8150	-.355	-.432	-.450
	.8350	-.272	-.302	-.485
	.8750	-.189	-.195	-.254
	.9250	-.107	-.136	-.154
	.9750	-.095	-.112	-.142
Lower	.8100	.183	.320	.260
	.8340	.237	.314	.278
	.9250	-.036	.112	.071
	.9750	-.012	-.000	.000

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.065	.172
	.0100	-2.213	-2.260
	.0250	-2.609	-2.905
	.0500	-2.337	-2.521
	.1000	-1.550	-1.651
	.2250	-1.065	-.970
	.4500	-.527	-.651
	.7500	-.290	-.438
	.8350	-.160	-.107
	.8750	-.065	-.083
	.9250	-.118	-.024
	.9750	.030	.018
Lower	.0500	.373	.521
	.1500	.107	.112
	.4000	-.036	-.101
	.7000	.036	-.006
	.8530	-.095	-.012
	.9250	-.207	-.012
	.9750	-.148	-.012

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.195	.515	.663
	.0100	-2.036	-1.592	-1.112
	.0250	-2.550	-2.296	-1.462
	.0500	-2.284	-2.089	-1.402
	.1000	-1.669	-1.290	-.982
	.2250	-.947	-.757	-.604
	.4500	-.580	-.414	-.296
	.7500	-.172	-.112	-.095
Lower	.0500	.450	.373	.213
	.1500	.118	.077	-.030
	.4000	-.059	-.071	-.095
	.7000	-.065	-.053	-.130
Aileron				
Upper	.8050	-.101	-.065	-.024
	.8150	-.107	-.065	-.036
	.8350	-.036	-.071	-.112
	.8750	-.006	-.059	-.160
	.9250	.024	.006	-.077
	.9750	.030	.071	-.018
Lower	.8100	-.047	-.012	-.083
	.8340	-.012	.201	-.012
	.9250	-.012	.006	-.118
	.9750	.059	.083	-.012

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = -15^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) $\alpha_f = -0.9^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.006
	.0100		-2.651
	.0250		-3.089
	.0500		-2.769
	.1000		-1.757
	.2250		-1.018
	.4500		-.580
	.7500		-.438
	.8350		-.047
	.8750		.006
	.9250		.024
	.9750		-.118
Lower	.0500		.574
	.1500		.142
	.4000		-.012
	.7000		-.118
	.8530		-.000
	.9250		-.036
	.9750		-.000

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.343	.296	.497
	.0100	-2.876	-2.041	-1.385
	.0250	-2.473	-2.645	-1.680
	.0500	-1.704	-2.379	-1.586
	.1000	-.923	-1.396	-1.053
	.2250	-.503	-.781	-.651
	.4500	.012	-.337	-.243
	.7500	.598	.124	.172
Lower	.0500	.189	.479	.266
	.1500	-.036	.130	.030
	.4000	-.178	-.089	-.101
	.7000	-.112	-.290	-.249
Aileron				
Upper	.8050	.047	.136	.166
	.8150	.107	.160	.213
	.8350	.172	.166	.237
	.8750	-.053	.112	.160
	.9250	-.018	.083	.089
	.9750	-.142	.053	.024
Lower	.8100	.006	-.420	-.432
	.8340	.059	-.183	-.308
	.9250	.041	-.053	-.112
	.9750	.083	-.006	-.041

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 10.2$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.207	.272
	.0100	-2.018	-2.083
	.0250	-2.462	-2.763
	.0500	-2.166	-2.379
	.1000	-1.450	-1.568
	.2250	-.888	-.870
	.4500	-.479	-.586
	.7500	-.213	-.408
	.8350	-.107	-.112
	.8750	-.124	-.041
	.9250	-.089	.012
	.9750	.006	.059
Lower	.0500	.426	.456
	.1500	.160	.107
	.4000	-.107	-.024
	.7000	-.000	.018
	.8530	-.142	-.006
	.9250	-.148	-.012
	.9750	-.101	-.024

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.308	.598	.751
	.0100	-1.858	-1.349	-.781
	.0250	-2.391	-2.047	-1.189
	.0500	-2.142	-1.882	-1.178
	.1000	-1.609	-1.231	-.905
	.2250	-.888	-.775	-.633
	.4500	-.615	-.485	-.391
	.7500	-.325	-.420	-.497
Lower	.0500	.438	.260	.053
	.1500	.112	.024	-.095
	.4000	-.012		-.095
	.7000	.142	.201	.018
Aileron				
Upper	.8050	-.361	-.396	-.467
	.8150	-.296	-.361	-.450
	.8350	-.243	-.249	-.456
	.8750	-.195	-.178	-.314
	.9250	-.148	-.142	-.225
	.9750	-.095	-.148	-.201
Lower	.8100	.219	.349	.213
	.8340	.314	.331	.225
	.9250	.018	.095	-.059
	.9750	-.012	-.024	-.160

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.959
	.0100		-3.988
	.0250		-4.343
	.0500		-3.870
	.1000		-2.231
	.2250		-1.243
	.4500		-.787
	.7500		-.379
	.8350		-.148
	.8750		-.041
	.9250		-.041
	.9750		-.030
Lower	.0500		.746
	.1500		.361
	.4000		.136
	.7000		-.018
	.8530		-.018
	.9250		-.041
	.9750		-.036

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.941	-.231	.136
	.0100	-3.746	-3.077	-2.053
	.0250	-4.041	-3.521	-2.207
	.0500	-3.568	-3.207	-1.953
	.1000	-2.195	-1.722	-1.237
	.2250	-1.178	-.929	-.728
	.4500	-.716	-.438	-.325
	.7500	-.124	-.095	-.172
Lower	.0500	.775	.692	.503
	.1500	.290	.325	.148
	.4000	.172	.077	-.071
	.7000	.024	.012	-.148
Aileron				
Upper	.8050	-.101	-.065	-.089
	.8150	-.018	-.059	-.089
	.8350	-.077	-.053	-.219
	.8750	.018	-.053	-.278
	.9250	.018	.065	-.195
	.9750	.065	.077	-.107
Lower	.8100	-.071	.065	-.047
	.8340	.053	.266	-.071
	.9250	.041	.036	-.148
	.9750	.036	.089	-.095

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) α_f = 2.9°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-1.136 -4.500 -4.818 -4.182 -2.273 -.909 -.636 -.409 -.136 .000 -.091 -.046
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.773 .500 .227 .000 -.046 -.091 -.091

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.500 -4.546 -4.864 -4.046 -2.500 -1.273 -.636 -.136	-.727 -4.046 -4.364 -4.046 -2.136 -1.136 -.455 -.046	-.091 -2.636 -2.773 -2.591 -1.591 -1.000 -.409 -.091
Lower	.0500 .1500 .4000 .7000	.818 .455 .046 -.136	.727 .273 -.091 -.364	.500 .091 -.227 -.364
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.182 -.091 -.046 .000 .000 .000	-.136 -.091 -.091 .046 .046 .046	.091 .136 .182 -.046 -.136 -.182
Lower	.8100 .8340 .9250 .9750	-.227 -.091 -.091 -.046	-.227 .091 -.091 -.046	-.227 -.182

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 10.2$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.852	-.769
	.0100	-3.515	-3.769
	.0250	-3.663	-4.095
	.0500	-3.195	-3.645
	.1000	-2.047	-2.083
	.2250	-1.302	-1.189
	.4500	-.663	-.710
	.7500	-.296	-.462
	.8350	-.000	-.047
	.8750	-.041	-.118
	.9250	-.065	-.036
	.9750	-.059	.006
Lower	.0500	.621	.740
	.1500	.373	.361
	.4000	.101	.107
	.7000	.083	.077
	.8530	-.018	-.024
	.9250	-.071	-.024
	.9750	-.083	-.065

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.698	-.012	.266
	.0100	-3.396	-2.627	-1.781
	.0250	-3.722	-3.195	-2.012
	.0500	-3.296	-2.923	-1.852
	.1000	-2.142	-1.686	-1.225
	.2250	-1.172	-.964	-.781
	.4500	-.686	-.562	-.408
	.7500	-.367	-.391	-.574
Lower	.0500	.775	.592	.408
	.1500	.355	.272	.095
	.4000	.112	.071	-.065
	.7000	.225	.231	-.041
Aileron				
Upper	.8050	-.331	-.325	-.598
	.8150	-.266	-.320	-.568
	.8350	-.225	-.219	-.669
	.8750	-.154	-.148	-.456
	.9250	-.124	-.142	-.337
	.9750	-.118	-.154	-.325
Lower	.8100	.213	.343	.118
	.8340	.296	.308	.130
	.9250	-.024	.112	-.089
	.9750	-.012	.006	-.183

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 10.2 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-2.083
	.0100		-5.633
	.0250		-5.538
	.0500		-4.450
	.1000		-2.503
	.2250		-1.172
	.4500		-.521
	.7500		-.065
	.8350		-.024
	.8750		.036
	.9250		.036
	.9750		-.065
Lower	.0500		1.018
	.1500		.734
	.4000		.574
	.7000		.414
	.8530		.018
	.9250		.018
	.9750		.018

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-2.172	-1.219	-.592
	.0100	-5.314	-4.497	-3.107
	.0250	-5.308	-4.716	-2.970
	.0500	-4.302	-4.402	-2.639
	.1000	-2.497	-2.136	-1.562
	.2250	-1.278	-1.095	-.840
	.4500	-.503	-.467	-.414
	.7500	-.077	-.112	-.254
Lower	.0500	.923	.882	.663
	.1500	.692	.497	.266
	.4000	.308	.136	-.024
	.7000	.148	-.006	-.213
Aileron				
Upper	.8050	.000	-.071	-.189
	.8150	-.047	-.047	-.178
	.8350	-.006	-.071	-.260
	.8750	.036	-.047	-.373
	.9250	.024	-.030	-.314
	.9750	.095	.036	-.201
Lower	.8100	.065	.047	-.118
	.8340	.107	.237	-.083
	.9250	.053	-.006	-.201
	.9750	.065	.036	-.154

TABLE XV.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 10.8 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) α_f = 6.7°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.8350 .8750 .9250 .9750		-.067 -.067 -.067 -.050
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.613 .383 .133 .096 .008 -.025 -.083

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.875 -4.017 -3.950 -3.175 -1.904 -.896 -.350 -.063	-1.100 -3.492 -3.517 -3.296 -1.521 -.717 -.238 -.038	-.625 -2.458 -2.267 -1.925 -1.088 -.554 -.213 .088
Lower	.0500 .1500 .4000 .7000	.608 .367 .150 -.067	.654 .392 .113 -.133	.529 .254 .013 -.196
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.150 -.067 -.025 -.025 -.021 -.021	-.042 -.004 .000 .021 .054 .054	.138 .100 .242 .046 -.033 -.117
Lower	.8100 .8340 .9250 .9750	-.025 -.033 -.025 -.017	-.275 -.088 -.042 -.004	-.313 -.171 -.183 -.104

TABLE XV.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 10.8$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(l) $\alpha_f = 6.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.308
	.0100		-3.854
	.0250		-3.896
	.0500		-3.154
	.1000		-1.754
	.2250		-.896
	.4500		-.429
	.7500		-.075
	.8350		-.021
	.8750		.029
	.9250		.025
	.9750		.029
Lower	.0500		.700
	.1500		.479
	.4000		.246
	.7000		.146
	.8530		.088
	.9250		.046
	.9750		-.038

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.417	-.738	-.329
	.0100	-3.625	-3.029	-2.050
	.0250	-3.696	-3.217	-2.038
	.0500	-3.042	-3.021	-1.788
	.1000	-1.779	-1.542	-1.071
	.2250	-.942	-.817	-.621
	.4500	-.375	-.408	-.321
	.7500	-.129	-.238	-.488
Lower	.0500	.683	.579	.438
	.1500	.446	.329	.150
	.4000	.225	.129	-.013
	.7000	.200	.204	-.046
Aileron				
Upper	.8050	-.121	-.179	-.583
	.8150	-.108	-.188	-.479
	.8350	-.071	-.146	-.621
	.8750	-.067	-.117	-.421
	.9250	-.075	-.117	-.300
	.9750	-.038	-.125	-.258
Lower	.8100	.283	.250	.025
	.8340	.283	.238	.013
	.9250	.033	.079	-.104
	.9750	.021	.021	-.175

TABLE XVI

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.7 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(a) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.636	.697
	.0100	-1.273	-1.364
	.0250	-1.848	-2.091
	.0500	-1.697	-1.879
	.1000	-1.182	-1.364
	.2250	-.758	-.818
	.4500	-.485	-.545
	.7500	-.242	-.485
	.8350	.000	.000
	.8750	-.030	.061
	.9250	.030	.061
	.9750	.061	.091
Lower	.0500	.182	.212
	.1500	.000	-.121
	.4000	-.152	-.121
	.7000	.030	-.121
	.8530	.030	-.061
	.9250	.030	-.030
	.9750	.030	.000

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.727	.879	.879
	.0100	-1.242	-.939	-.545
	.0250	-1.909	-1.697	-1.030
	.0500	-1.727	-1.667	-1.030
	.1000	-1.394	-1.091	-.879
	.2250	-.788	-.727	-.606
	.4500	-.545	-.424	-.303
	.7500	-.121	-.152	-.061
Lower	.0500	.182	.030	-.091
	.1500	-.030	-.182	-.212
	.4000	-.182	-.242	-.182
	.7000	-.091	-.182	-.152
Aileron				
Upper	.8050	-.091	-.091	.000
	.8150	-.121	-.091	.030
	.8350	-.030	-.091	-.121
	.8750	-.030	-.121	-.152
	.9250	-.030	-.061	-.061
	.9750	.000	.000	.000
Lower	.8100	-.091	-.091	-.121
	.8340	.000	.000	-.091
	.9250	-.030	-.061	-.091
	.9750	.030	.000	.000

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 14.7 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(b) α_f = -4.8°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.455
	.0100		-1.754
	.0250		-2.500
	.0500		-2.197
	.1000		-1.434
	.2250		-.881
	.4500		-.664
	.7500		-.475
	.8350		-.131
	.8750		-.053
	.9250		-.053
	.9750		-.020
Lower	.0500		.344
	.1500		-.029
	.4000		-.135
	.7000		-.143
	.8530		-.074
	.9250		-.074
	.9750		-.057

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.406	.697	.795
	.0100	-1.705	-1.246	-.816
	.0250	-2.275	-1.971	-1.213
	.0500	-1.992	-1.803	-1.168
	.1000	-1.541	-1.148	-.889
	.2250	-.824	-.668	-.566
	.4500	-.520	-.344	-.242
	.7500	-.156	.025	.123
Lower	.0500	.320	.234	.082
	.1500	-.008	.004	-.098
	.4000	-.242	-.156	-.135
	.7000	-.242	-.221	-.197
Aileron				
Upper	.8050	-.172	.066	.156
	.8150	-.074	.082	.172
	.8350	-.074	.061	.152
	.8750	-.025	.033	.074
	.9250	-.041	.033	.045
	.9750	-.049	.041	-.004
Lower	.8100	-.012	-.008	-.275
	.8340	-.012	-.102	-.230
	.9250	-.066	-.041	-.086
	.9750	.029	.033	-.029

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 14.7$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(c) $\alpha_f = -4.7^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		.402 -.742 -1.455 -1.361 -1.164 -.885 -.701 -.545 -.045 -.057 -.107 -.082
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		-.066 -.369 -.307 -.156 -.070 -.045 -.008

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	.885 -.209 -.988 -.926 -1.086 -.668 -.553 -.447	.975 .107 -.598 -.619 -.594 -.459 -.381 -.480	.918 .279 -.205 -.303 -.443 -.414 -.344 -.451
Lower	.0500 .1500 .4000 .7000	-.238 -.307 -.156 .111	-.463 -.365 -.184 .234	-.545 -.361 -.131 .082
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.533 -.500 -.459 -.348 -.266 -.164	-.451 -.414 -.270 -.254 -.238 -.250	-.480 -.471 -.545 -.307 -.148 -.082
Lower	.8100 .8340 .9250 .9750	.213 .451 .016 -.066	.336 .336 .119 -.037	.246 .258 .057 -.004

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.4 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(d) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	.042	.096
	.0100	-2.217	-2.454
	.0250	-2.467	-2.904
	.0500	-2.208	-2.583
	.1000	-1.500	-1.571
	.2250	-.921	-.908
	.4500	-.517	-.546
	.7500	-.154	-.350
	.8350	-.071	-.071
	.8750	-.017	.008
	.9250	.063	.100
	.9750	.054	.150
Lower	.0500	.517	.621
	.1500	.267	.163
	.4000	.042	.017
	.7000	.042	.017
	.8530	-.021	.038
	.9250	-.088	.017
	.9750	.033	.038

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.075	.429	.579
	.0100	-2.321	-1.804	-1.150
	.0250	-2.775	-2.475	-1.479
	.0500	-2.442	-2.238	-1.396
	.1000	-1.750	-1.358	-1.000
	.2250	-.908	-.821	-.642
	.4500	-.638	-.433	-.321
	.7500	-.221	-.233	-.150
Lower	.0500	.550	.421	.213
	.1500	.175	.129	-.004
	.4000	-.046	-.029	-.088
	.7000	.046	.033	-.113
Aileron				
Upper	.8050	-.221	-.204	-.100
	.8150	-.179	-.179	-.104
	.8350	-.138	-.154	-.213
	.8750	-.046	-.096	-.208
	.9250	.021	-.013	-.129
	.9750	.033	.067	-.063
Lower	.8100	.146	.096	.021
	.8340	.146	.192	-.058
	.9250	-.029	.021	-.108
	.9750	.029	.058	-.042

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 14.7 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(e) α_f = -1.0°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.373	-.242
	.0100	-2.803	-3.225
	.0250	-3.041	-3.664
	.0500	-2.697	-3.234
	.1000	-1.627	-1.770
	.2250	-.963	-.943
	.4500	-.520	-.607
	.7500		-.332
	.8350	.000	-.115
	.8750	-.025	-.049
	.9250	-.016	-.037
	.9750	.012	.041
Lower	.0500	.512	.680
	.1500	.266	.270
	.4000	.086	.045
	.7000	-.016	-.016
	.8530	-.082	-.037
	.9250	-.131	-.086
	.9750	-.033	-.066

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.336	.094	.348
	.0100	-3.057	-2.488	-1.701
	.0250	-3.414	-3.053	-1.914
	.0500	-2.975	-2.779	-1.758
	.1000	-1.930	-1.553	-1.143
	.2250	-1.029	-.848	-.672
	.4500	-.566	-.381	-.270
	.7500	-.156	-.008	.066
Lower	.0500	.631	.590	.414
	.1500	.270	.234	.111
	.4000	.025	-.004	-.070
	.7000	-.041	-.160	-.225
Aileron				
Upper	.8050	-.066	.045	.127
	.8150	-.066	.053	.201
	.8350	.000	.057	.148
	.8750	.020	.029	.033
	.9250	.016	.049	-.012
	.9750	.045	.070	-.061
Lower	.8100	.033	.053	-.299
	.8340	.049	.020	-.172
	.9250	-.029	-.029	-.176
	.9750	.020	.049	-.090

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 14.7$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(f) $\alpha_f = -1.0^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		.324
	.0100		-2.160
	.0250		-2.787
	.0500		-2.357
	.1000		-1.561
	.2250		-.881
	.4500		-.594
	.7500		-.459
	.8350		-.057
	.8750		-.057
	.9250		-.037
	.9750		-.037
Lower	.0500		.328
	.1500		.070
	.4000		.004
	.7000		-.070
	.8530		-.008
	.9250		.025
	.9750		-.020

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	.553	.795	.877
	.0100	-1.619	-.934	-.471
	.0250	-2.221	-1.725	-.902
	.0500	-1.980	-1.639	-.934
	.1000	-1.557	-1.086	-.836
	.2250	-.861	-.709	-.590
	.4500	-.586	-.455	-.348
	.7500	-.398	-.414	-.443
Lower	.0500	.398	.143	-.045
	.1500	.070	-.029	-.135
	.4000	.020	-.037	-.107
	.7000	.217	.242	.037
Aileron				
Upper	.8050	-.303	-.352	-.406
	.8150	-.373	-.361	-.426
	.8350	-.287	-.234	-.537
	.8750	-.266	-.184	-.328
	.9250	-.266	-.184	-.184
	.9750	-.168	-.176	-.156
Lower	.8100	.303	.373	.242
	.8340	.516	.385	.258
	.9250	.074	.127	-.025
	.9750	.020	-.033	-.090

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 0^o$; cable configuration 2-5-8; q_∞ = 14.7 lb/sq ft, belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(g) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-.182	-.061
	.0100	-2.727	-3.061
	.0250	-2.939	-3.576
	.0500	-2.667	-3.152
	.1000	-1.818	-2.242
	.2250	-.848	-1.000
	.4500	-.424	-.576
	.7500	-.303	-.394
	.8350	.061	-.091
	.8750	-.030	.000
	.9250	-.182	.000
	.9750	-.121	.152
Lower	.0500	.606	.788
	.1500	.121	.303
	.4000	.030	.152
	.7000	.091	.121
	.8530	-.091	.030
	.9250	-.030	-.030
	.9750	.061	-.030

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.091	.242	.455
	.0100	-2.818	-2.364	-1.606
	.0250	-3.333	-3.030	-1.939
	.0500	-2.849	-2.727	-1.818
	.1000	-1.939	-1.636	-1.242
	.2250	-1.000	-.970	-.758
	.4500	-.667	-.515	-.364
	.7500	-.121	-.182	-.121
Lower	.0500	.667	.485	-.091
	.1500	.303	.182	.364
	.4000	.031	-.061	.061
	.7000	.000	-.091	-.121
				-.182
Aileron				
Upper	.8050	-.091	-.091	-.091
	.8150	-.091	-.091	
	.8350	-.030	-.091	-.182
	.8750	.000	-.091	-.212
	.9250	.030	-.030	-.152
	.9750	.061	.061	-.091
Lower	.9100	-.081	.030	-.121
	.8340	.030	.182	-.061
	.9250	.000	.000	-.152
	.9750	.061	.061	-.121

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[δ_a = -15°; cable configuration 2-5-8; q_∞ = 14.7 lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(h) α_f = 2.8°

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500 .8350 .8750 .9250 .9750		-1.635 -5.373 -5.307 -4.410 -2.414 -1.246 -.734 -.213 -.115 -.098 -.004 -.025
Lower	.0500 .1500 .4000 .7000 .8530 .9250 .9750		.951 .459 .148 .107 -.008 -.045 -.098

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000 .0100 .0250 .0500 .1000 .2250 .4500 .7500	-1.984 -5.078 -5.078 -4.270 -2.520 -1.246 -.639 -.127	-1.078 -4.377 -4.594 -4.311 -2.094 -1.025 -.418 -.008	- .513 -1.823 -1.994 -1.852 -1.254 - .826 - .399 - .028
Lower	.0500 .1500 .4000 .7000	.873 .516 .143 .008	.877 .516 .143 -.078	+.627 .199 .000 -.228
Aileron				
Upper	.8050 .8150 .8350 .8750 .9250 .9750	-.078 -.078 -.049 -.025 .016 .049	.016 .037 .045 .053 -.057 -.057	+.057 .114 .114 -.057 -.085 -.142
Lower	.8100 .8340 .9250 .9750	.086 .049 -.029 .045	000 -.114 -.171 -.085	-.171 -.313 -.256

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 14.7$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(i) $\alpha_f = 2.8^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-.750
	.0100		-3.951
	.0250		-4.266
	.0500		-3.742
	.1000		-2.225
	.2250		-1.287
	.4500		-.807
	.7500		-.484
	.8350		-.221
	.8750		-.098
	.9250		-.033
	.9750		-.041
Lower	.0500		.758
	.1500		.205
	.4000		.135
	.7000		.016
	.8530		-.041
	.9250		-.029
	.9750		-.037

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-.533	.205	.484
	.0100	-3.389	-2.389	-1.500
	.0250	-3.832	-3.082	-1.816
	.0500	-3.193	-2.791	-1.672
	.1000	-2.156	-1.656	-1.189
	.2250	-1.193	-.984	-.775
	.4500	-.775	-.566	-.406
	.7500	-.410	-.393	-.570
Lower	.0500	.553	.566	.311
	.1500	.307	.221	.049
	.4000	.082	.045	-.074
	.7000	.184	.266	-.061
Aileron				
Upper	.8050	-.398	-.320	-.545
	.8150	-.352	-.320	-.504
	.8350	-.324	-.225	-.602
	.8750	-.303	-.172	-.393
	.9250	-.246	-.156	-.279
	.9750	-.180	-.168	-.279
Lower	.8100	.291	.402	.143
	.8340	.553	.434	.143
	.9250	.045	.139	-.131
	.9750	-.037	.008	-.242

TABLE XVI.- Continued

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = -15^\circ$; cable configuration 2-5-8; $q_\infty = 14.7$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(j) $\alpha_f = 7.3^\circ$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000	-1.994	-1.610
	.0100	-5.640	-5.799
	.0250	-5.451	-5.933
	.0500	-4.530	-5.591
	.1000	-2.823	-2.329
	.2250	-1.677	-1.659
	.4500	-.945	-1.122
	.7500	-.610	-.549
	.8350	-.384	-.494
	.8750	-.573	-.494
	.9250	-.537	-.421
	.9750	-.500	-.311
Lower	.0500	.933	1.311
	.1500	.390	.860
	.4000	.159	.360
	.7000	.067	.067
	.8530	-.177	-.098
	.9250	-.457	-.220
	.9750	-.274	-.445

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.823	-1.335	-.750
	.0100	-4.610	-5.616	-4.311
	.0250	-3.646	-6.000	-4.055
	.0500	-2.896	-5.360	-3.543
	.1000	-2.012	-2.573	-2.134
	.2250	-1.591	-1.390	-1.122
	.4500	-1.085	-.732	-.518
	.7500	-.530	-.409	-.366
Lower	.0500	1.177	1.177	.890
	.1500	.768	.677	.360
	.4000	.165	.165	-.067
	.7000	-.024	-.104	-.421
Aileron				
Upper	.8050	-.695	-.427	-.354
	.8150	-.524	-.311	-.274
	.8350	-.537	-.323	-.165
	.8750	-.506	-.238	-.171
	.9250	-.457	-.226	-.287
	.9750	-.415	-.226	-.530
Lower	.8100	-.280	-.018	-.341
	.8340	.165	-.152	-.384
	.9250	-.232	-.183	-.299
	.9750	-.311	-.177	

TABLE XVI.- Concluded

CHORDWISE PRESSURE COEFFICIENTS FOR WING AND AILERON

[$\delta_a = 26^o$; cable configuration 2-5-8; $q_\infty = 14.7$ lb/sq ft; belly plate on; p = 7.0 lb/sq in.; forward guy cables, heavily tightened]

(k) $\alpha_f = 7.4^o$

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:	
		0.1621	0.2895
Wing			
Upper	.0000		-1.246
	.0100		-4.635
	.0250		-4.746
	.0500		-3.992
	.1000		-2.193
	.2250		-1.135
	.4500		-.746
	.7500		-.459
	.8350		-.275
	.8750		-.201
	.9250		-.201
	.9750		-.131
Lower	.0500		.799
	.1500		.389
	.4000		.180
	.7000		-.004
	.8530		-.107
	.9250		-.135
	.9750		-.135

Surface	$\frac{x}{c}$	C_p for values of $y/\frac{b}{2}$ of:		
		0.5000	0.6800	0.8920
Wing				
Upper	.0000	-1.430	-.484	-.070
	.0100	-4.545	-3.557	-2.324
	.0250	-4.877	-4.045	-2.434
	.0500	-3.963	-3.717	-2.160
	.1000	-2.418	-1.980	-1.385
	.2250	-1.230	-1.074	-.824
	.4500	-.693	-.549	-.439
	.7500	-.344	-.352	-.578
Lower	.0500	.754	.766	.541
	.1500	.520	.398	.172
	.4000	.098	.131	-.041
	.7000	.184	.225	-.143
Aileron				
Upper	.8050	-.348	-.291	-.607
	.8150	-.324	-.283	-.541
	.8350	-.270	-.189	-.725
	.8750	-.217	-.131	-.582
	.9250	-.217	-.123	-.377
	.9750	-.168	-.131	-.266
Lower	.8100	.369	.369	.016
	.8340	.520	.377	.041
	.9250	.025	.127	-.221
	.9750	-.078	.016	-.221

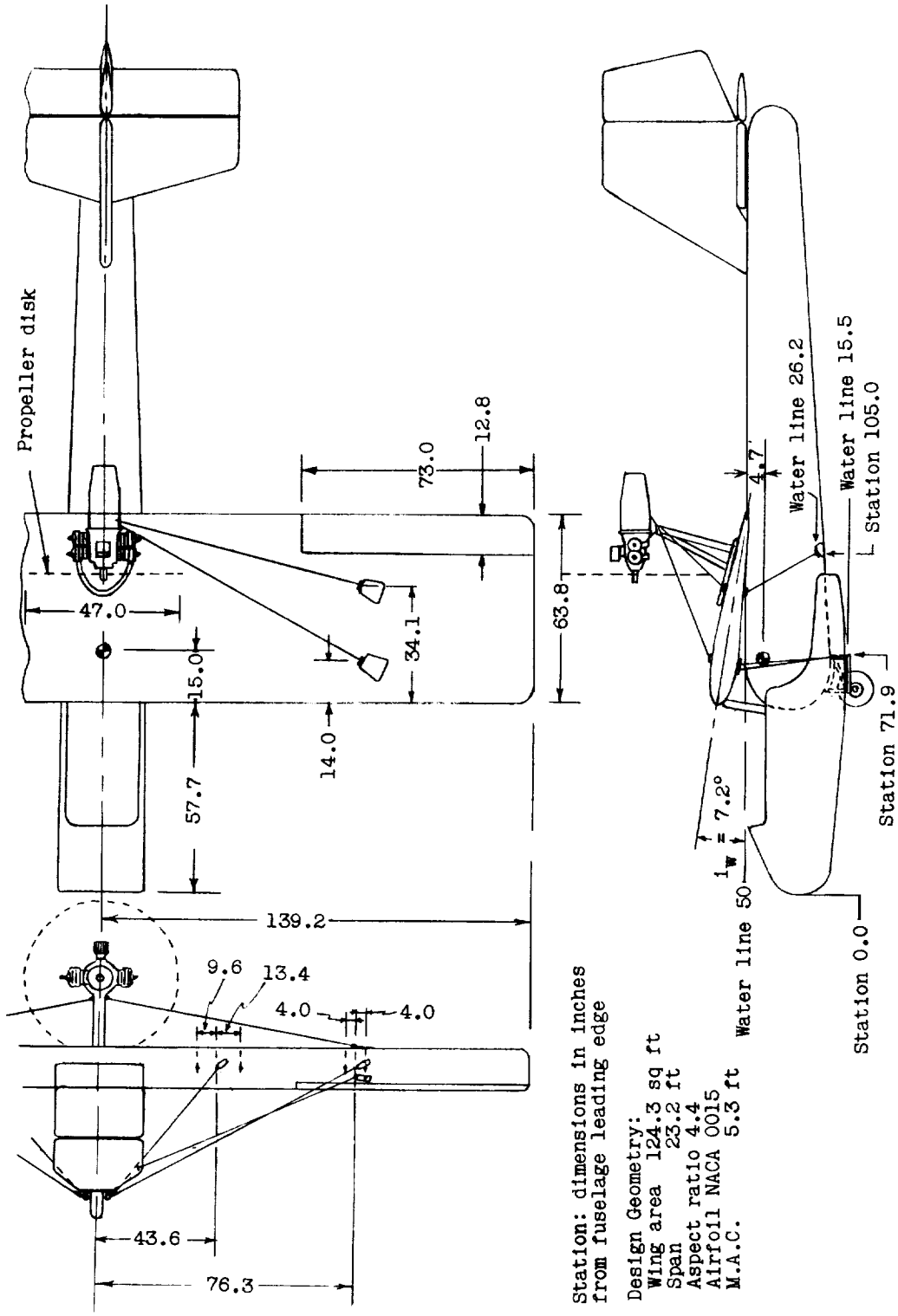

Figure 1.- Geometric characteristics of Inflatoplane II. All dimensions are in inches.

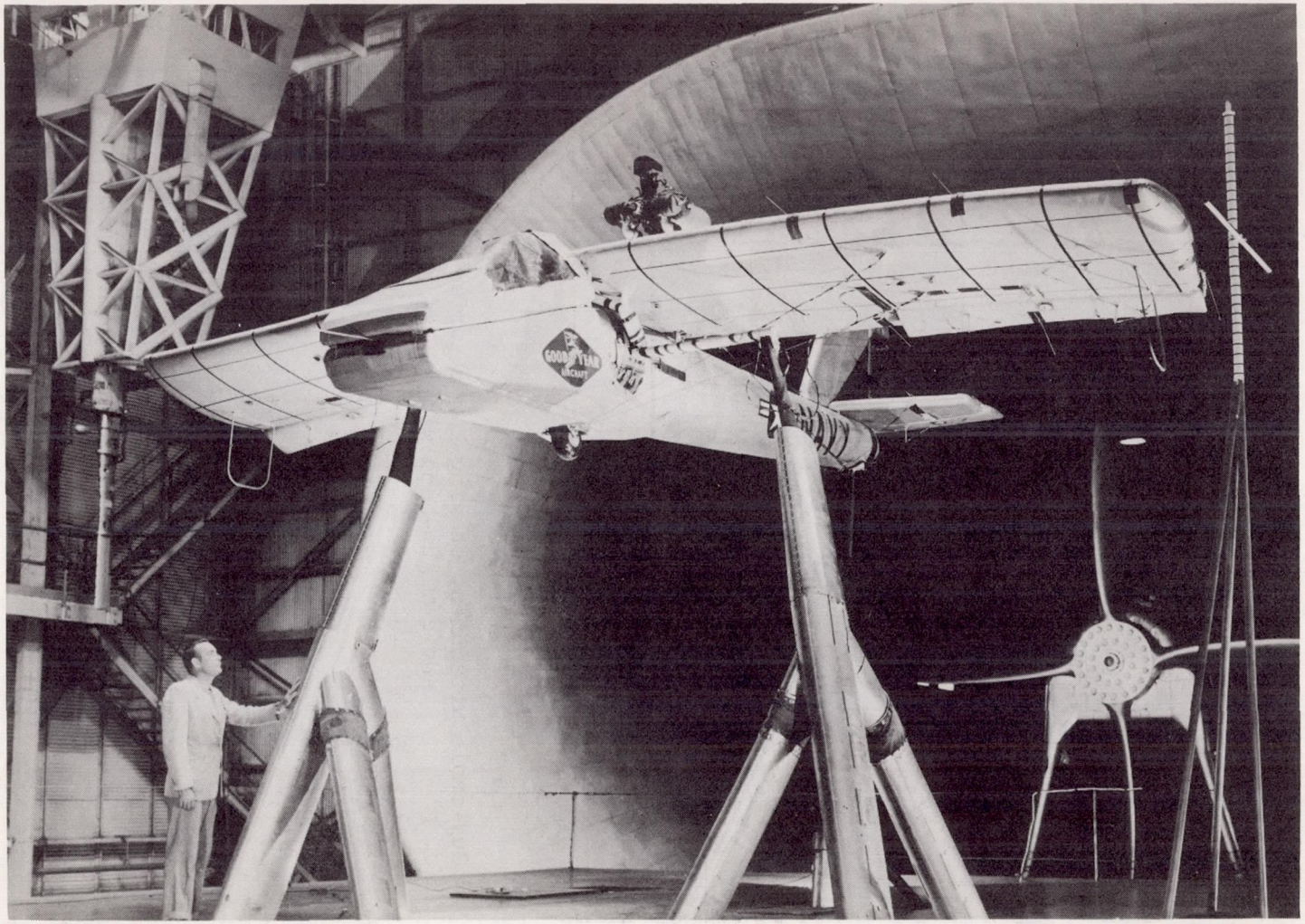

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Figure 2.- Inflatoplane I of reference 1 mounted for tests in the Langley full-scale tunnel. (Airplane is similar to the one used for the present investigation.)

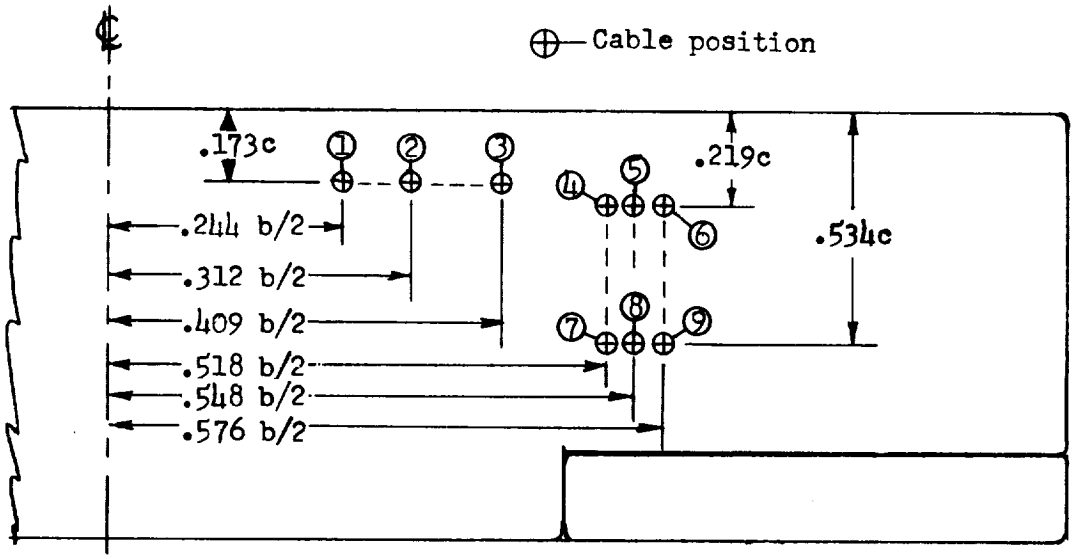


Figure 3.- Guy-cable attachment on wing lower surface. View is from above right-hand wing.

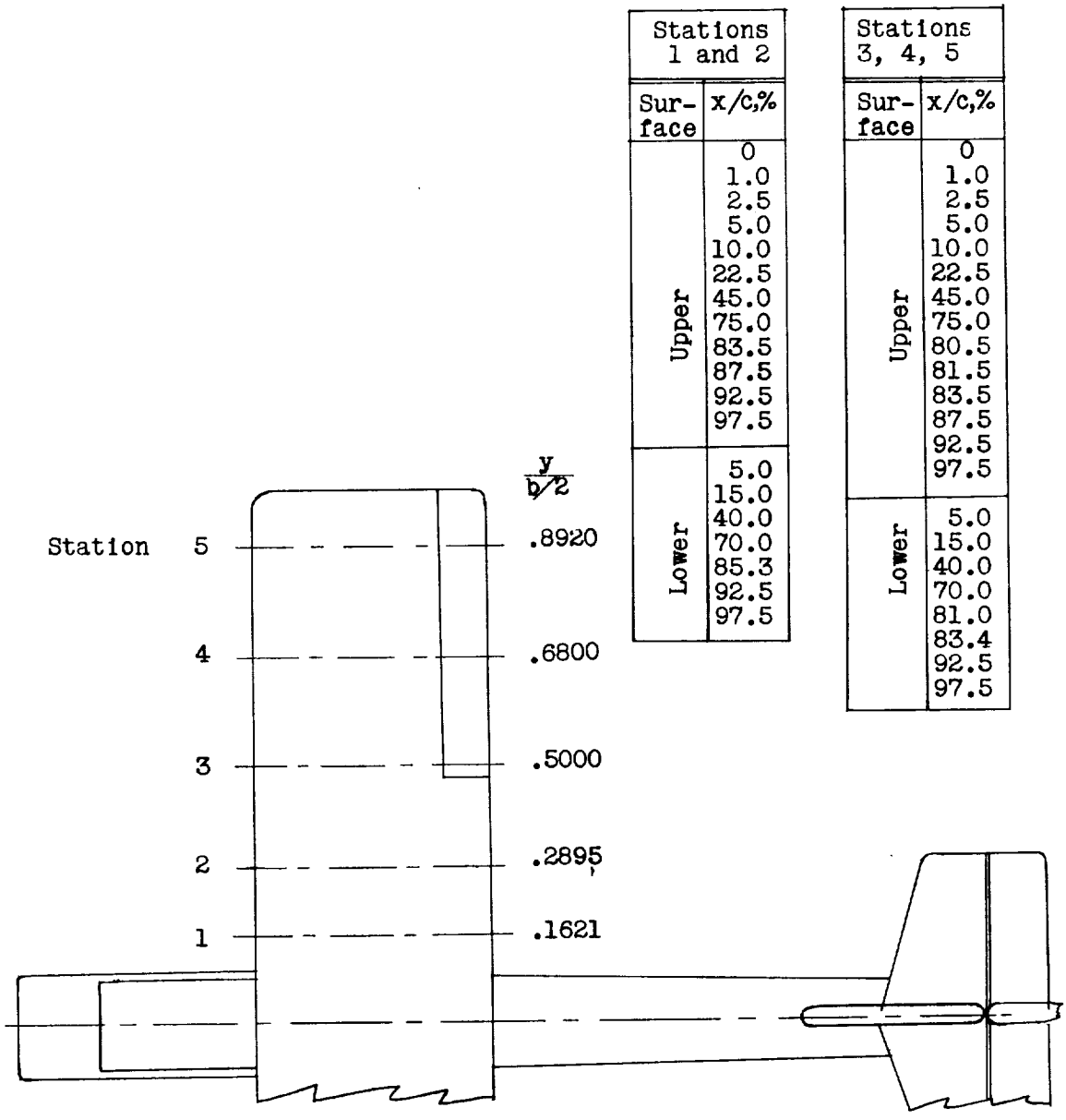


Figure 4.- Spanwise and chordwise locations of the surface-pressure orifices.

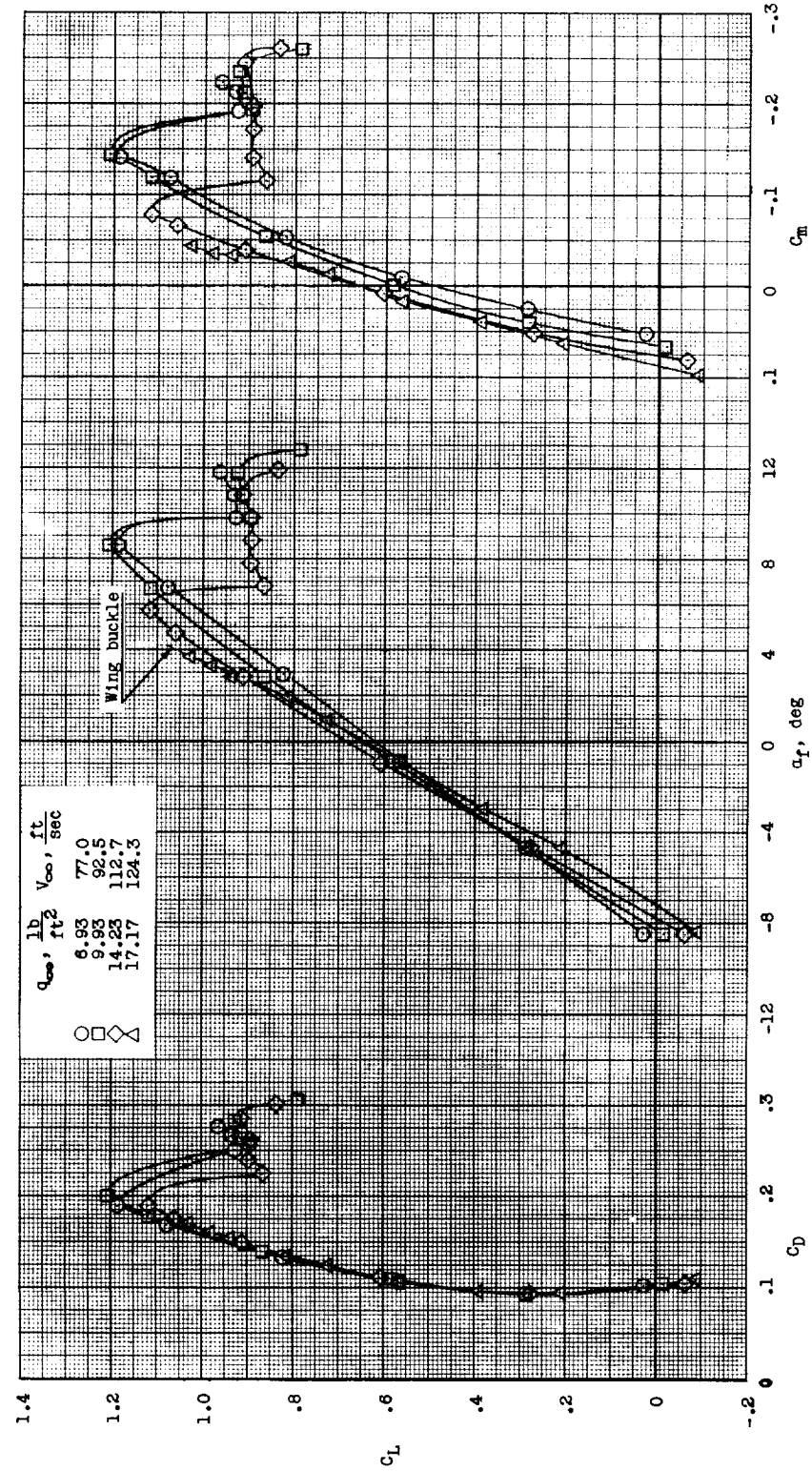


Figure 5.- Aerodynamic characteristics for several tunnel speeds. Cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened in static condition; p = 7.0 lb/sq in.

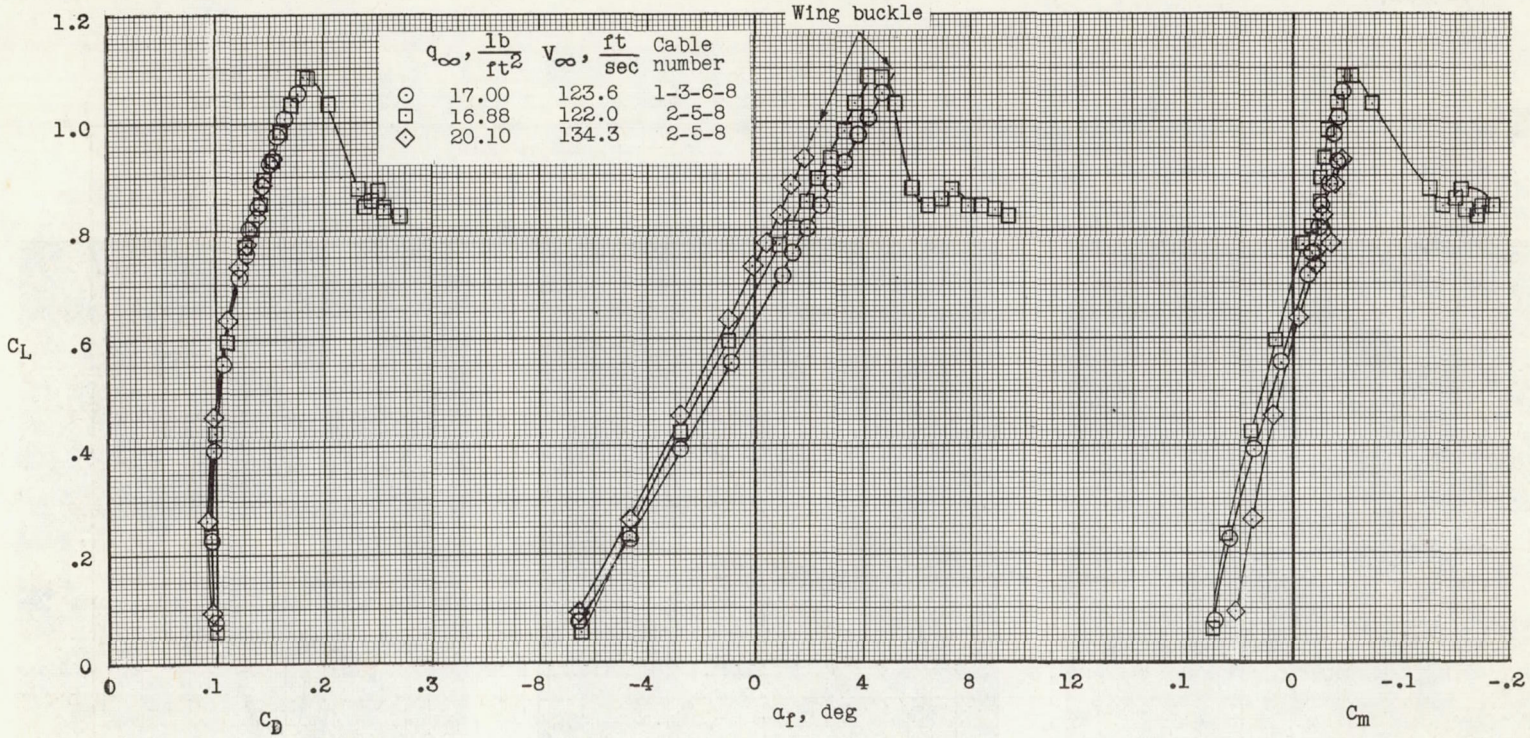
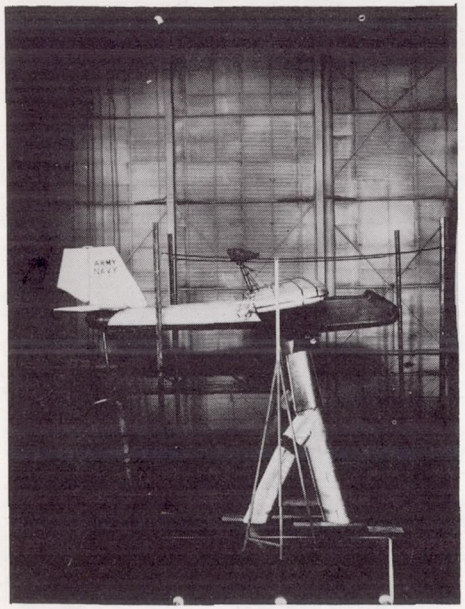
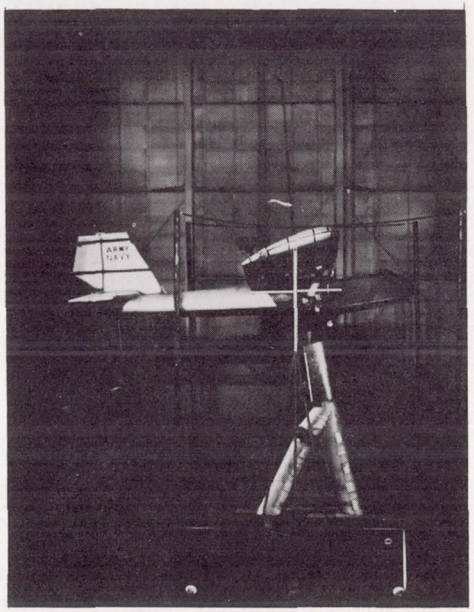


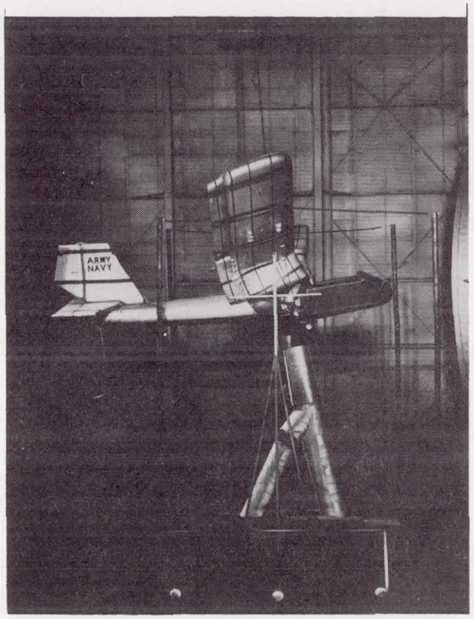
Figure 6.- Aerodynamic characteristics of the airplane for two cable configurations. Belly plate installed; lower forward cables heavily tightened in static condition; p = 7.0 lb/sq in.



$V_\infty = 0 \frac{ft}{sec}$
$\alpha_f = 3.5°$
$n = 0$



$V_\infty = 123.6 \frac{ft}{sec}$
$\alpha_f = 4.2°$
$n = 3.98$



$V_\infty = 123.6 \frac{ft}{sec}$
$\alpha_f = 5.2°$
n (just prior to buckle) = 4.22

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Figure 7.- Photographs of the airplane showing a typical wing deflection and buckle during tests. (Cross-pylon is used for deflection reference.)

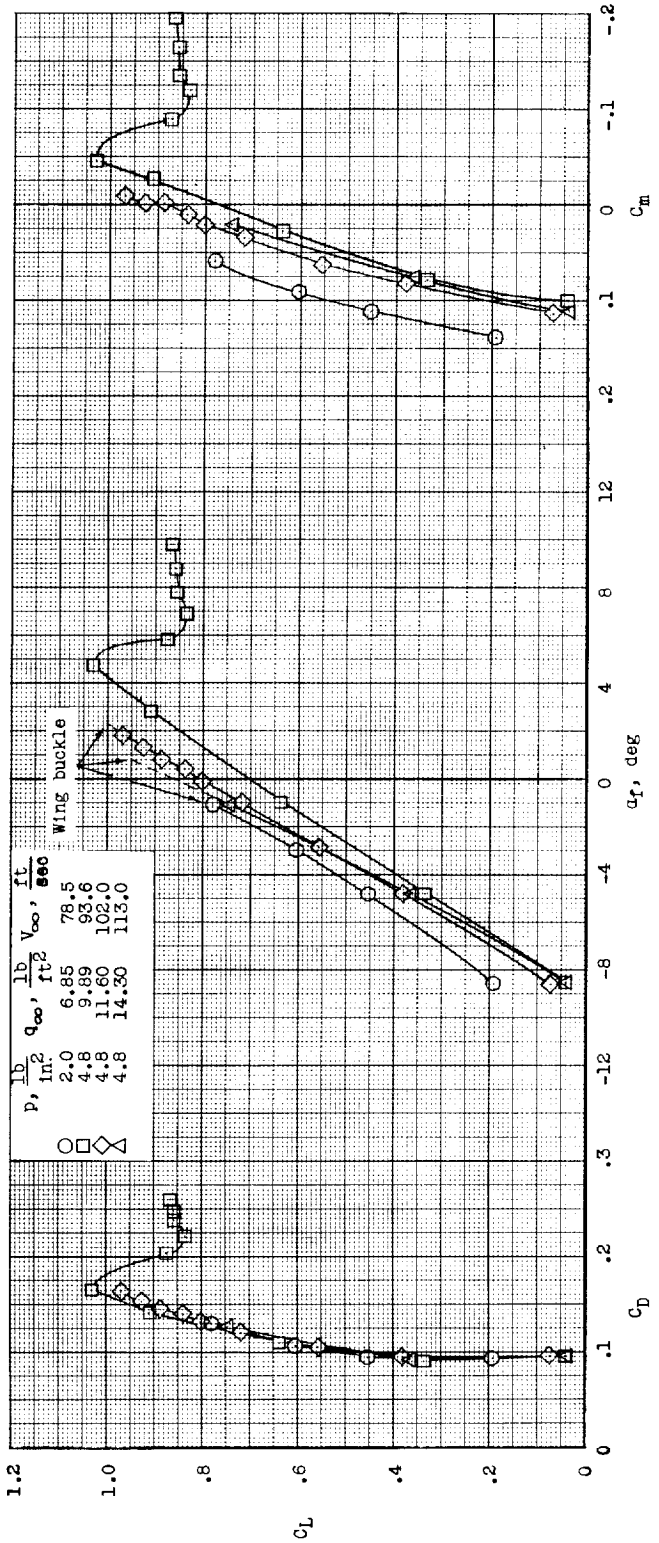


Figure 8.- Aerodynamic characteristics of the airplane at reduced inflation pressure. Cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened in static condition.

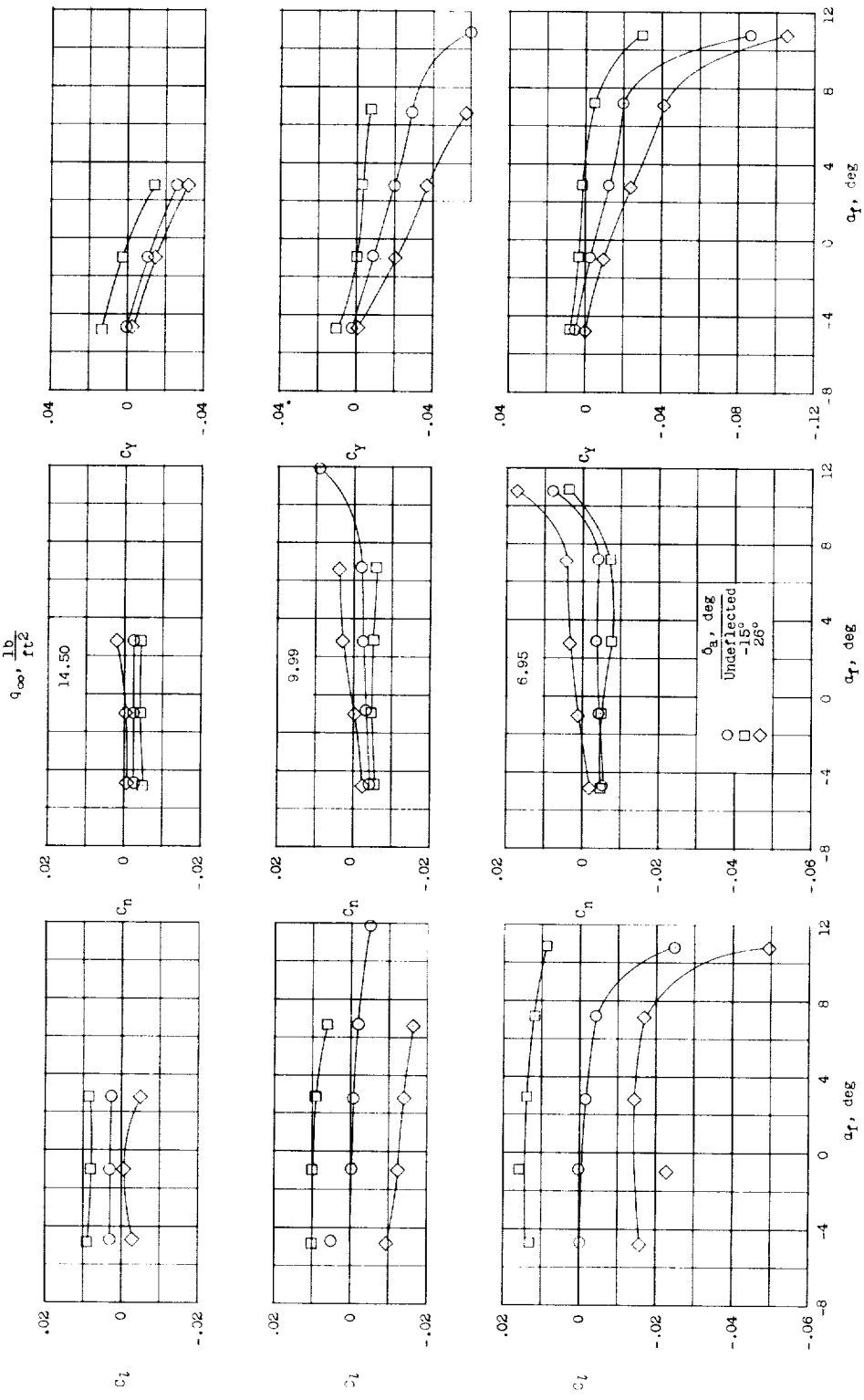


Figure 9.- Variation of lateral characteristics with angle of attack and right-hand-aileron deflection for three airspeed conditions. Cable configuration 2-5-8 with belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.

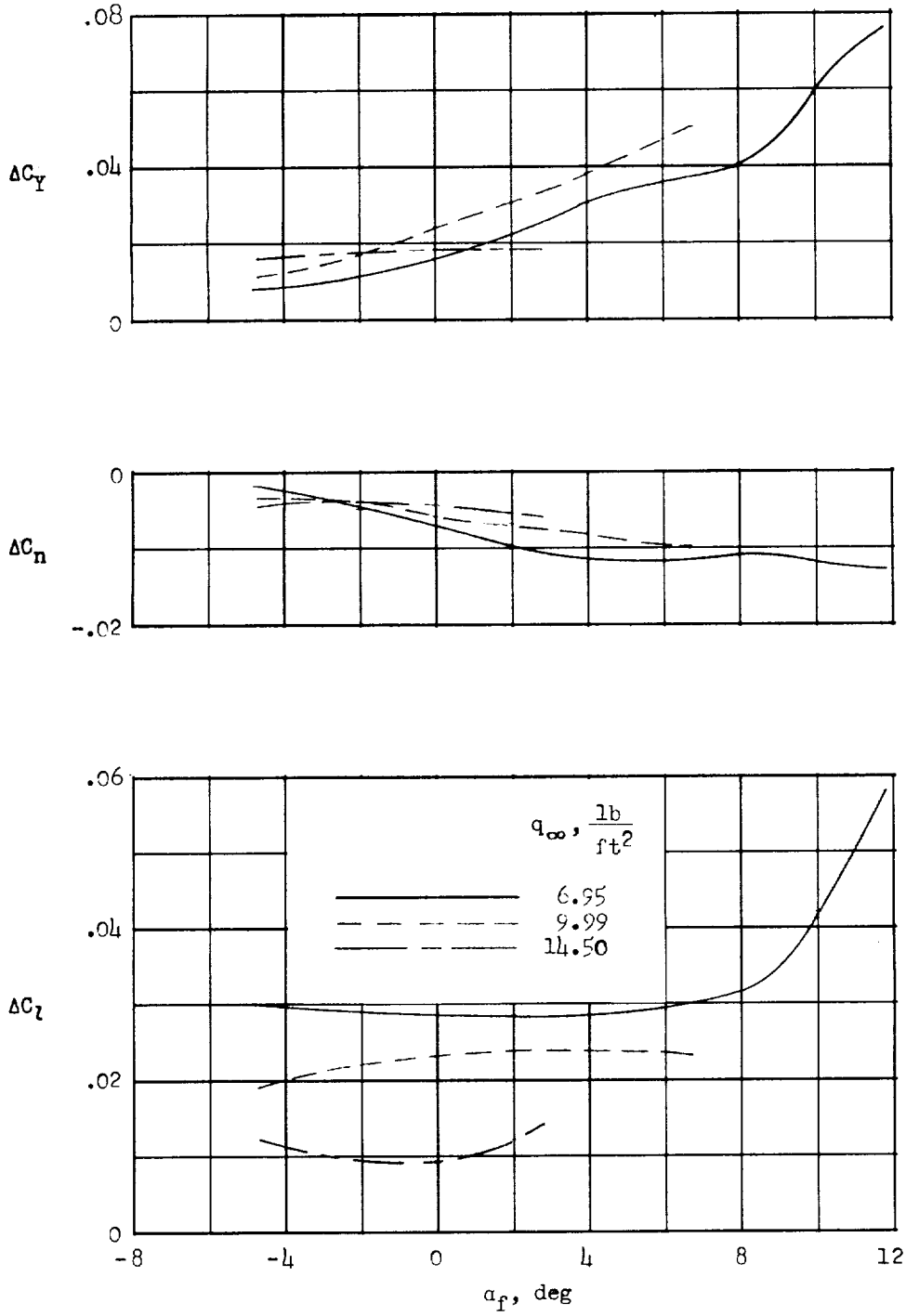


Figure 10.- Increment of C_l, C_n, and C_Y produced by deflecting the right-hand aileron up 15° and the left-hand aileron down 26°.

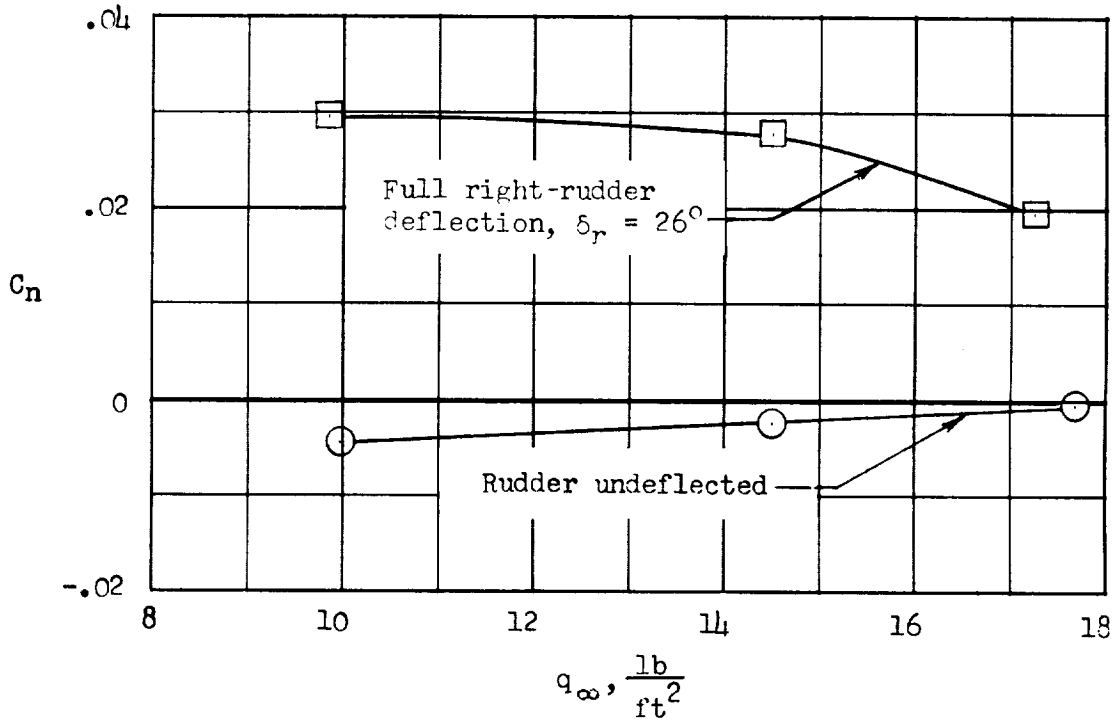


Figure 11.- Effect of full right-rudder deflection on the yawing-moment characteristics of the airplane at an angle of attack of the fuselage of about -7.5°. Cable configuration 2-5-8 with belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.

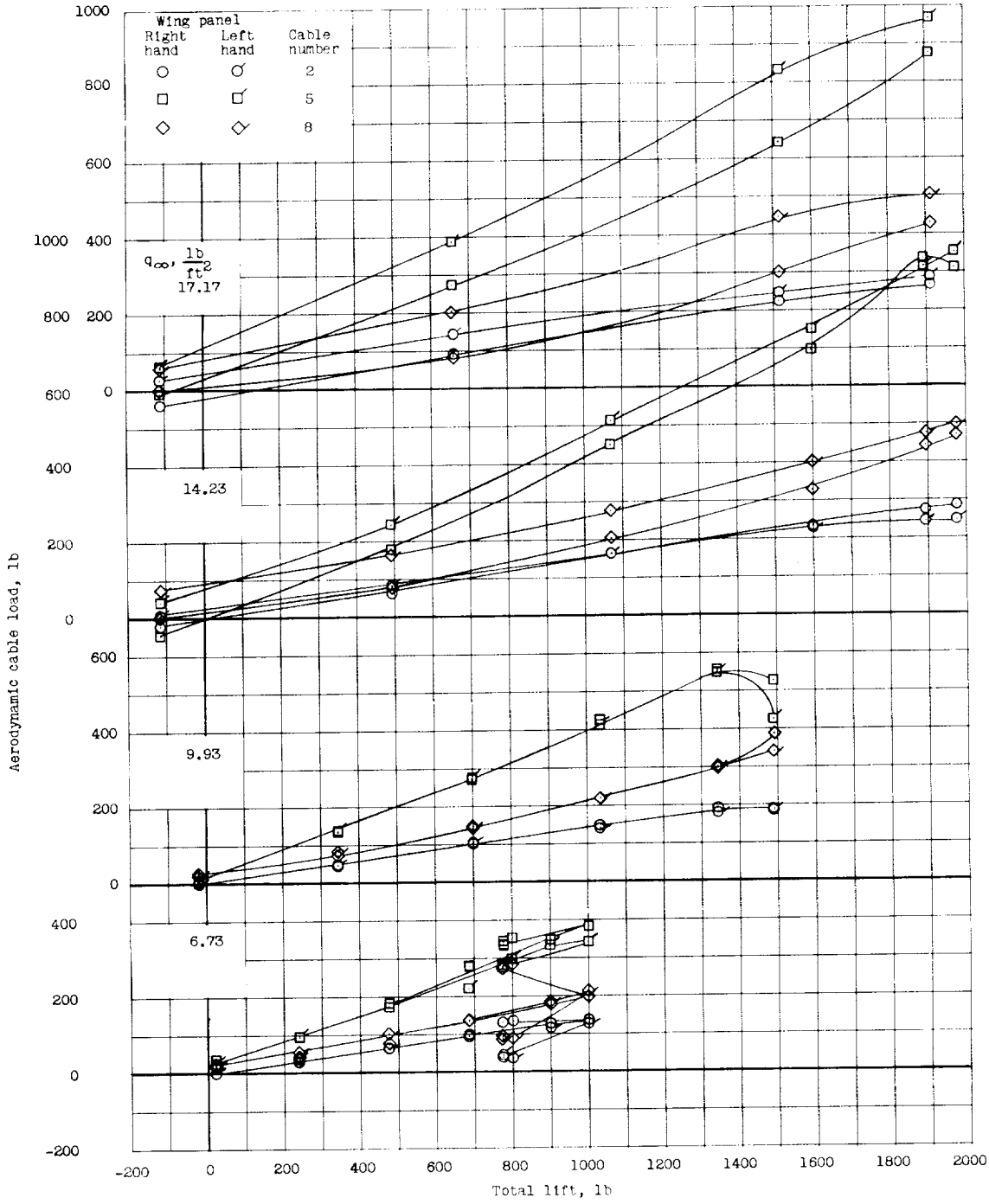


Figure 12.- Variation of the wing-guy-cable loads with total lift of the configuration for several airspeeds. Original cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened; p = 7.0 lb/sq in.

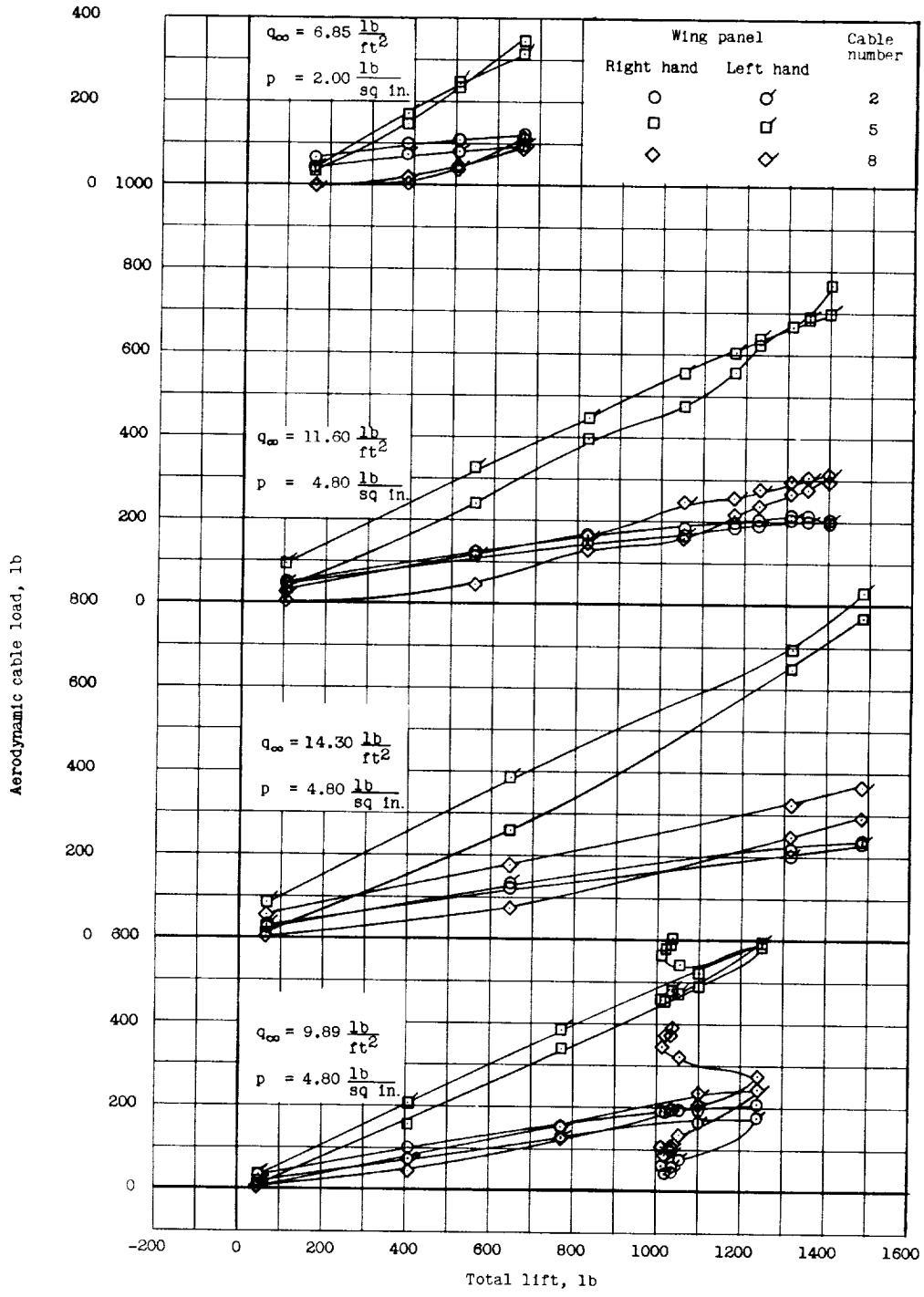


Figure 13.- Variation of the wing-guy-cable loads with total lift of the configuration for reduced inflation pressures. Original cable configuration 2-5-8 without belly plate installed; lower forward cables lightly tightened.

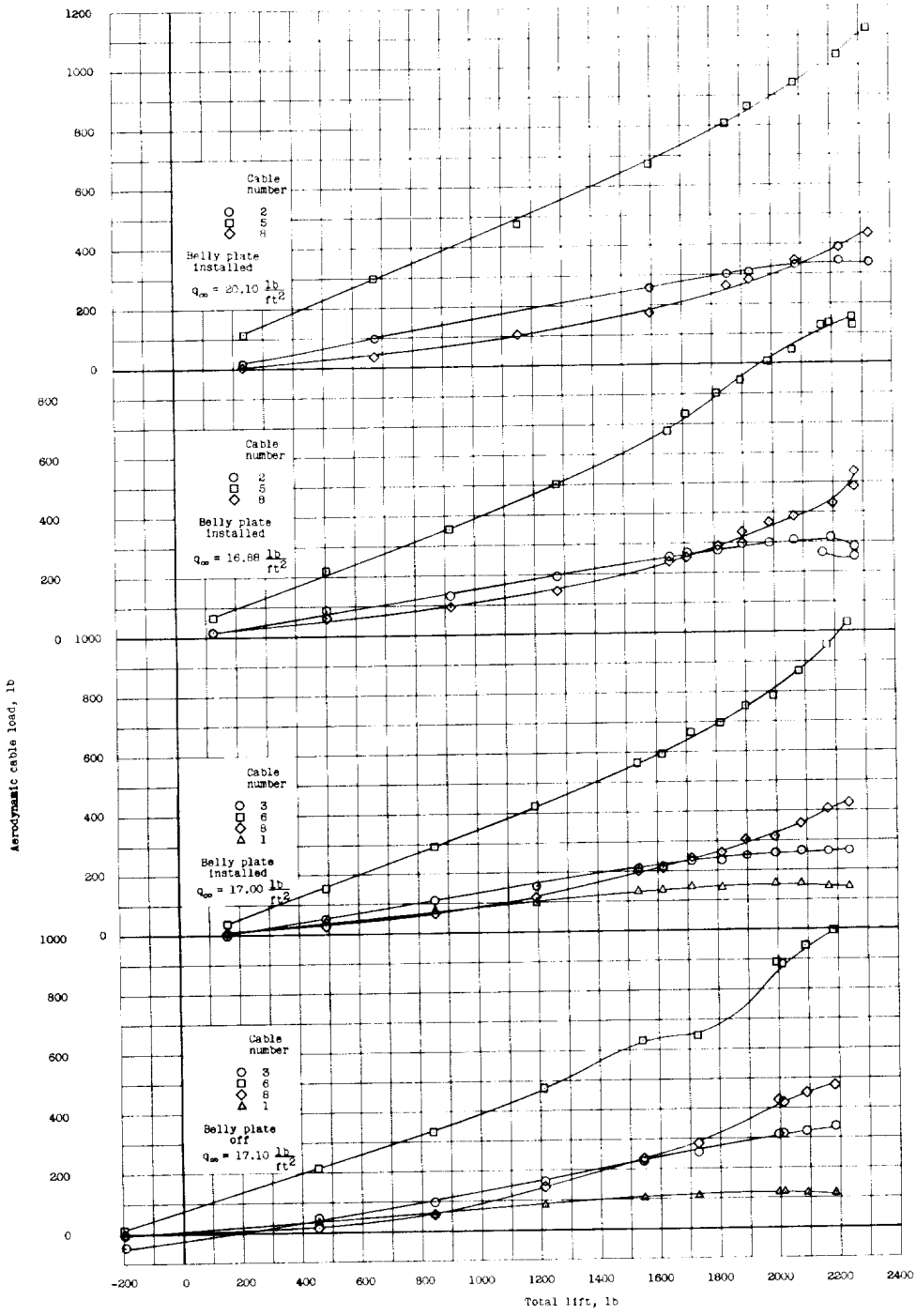


Figure 14.- Variation of the right-hand-wing guy-cable loads with total lift of the configuration. Lower forward cables heavily tightened; p = 7.0 lb/sq in.

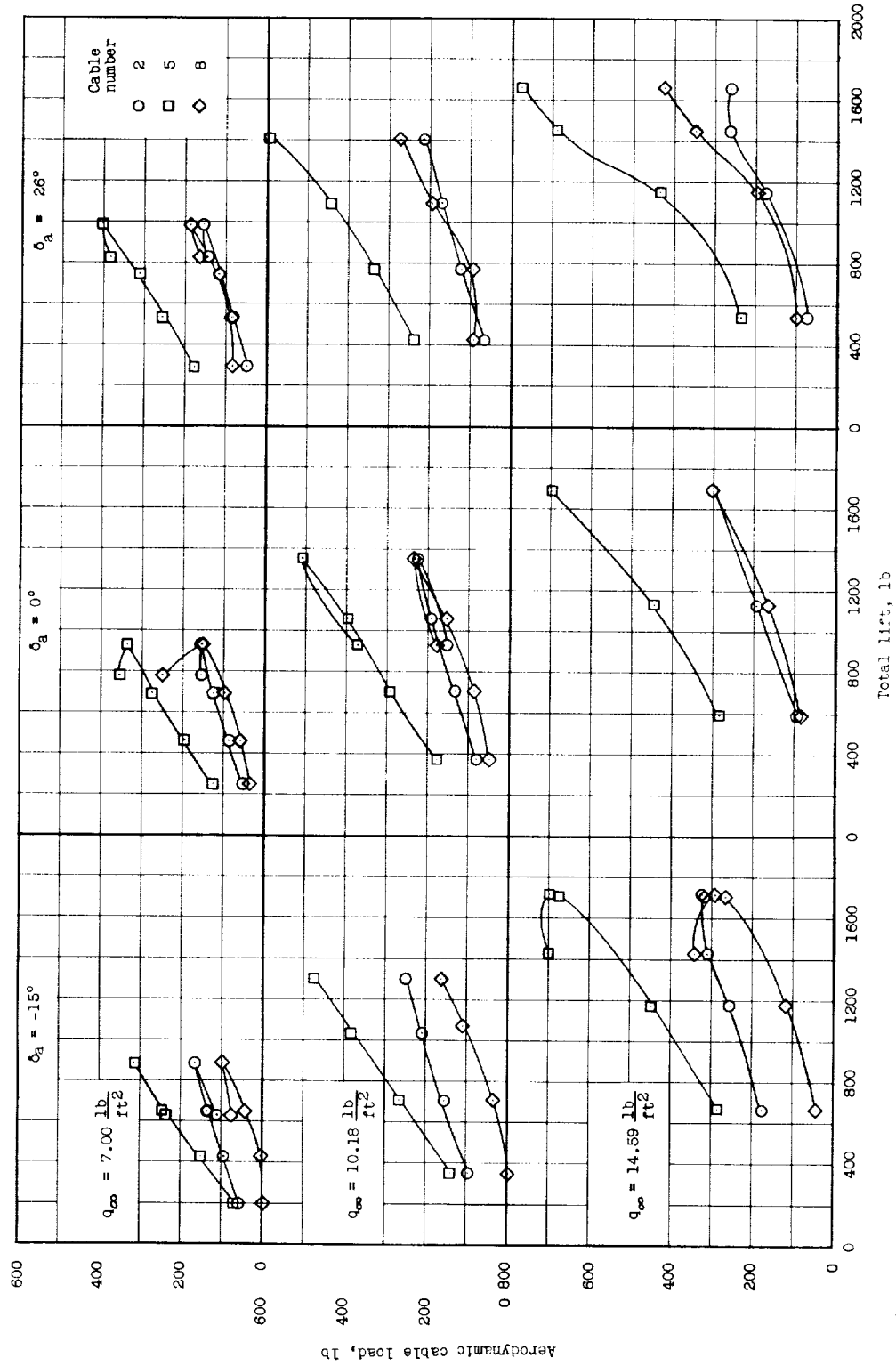
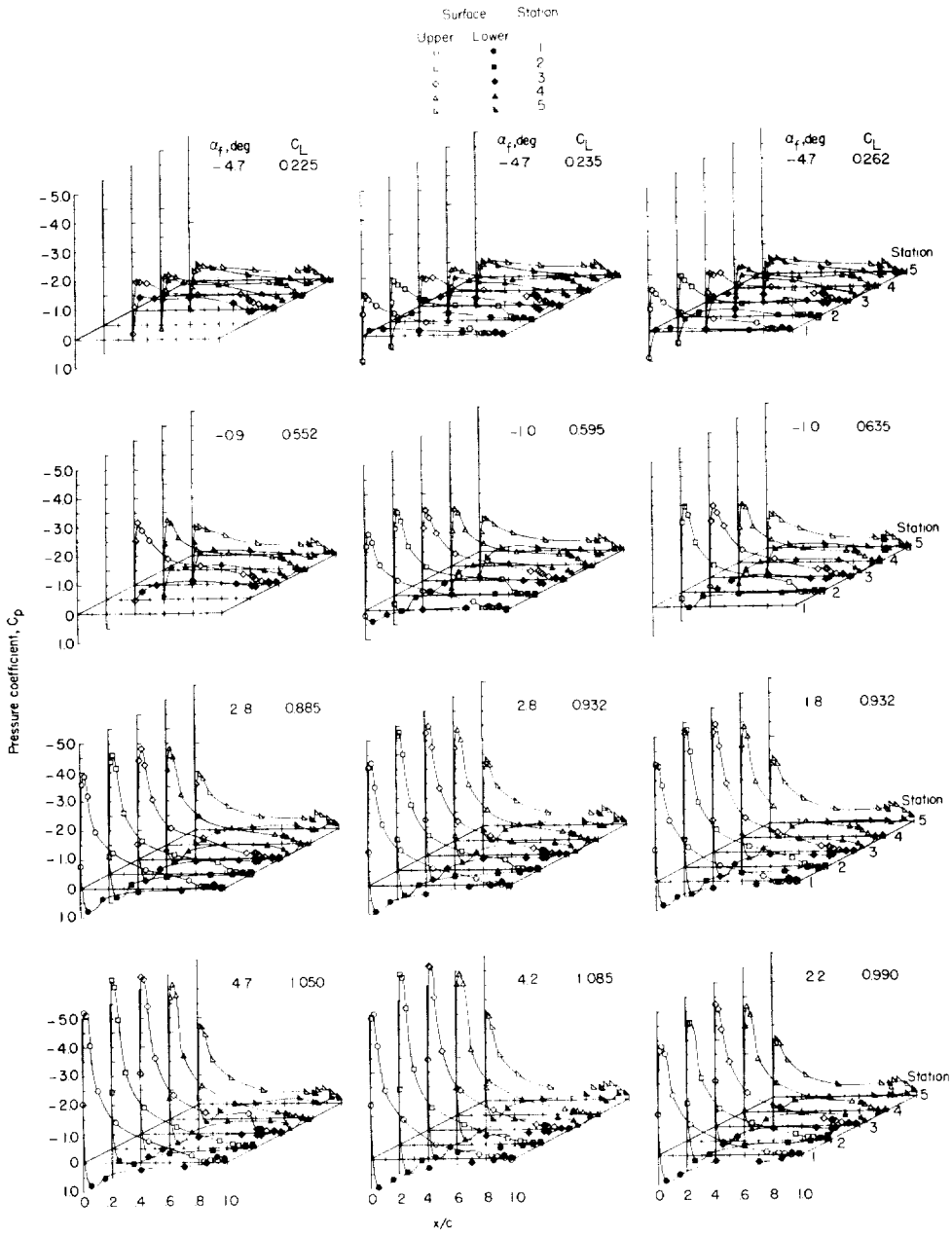


Figure 15.- Variation of the right-hand-wing cable loads with total lift of the configuration for several right-hand-aileron deflections and for several tunnel speeds. Belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.

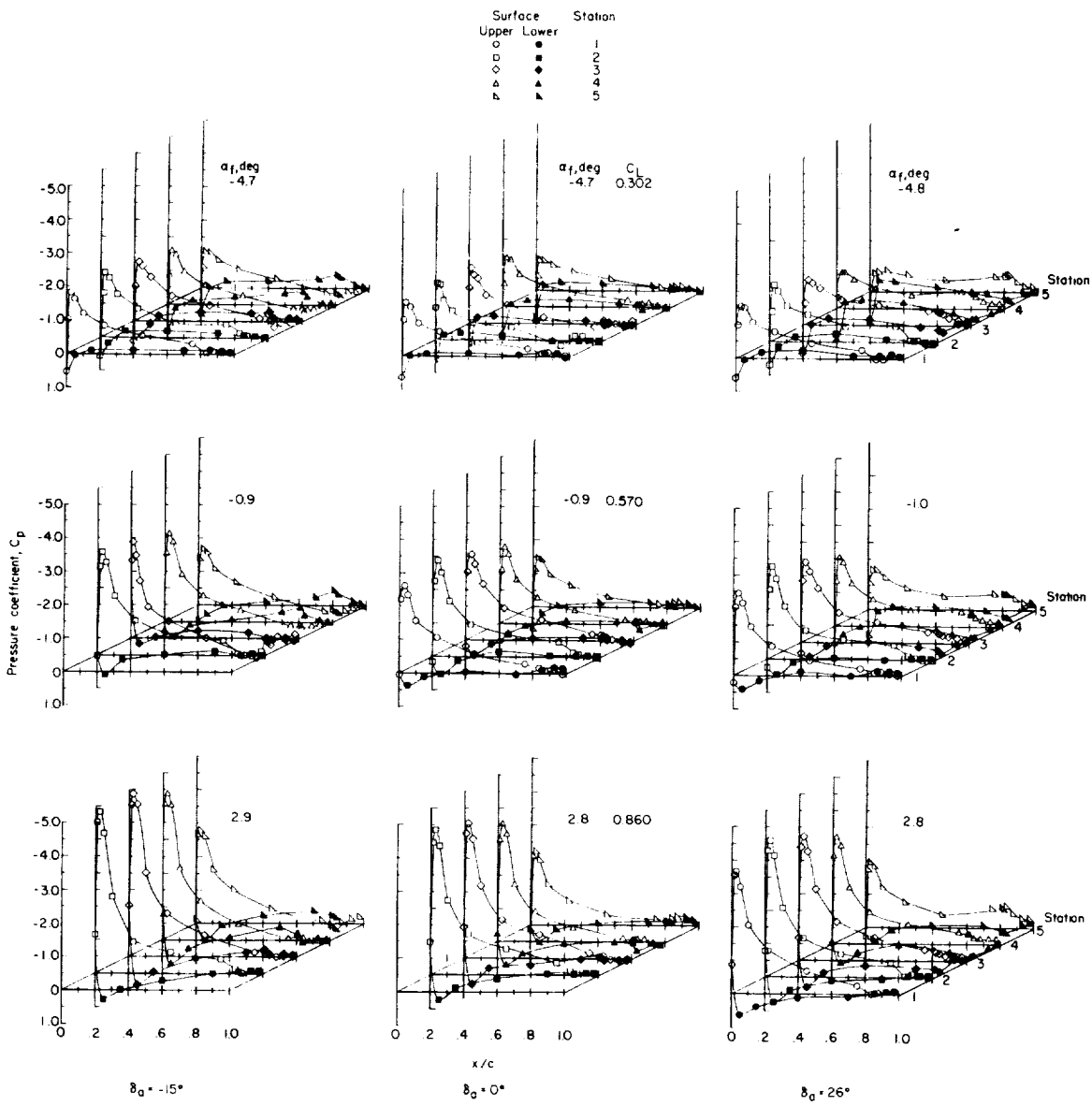


(a) Cable configuration 1-3-6-8; V_∞ = 123.6 ft/sec.

(b) Cable configuration 2-5-8; V_∞ = 122.0 ft/sec.

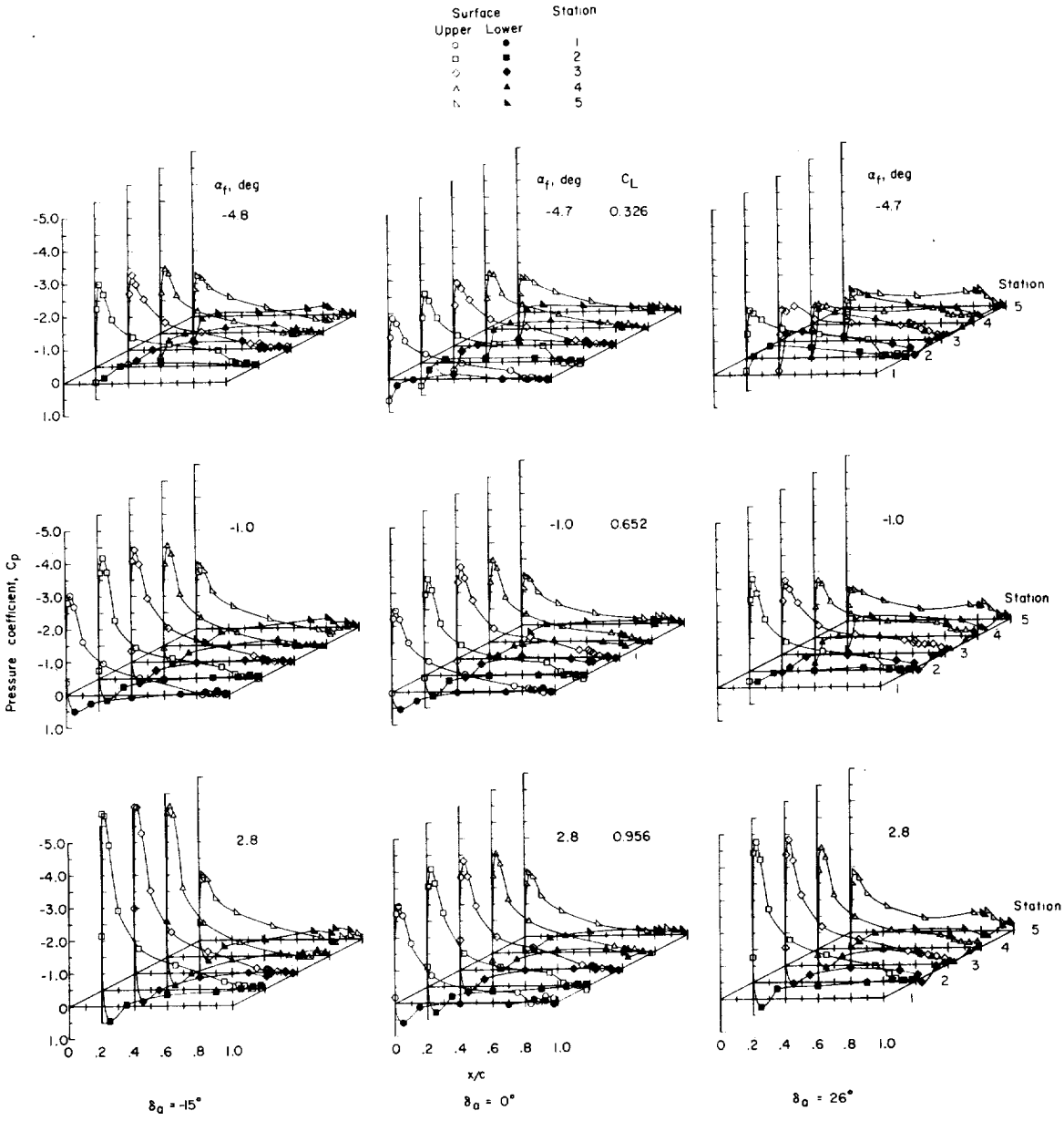
(c) Cable configuration 2-5-8; V_∞ = 134.3 ft/sec.

Figure 16.- Variation of the chordwise pressure distribution for two wing cable configurations and for increased test velocities. Belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.



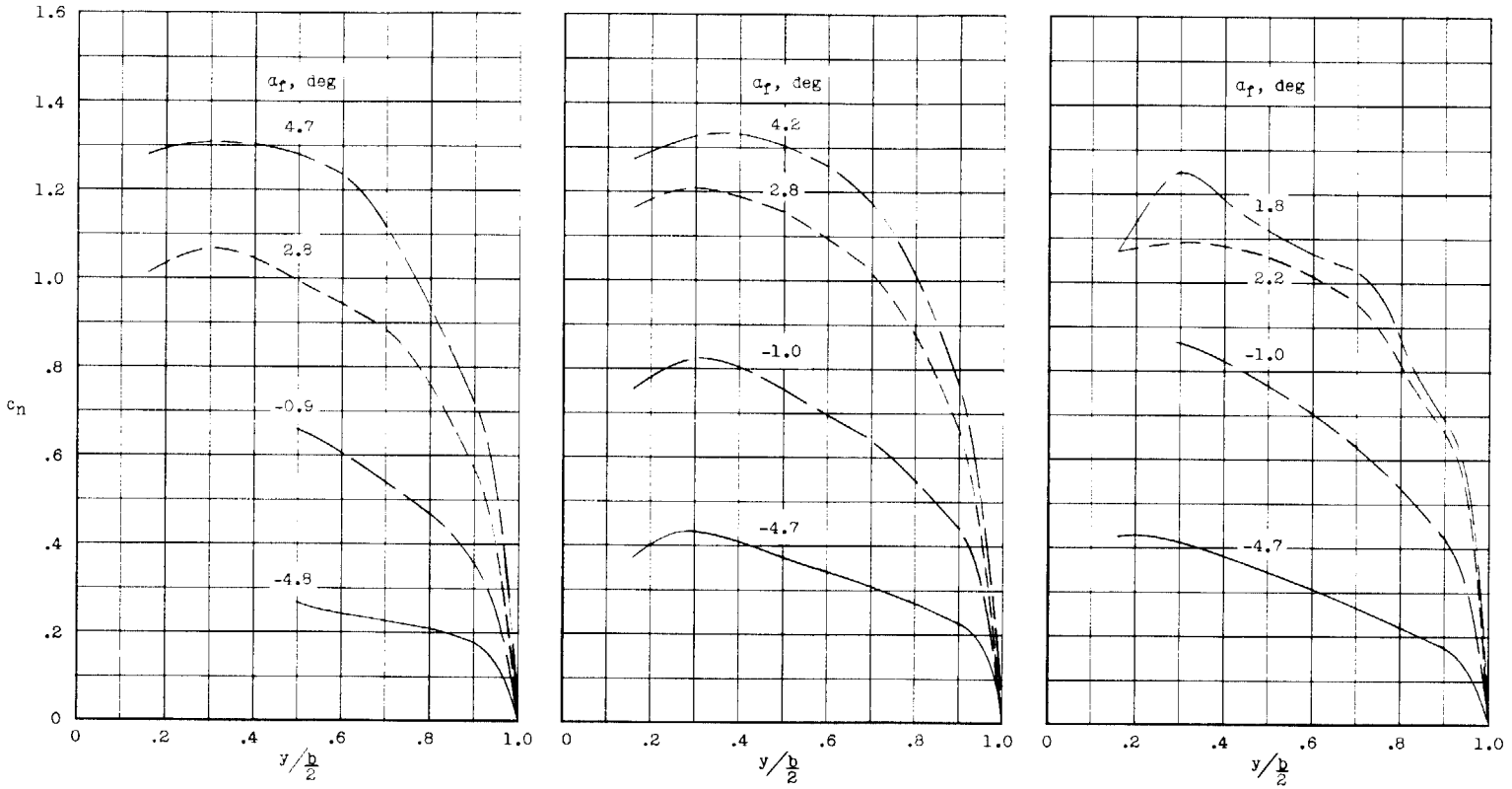
(a) V_∞ = 94.0 ft/sec.

Figure 17.- Variation of the chordwise pressure distributions with aileron deflection of cable configuration 2-5-8. Belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.



(b) V_∞ = 113.5 ft/sec.

Figure 17.- Concluded.

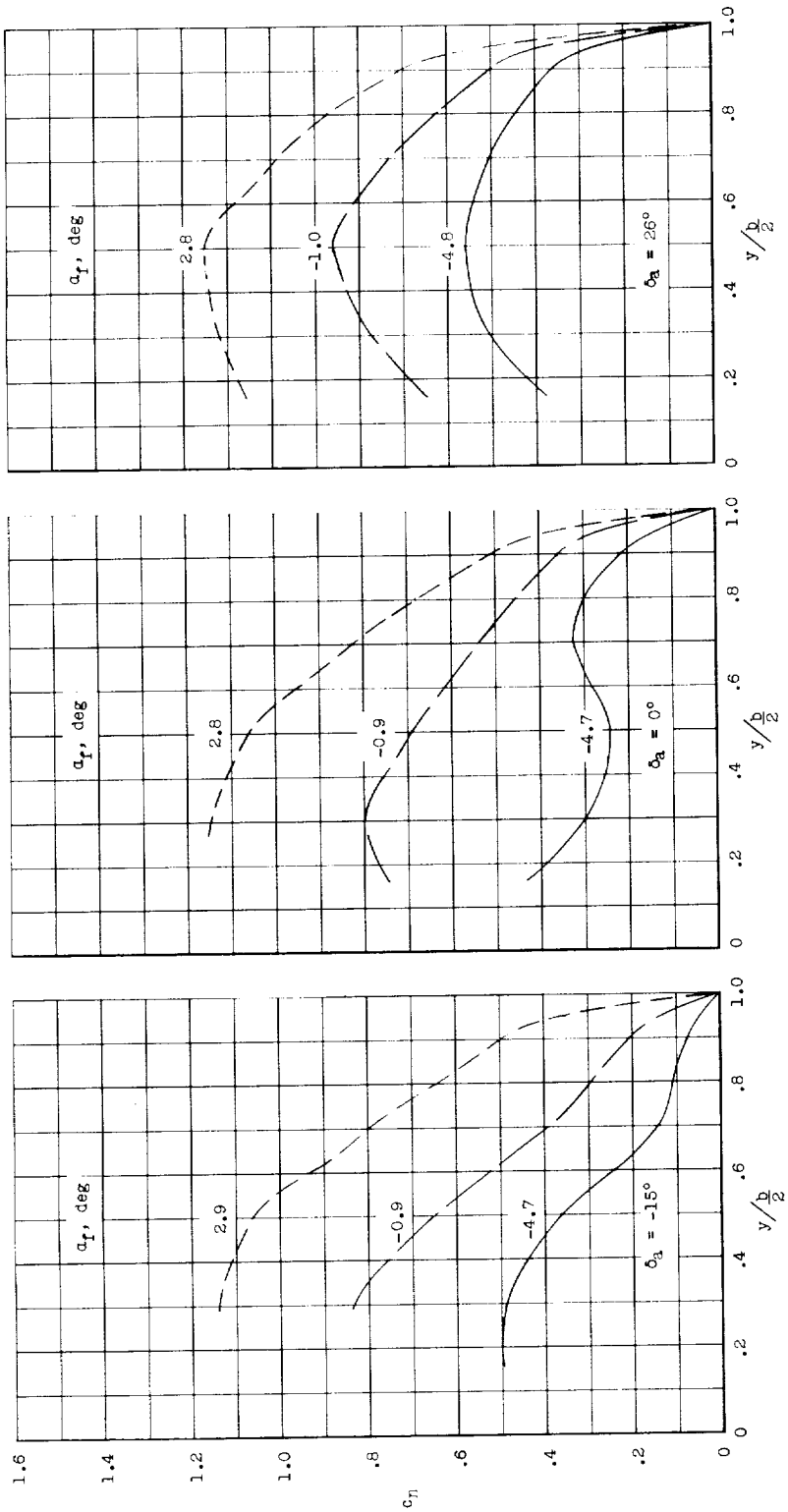


(a) Cable configuration 1-3-6-8; V_∞ = 123.6 ft/sec.

(b) Cable configuration 2-5-8; V_∞ = 122.0 ft/sec.

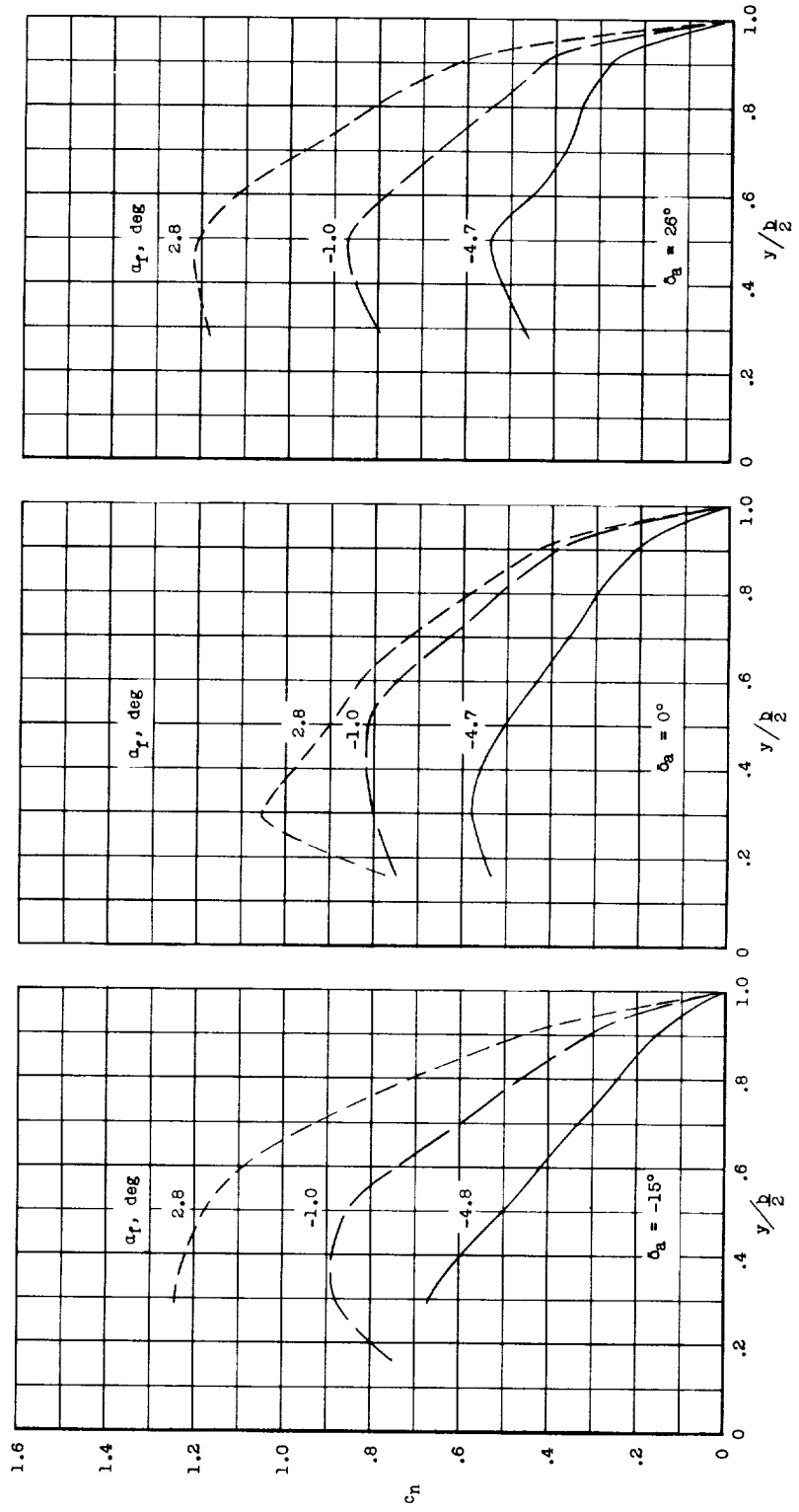
(c) Cable configuration 2-5-8; V_∞ = 134.3 ft/sec.

Figure 18.- Span loading characteristics of two cable configurations. Belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.



(a) V_∞ = 94.0 ft/sec.

Figure 19.- Span loading characteristics of cable configuration 2-5-8. Belly plate installed; lower forward cables heavily tightened; p = 7.0 lb/sq in.



(b) V_∞ = 113.7 ft/sec.

Figure 19.- Concluded.